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# April 2002 Working Group Meeting on Heavy Vehicle Aerodynamic Drag: Presentations and Summary of Comments and Conclusions

*R. McCallen, K. Salari, J. Ortega, T. Dunn, LLNL; F. Browand, D. Arcas, M. Hammache, USC; A. Leonard, P. Chatelain, M. Rubel, Caltech; M. McWherter-Payne, C. Roy, W. Rutledge, SNL; J. Ross, D. Satran, J.T. Heineck, B. Storms, NASA; D. Pointer, T. Sofu, D. Weber, ANL; E. Chu, P. Hancock, B. Bundy, PACCAR; B. Englar, GTRI*

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U.S. Department of Energy

Lawrence Livermore National Laboratory

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# April 2002
# Working Group Meeting on
# Heavy Vehicle Aerodynamic Drag:
# Presentations and Summary of Comments and Conclusions

***Jointly written by***
Lawrence Livermore National Laboratory
Sandia National Laboratories
University of Southern California
California Institute of Technology
NASA Ames Research Center
Georgia Tech Research Institute
Argonne National Laboratory

A Working Group Meeting on Heavy Vehicle Aerodynamic Drag was held at Lawrence Livermore National Laboratory on April 3 and 4, 2002. The purpose of the meeting was to present and discuss technical details on the experimental and computational work in progress and future project plans. Representatives from the Department of Energy (DOE) Office of Transportation Technology Office of Heavy Vehicle Technology (OHVT), Lawrence Livermore National Laboratory (LLNL), Sandia National Laboratories (SNL), NASA Ames Research Center, University of Southern California (USC), and California Institute of Technology (Caltech), Georgia Tech Research Institute (GTRI), and Argonne National Laboratory (ANL), Volvo Trucks, and Freightliner Trucks presented and participated in discussions. This report contains the technical presentations (viewgraphs) delivered at the Meeting, briefly summarizes the comments and conclusions, and outlines the future action items.

## Introduction, Overview of the Project, and Summary

The meeting began with an introduction by LLNL's Deputy Associate Director of the Energy and Environmental Directorate, Ray Smith, where he emphasized that the Nations dependence on oil is a national security issue and that minimizing vehicle aerodynamic drag will significantly reduce the dependence on foreign oil resources. Rose McCallen of LLNL followed with an overview of the DOE project goals, deliverables, and FY02 activities. The viewgraphs for the project introduction and LLNL overview are attached at the end of this report.

Sid Diamond of DOE OHVT announced to the participants that OTT was being reorganized and that certain key aspects of OTT such as OHVT have been incorporated into the FreedomCAR and Vehicle Technologies Programs. This represents a reduction from 6 to 2 Deputy Assistant Secretaries and a reduction of 31 to 11 offices. He assured all that the FY03 budget was secure and that information about FY04 would be forthcoming. Sid also emphasized the importance of reducing energy use to reduce our nations dependence on oil and the relation to national security. In addition to aerodynamic drag reduction, Sid mentioned the importance of developing means for high-density energy storage and efficient energy conversion. Jules Routbort of DOE OHVT/ANL also discussed the push for more electronics in vehicles because of lighter weight and durability.

In summary, the technical presentations at the meeting included a review of experimental results and plans by USC and NASA Ames, the computational results from LLNL and SNL for the integrated tractor-trailer benchmark geometry called the Ground Transportation System (GTS) Model, and turbulence model development and benchmark simulation for rounded cube shapes representative of a tractor and trailer being investigated by Caltech. NASA Ames also presented information on the new geometry called the Generic Conventional Model (GCM) that was evaluated last year in the 7-ft. x 10-ft. wind tunnel at NASA and plans for testing in the 12-ft pressure wind tunnel this year. USC is also investigating an acoustic drag reduction device that has been named 'Mozart' and GTRI continues their investigation of a blowing device. ANL presented their plans for a DOE supported Cooperative Research and Development Agreement (CRADA) with Paccar Truck Company utilizing commercial software tools to simulate the flow and drag for an actual Tractor. Much of the discussion involved wind tunnel testing plans, analysis of existing experimental data, investigations of drag reduction devices, simulation results, and needed modeling improvements. Further details are provided in the attached viewgraphs.

## Project Goals, Deliverables, and Future Activities

Based on discussions at the Meeting, the project goals remain unchanged:

- Perform heavy vehicle computations to provide guidance to industry
- Using experimental data, validate computations
- Provide industry with design guidance and insight into flow phenomena from experimental and computations
- Investigate aero devices (e.g., boattail plates, side extenders, blowing and 'Mozart' device)

The following additional activities were identified:

1) Invite industries' overseas R&D contacts to UEF Conference.
2) All DOE Team members submit abstracts to UEF Conference.
3) Obtain more funding for UEF Conference.
4) Submit papers for SAE March 2003 conference. The paper submission deadline is June 1st and final manuscripts are due December 10. (Participation by the Team may be limited because of demand by UEF Conference.)
5) Respond to DOE/OHVT request for proposals (RFP) in collaboration with Freightliner on topic of full-scale experiments, instrumentation techniques, and computations.
6) Discuss with International a possible RFP on splash and spray. (USC has a small moving-ground-plane wind tunnel coming online in about six months and LLNL is interested in spray modeling.)
7) LLNL will consult Caltech on guidance in improving boundary layer (near wall) treatment with LES.
8) Demonstrate use of smaller machines (e.g., Linux/PC clusters).
9) RANS for FY02

a) SNL: Simulate GTS at 0 degree yaw using 1) Wilcox k-omega, 2) Spalart-Almaras, and maybe 3) k-epsilon turbulence models for a minimum of 2 grids and if possible, 3 grids for each.
b) LLNL:
   i) Document GTS and Texas A&M simulations using Spalart-Almaras model with 2 grids at 0 degree yaw and 1 grid at 10 degree yaw
   ii) Attempt GCM simulation using Overflow code with RANS k-omega turbulence model at 0 degree yaw with 1 grid

## Technical Discussion Highlights

### Analysis of NASA's Experimental Data on GTS and GCM Geometries in the NASA 7-ft x 10-ft Wind Tunnel

Jim Ross of NASA Ames provided some interesting findings through their analysis of the data from tests done on the GTS geometry in the 7-ft x 10-ft wind tunnel at NASA Ames. The instantaneous PIV measurements of the wake flow were evaluated by 'conditioned sampling'. Condition sampling is performed by calculating the instantaneous vorticity from the measured instantaneous velocity, then searching the results for the maximum vorticity location. This location should point to the center of an eddy, thus, capturing the vortex shedding from the rear of the trailer.

Analysis of the results indicate a Strouhal number St = FL/V = 0.128 where F is the vortex shedding frequency (approximately 1180 Hz), L is the boundary layer thickness upstream, and V is the freestream velocity (approximately 92 m/s). It is also observed that there is not a strong correlation of the vortex shedding from the top and bottom of the trailer and that boattail plates not only narrow the wake, but they stabilize it as indicated by a reduction in wake flapping with the bottail plates.

Analysis of PIV data in the gap of the GCM geometry indicates a hysteresis in the flow. It was found that the established recirculating gap flow persists for variations in yaw until the flow finally 'blows through' at the highest yaw angles. What is important to note is that the vehicle exhibits the lowest drag at the yaw angle where blow through occurs. If this blow through characteristic can be artificially reproduced, it can provide a significant reduction in drag. It was also noted that side extenders significantly reduce drag and do not exhibit flow hysteresis.

### Full-Scale Experimental Demonstration of Pneumatic Heavy Vehicles

The team of GTRI, Novatek Inc, Volvo Trucks, Great Dane Trailers and American Trucking Associations has designed and assembled a full-scale demonstrator to evaluate the blown Pneumatic Heavy Vehicle (PHV) during road tests. This configuration is based on the GTRI-developed and tested model configuration with blowing applied to all 4 specially-modified trailing edges. Previous tunnel tests had verified 45-50% reduction in drag coefficient using only 1/2 psi of blowing pressure, and up to 84% $C_D$ reduction if more pressure were available (say from the turbocharger), with a measured $C_D$ less than any production automobile (see attached viewgraphs). After the PHV test vehicle modifications were completed, two preliminary road tests were conducted in early 2002 to confirm that all blowing and data systems were operational and to optimize the blowing systems for the planned SAE Type-II fuel economy testing. These preliminary tests at Volvo's facilities showed initial drag reductions yielding fuel economy improvements of

10-15%, depending upon blowing rate and vehicle speed. Minor system improvements have been made in preparation for the SAE Type-II fuel economy test scheduled for July 2002 at the Transportation Research Center's 7.5 mile test track in Ohio.

**Determining Weaknesses and Strengths of Reynolds-Averaged Navier-Stokes (RANS) Turbulence Modeling**

Walt Rutledge of SNL discussed the wall resolution requirements for RANS turbulence modeling. Calculations indicate that RANS simulations do not show convergence to a steady solution if the y+ of the grid is too large ($y+ = u_\tau y/\nu$, where $u_\tau = (\tau_w/\rho)^{1/2}$ the friction velocity) and is a measure of how well the flow boundary layer is being captured). With the Wilcox k-omega model, a y+ of 2 or less is required for solution convergence, whereas the standard k-epsilon model requires a y+ of 10 or less for solution convergence.

**Advantages of Overset Grid Technology**

Dora Yen-Nakafuji of LLNL demonstrated the benefits of using overset grid technology. Overset grids provide the flexibility of defining a simple regular grid for the freestream flow in the wind tunnel while allowing the user to separately specify and overlay a fine grid around the vehicle geometry. Thus, the addition of even more detailed components, like side mirrors, is trivial. This technology is currently being utilized by the industry in evaluating production aircraft.

In addition to their work with finite element methods and large-eddy simulation, the LLNL Team has recently been applying the NASA Overflow code, which uses overset grids with a steady Spalart-Allmaras (RANS) turbulence model. Preliminary simulations of the wind tunnel and GTS geometry show impressive performance (i.e., efficient use of computational resources and run time speed). The simulation runs well on a single processor PC and setup time is minimal. The LLNL Team plans to further investigate this technology for application to heavy vehicles and work with NASA to possibly incorporate an advance turbulence modeling technique for large-eddy simulation with the overset technology.

**Truck Aero Team Meeting Attendees**

**LLNL, Livermore, CA**

**April 3 & 4, 2002**

| Attendee | Organization | e-mail address and phone |
|---|---|---|
| Diego Arcas | USC | arcasrod@usc.edu, 323-936-7298 |
| Fred Browand | USC | browand@spock.usc.edu, 213-740-5359 |
| Sid Diamond | DOE | sid.diamond@ee.doe.gov, 202-586-8032 |
| Tim Dunn | LLNL | dunn13@llnl.gov, 925-422-5258 |
| Bob Englar | GTRI | bob.englar@grti.gatech.edu, 770-528-3222 |
| Mustapha Hammache | USC | hammache@usc.edu, 213-740-5377 |
| J. T. Heineck | NASA ARC | jheineck@mail.arc.nasa.gov, 650-604-0368 |
| Tsun-Ya Hsu | USC | tsunyah@spock.usc.edu, 213-740-0516 |
| Tony Leonard | Caltech | tony@galcit.caltech.edu, 626-395-4465 |
| Matt Markstaller | Freightliner | MattMarkstaller@freightliner.com, 503-745-6857 |
| Rose McCallen | LLNL | mccallen1@llnl.gov, 925-423-0958 |
| Mary McWherter-Payne | SNL | mapayne@sandia.gov, 505-844-8500 |
| Dora Nakafuji | LLNL | nakafuji2@llnl.gov, 925-422-8670 |
| Jason Ortega | LLNL | ortega17@llnl.gov, 925-423-3824 |
| David Pointer | ANL | dpointer@anl.gov, 630-252-1052 |
| Jim Ross | NASA ARC | jcross@mail.arc.nasa.gov, 650-604-6722 |
| Jules Routbort | ANL/DOE | routbort@anl.gov, 630-252-5065 |
| Chris Roy | SNL | cjroy@sandia.gov, 505-844-9904 |
| Mike Rubel | Caltech | mrubel@caltech.edu, 626-395-8310 |
| Walt Rutledge | SNL | whrutle@sandia.gov, 505-844-6548 |
| Kambiz Salari | LLNL | salari1@llnl.gov, 925-424-4635 |
| Dale Satran | NASA ARC | dsatran@mail.arc.nasa.gov, 650-604-5879 |
| Ray Smith | LLNL | smith40@llnl.gov, 925-422-7802 |
| Dave Weber | ANL | dpweber@anl.gov, 630-252-8175 |
| Skip Yeakel | Volvo | **skip.yeakel@volvo.com** |

AGENDA

Heavy Vehicle Aerodynamic Drag: Working Group Meeting
Lawrence Livermore National Laboratory
Livermore, CA

April 3 & 4, 2002
Building 123, Conf. Room A

**Purpose of Meeting**

Presentation & discussion of industry's perspective and activities
Presentation & discussion of technical details of work in progress & future plans

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Wednesday, April 3

| | | |
|---|---|---|
| 7:30 — 8:00 | Badging at West Badge Office (Building 71) and travel to conference room | |

**Introduction**

| | | |
|---|---|---|
| 7:45 — 8:15 | Continental breakfast served in meeting room | |
| 8:15 — 8:30 | Welcome & introduction | Ray Smith, Rose McCallen |
| 8:30 — 9:00 | DOE/OHVT update & budget | Sid Diamond, Jules Routbort |

Work Plans and Progress: Experimental Effort and Devices

| | | |
|---|---|---|
| 9:00 — 9:15 | Overview and objectives | Rose McCallen |
| 9:15 —10:15 | NASA data reduction, analysis, documentation, & test plans | JT Heineck, Jim Ross, Dale Satran |
| 10:15 — 10:30 | Break | |
| 10:30 — 11:30 | USC experimental & numerical results for trailer-base add-ons: a progress report | Diego Arcas, Fred Browand, Mustapha Hammache, Tsun-Ya Hsu |
| 11:30 — 12:30 | GTRI test results & plans for aero device | Bob Englar |
| 12:30 — 1:15 | Lunch at LLNL served in meeting room | |

Work Plans and Progress: Computational Effort

| | | |
|---|---|---|
| 1:15 — 1:30 | Overview and objectives | Rose McCallen |
| 1:30 — 2:30 | SNL RANS computations, analysis & DES development | Walt Rutledge, Mary McWherter-Payne, Chris Roy |
| 2:30 — 3:30 | LLNL LES/DES incompressible computations/analysis & development | Kambiz Salari, Jason Ortega, Dora Yen-Nakafuji, Tim Dunn |
| 3:30 — 3:45 | Break | |
| 3:45 — 4:45 | Caltech vortex method development & computations | Philippe Chatelain, Tony Leonard, Mike Rubel |
| 4:45 — 5:45 | Results with a commercial tool | Dave Weber, Dave Pointer |

| | | |
|---|---|---|
| 5:45 — 6:00 | Discussion and Wrap-up | |
| 7:00 | Dinner at Kawa Sushi in Livermore | |

Thursday, April 4

| | | |
|---|---|---|
| 7:30 — 8:00 | Continental Breakfast | |

**Summary and Discussion**

| | | |
|---|---|---|
| 8:00 — 8:30 | Summary of issues from previous day, discussion | Rose McCallen |

Industry Perspective & Activities

| | | |
|---|---|---|
| 8:30 — 9:00 | Volvo | Skip Yeakel |
| 9:00 —10:00 | Overflow from previous day | |
| 10:00 — 10:15 | Break | |
| 10:15 —12:00 | Discussion & wrap up | |

**‘Working Group Meeting’**
**Consortium for Aerodynamic Drag of Heavy Vehicles**
**Department of Energy, Office of Heavy Vehicle Technology**
**April 3-4, 2002**

**Rose McCallen, Kambiz Salari, Tim Dunn, Jason Ortega, Dora Yen Nakafuji** — University of California, Lawrence Livermore National Laboratory

**Walter Rutledge, Mary McWherter-Payne, Chris Roy, David Kuntz** — Sandia National Laboratories

**James Ross, Dale Satran, J.T. Heineck, Bruce Storms, David Driver, James Bell, Steve Walker, Gregory Zilliac** — National Aeronautics & Space Administration

**Mustapha Hammache, Fred Browand, Tsun-Ya Hsu, Diego Arcas** — USC University of Southern California

**Anthony Leonard, Mike Rubel, Philippe Chatelain** — Caltech, California Institute of Technology

**Robert Englar** — Georgia Tech Research Institute

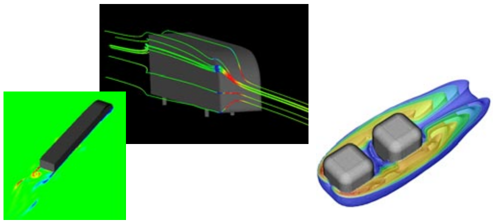
**David Weber, David Pointer, Tanju Sofu** — Argonne National Laboratory

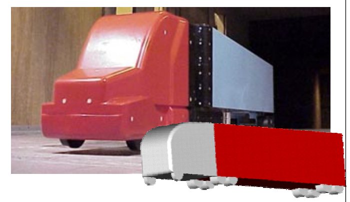
*Work performed under the auspices of the U.S. Department of Energy by the University of California, Lawrence Livermore National Laboratory under Contract W-7405-ENG-48.

## The consortium was formed to provide advanced technology to industry.


**Needed for significant impact on drag**
- Integrated tractor-trailer
- Drag reduction devices
- Aerodynamic
  - Front-end shape
  - trailer-base
  - components
  - underbody
- Improved thermal management (underhood flow)

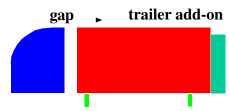
**Needed Technologies**
- Coupling experiments and computations for design guidance
  - Advanced computational methods and tools
  - Experimental validation
  - State-of-the-art experimental techniques
- Design and testing

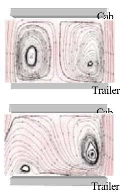




**LLNL, SNL, ANL, Caltech**
High quality numerical computations
Guidance on computational tools

**NASA, USC**
Data base of high quality
wind tunnel experiments

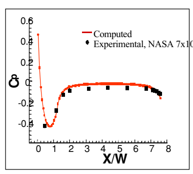


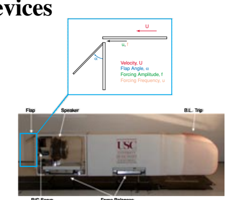




**USC, NASA, LLNL, SNL**
Comparisons and analyses
Insight into flow phenomena

**TEAM, Industry**
Information exchange

**USC, GTRI**
Concepts and designs of
aero devices



## The FY02 near-term deliverables include experiments, computations, design, and information exchange with industry.

**Guidance for the design of heavy vehicles**

- Analysis of existing experimental data
- Comparison to RANS, LES, and DES computations
- New Experiments: Re sensitivity, aero devices, gap and base drag, etc.

**Device to reduce base drag**

- Experimental validation of an acoustic device
- Full-scale road experiments on blowing device

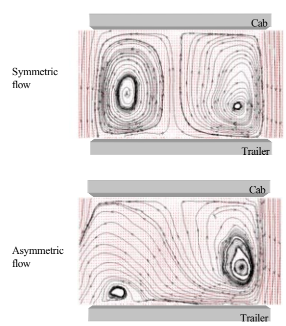
**Model development**

**Information exchange with industry**

- Working group meetings, conference papers, site visits

**Engineering Foundation Conference**

- "Aerodynamics of Trucks, Busses, and Railcars"



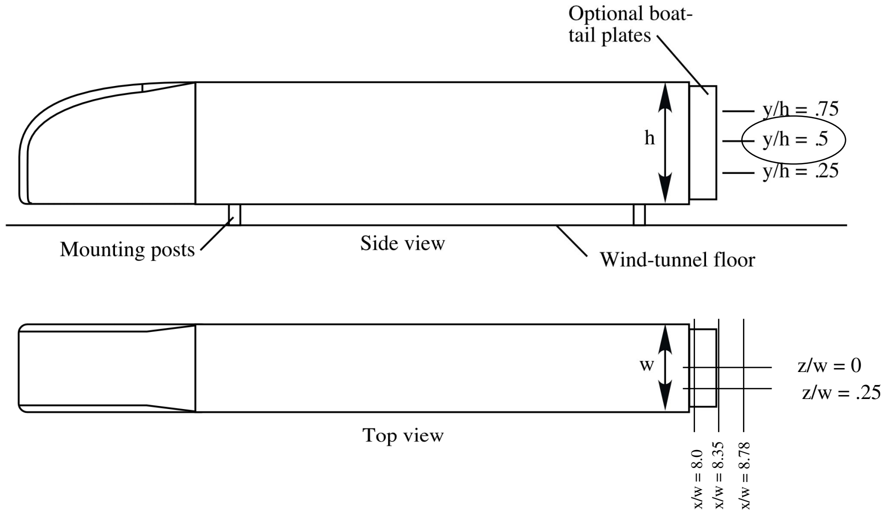
# GTS Wake Analysis
## Flow Structures and Effect of Boat-Tail Plates



# Outline

- **Analysis method to facilitate comparison between instantaneous PIV and LES results**
- **Look at how boat-tail plates modify wake**
- **Corrections to PIV data to fix Δt uncertainty**

# GTS Wake Measurement Planes

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# Conditional Sampling of PIV Data

- **Accepted a data set based on level of vorticity in a prescribed area in wake shear layer**
- **Can be sampled for both left and right shedding events**
- **Proper selection of level and sample area gave 6-12 hits per 100 data sets**

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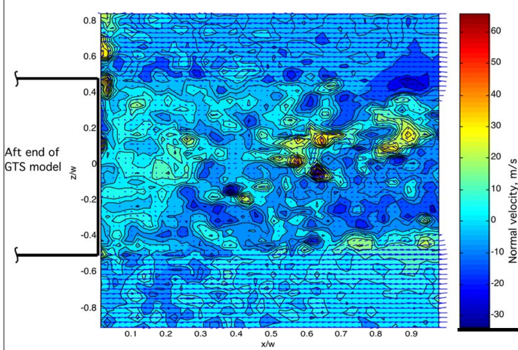
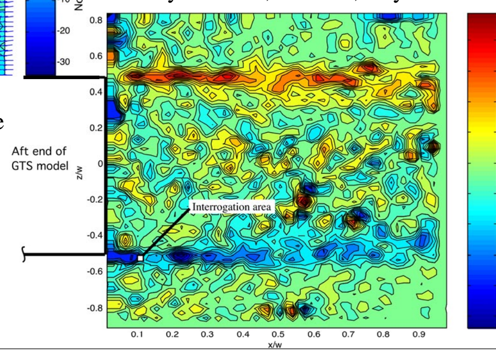
# Instantaneous PIV Data



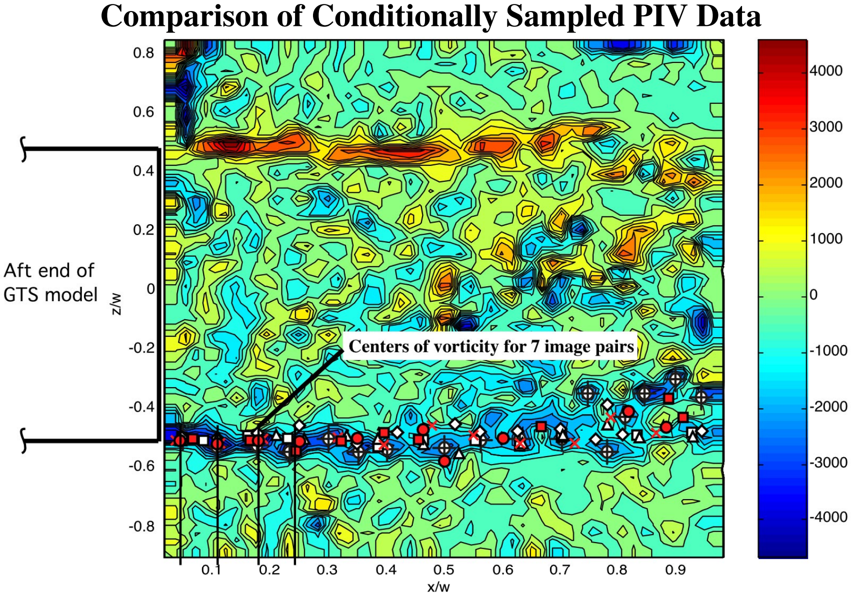

Instantaneous velocity field in wake
y/h = 0.75, M = .27, 0° yaw.

Data is on 56x66 grid (3696 points)

Instantaneous normal vorticity in wake
y/h = 0.75, M = .27, 0° yaw.




# Comparison of Conditionally Sampled PIV Data

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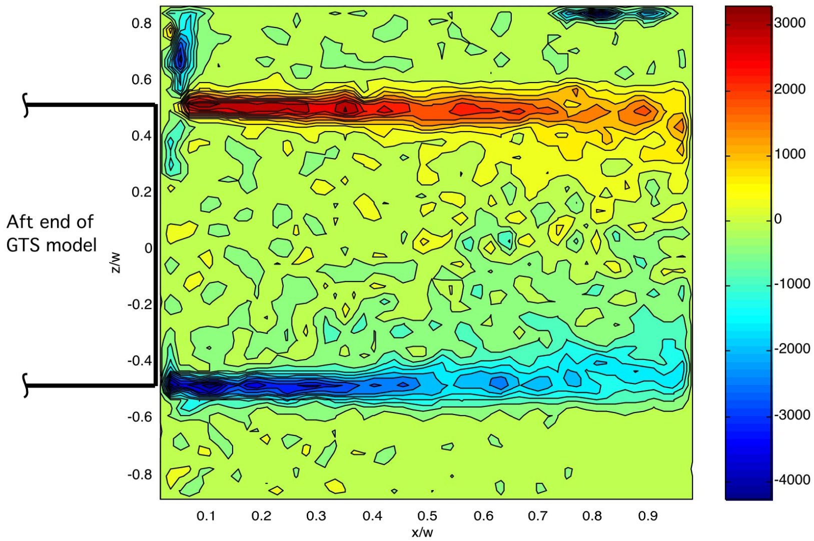
## “Phase” Averaged Vorticity

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## Strouhal Number of Shear Layer Flow Structure

$$St = \frac{f\,L}{V}$$

$f$ = frequency, $L$ = characteristic length, $V$ = reference velocity

For a turbulent shear layer, $St$ = 0.128 where L = Maximum slope thickness, and $V = V_\infty$ (Browand & Trout)

With V = 92 m/s, the shedding should occur at a frequency of ~ 1180 Hz.

The spacing between eddies is 0.021m giving a convection velocity of 25 m/s

For this kind of shear layer, the convection velocity should be 50-60% of free stream so …?

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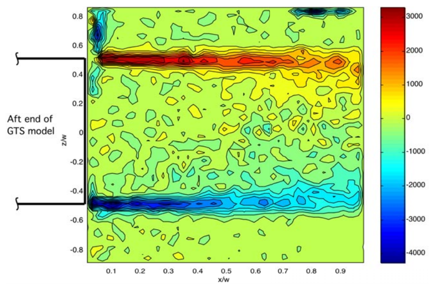
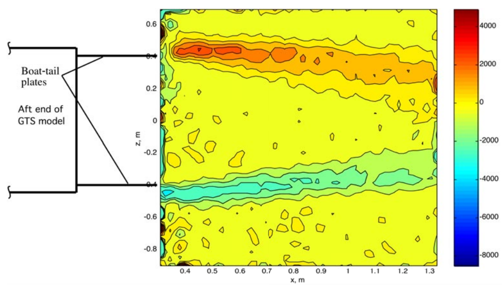
# Effect of Boat-Tail Plates on Wake

- Boat-tail plates cause wake to close more quickly
- Also stabilize the wake, reducing the lateral oscillations

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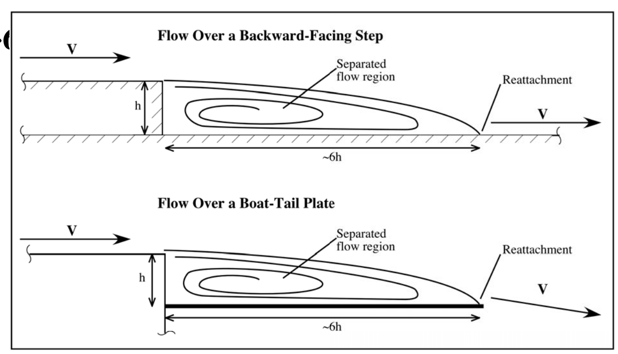
# Flow Mechanism Responsible for Boat-Tail Plate Effect on the wake

- Acts like backward-facing step
- Flow reattaches at ~6 step heights
- If plate ends near reattachment, wake closes due to fluid momentum toward model centerline
- Full-scale data indicates best drag reduction for plates 5-6 step heights in length

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# PIV Data Correction

- Errors in reported velocity measurements identified
  - Seems to be a problem with $\Delta t$ so it is an incorrect scaling, not an offset
  - Free stream ~10% off for horizontal and streamwise planes - up to 25% for cross-stream planes
- Data has been re-reduced to report 3 velocity components normalized by "free stream"
  - Location of free-stream identified for each measurement plane
  - Comparisons with CFD still possible if similarly normalized
  - Will distribute normalized data on CD



# Generic Conventional Model (GCM) Truck Test in 7x10 and12-Ft.

Dale Satran
dsatran@mail.arc.nasa.gov
650-604-5879

# Deliverables

- Digitized model geometry
- CFD validation data
- Reynolds Number effects
- Drag reduction
- PIV data
- Final reports

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# Actions

- Digitize model
- Analyze 7 x 10 results
- Modify model based on 7 x 10 results
- Modify model for mounting in 12-Ft.
- Restore instrumentation
- Conduct test
- Analyze results
- Prepare final report

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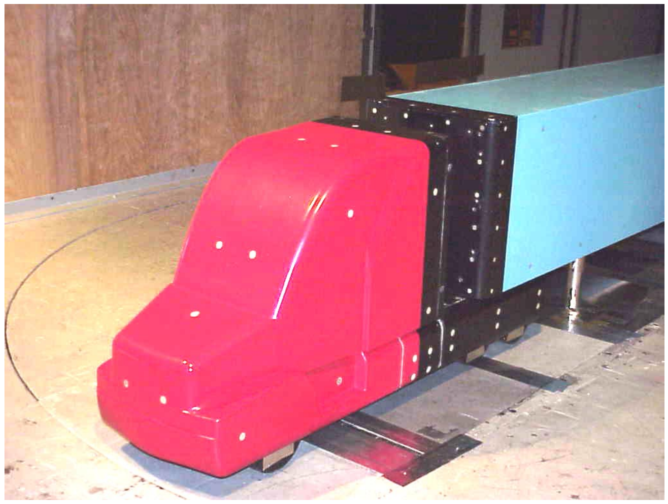
# Basic Model



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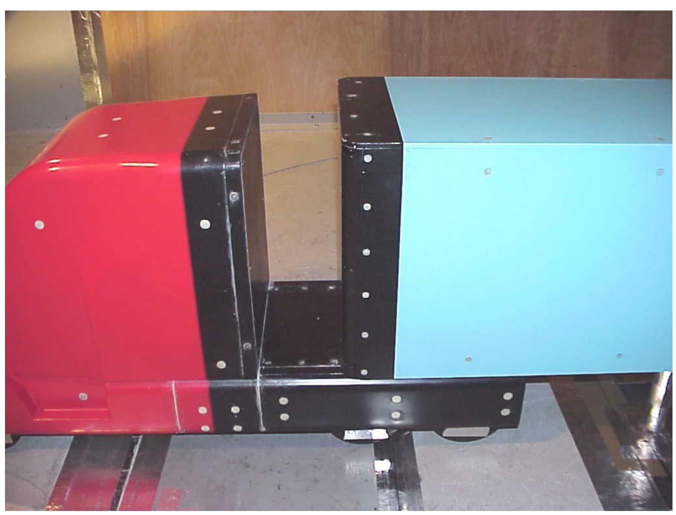
# Basic Model - Gap



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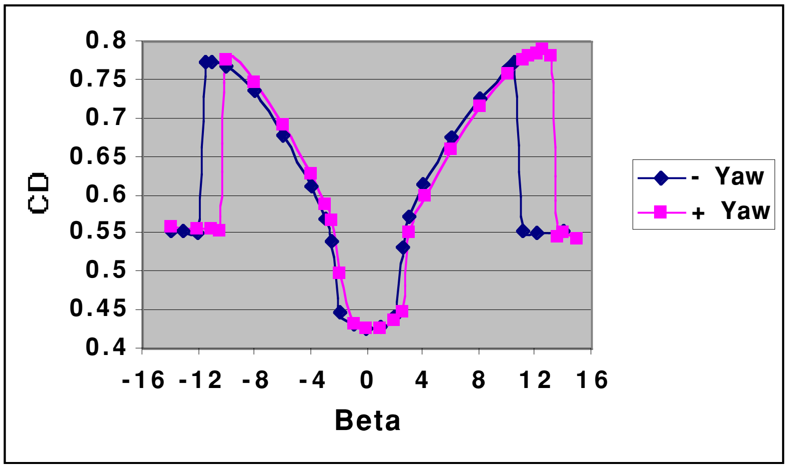
# Basic Model - Hysteresis

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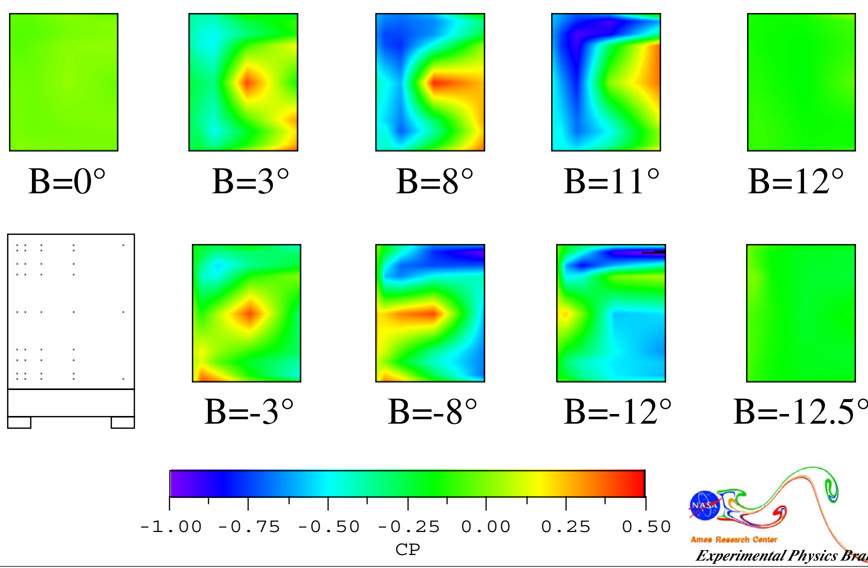
# Basic Model
# Tractor Rear Pressures

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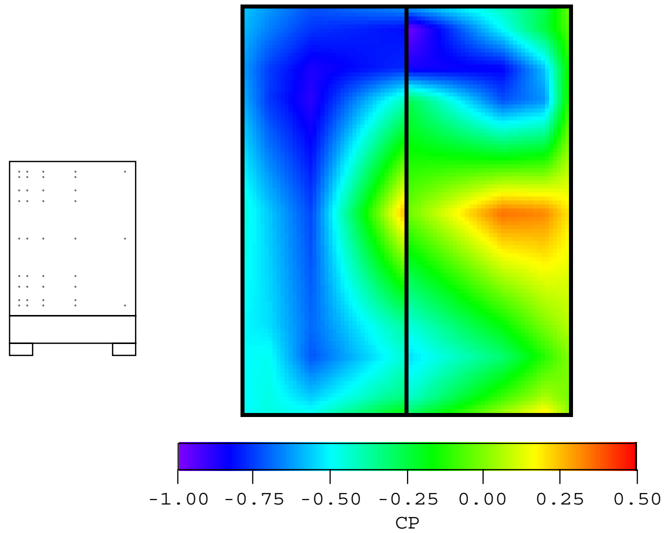
# Composite TractorPressures for Beta=10°

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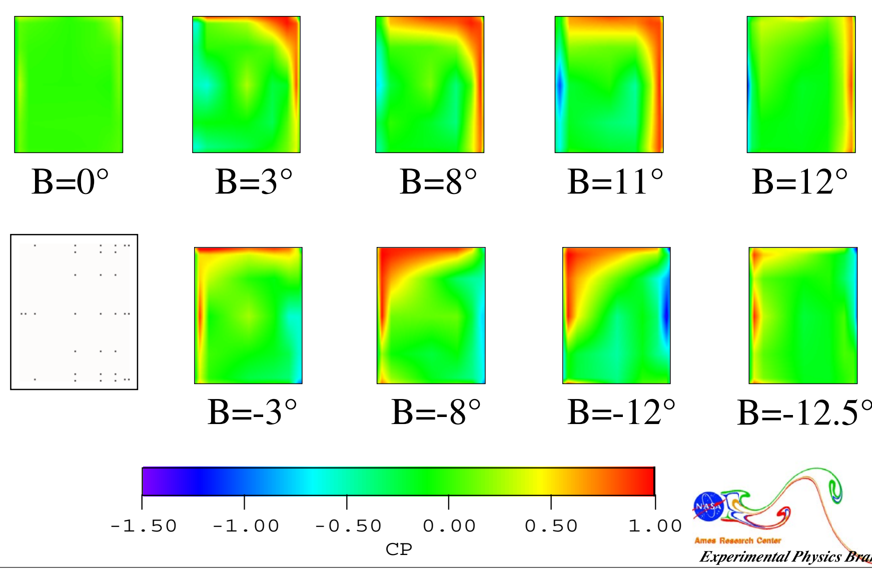
# Basic Model Trailer Front Pressures

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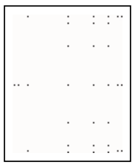
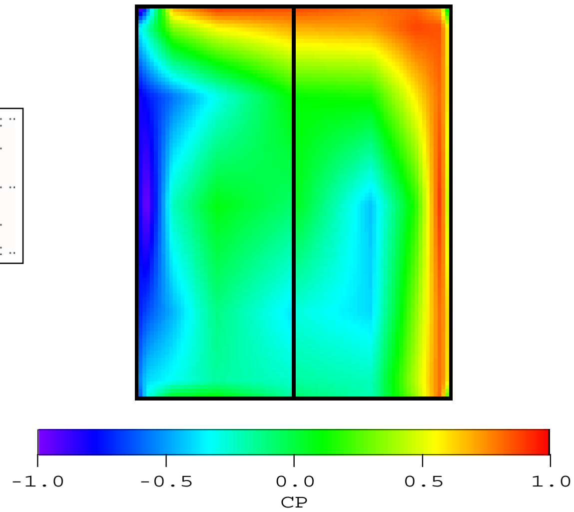
# Composite TrailerPressures for Beta=10°



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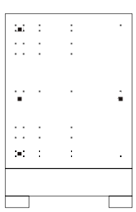
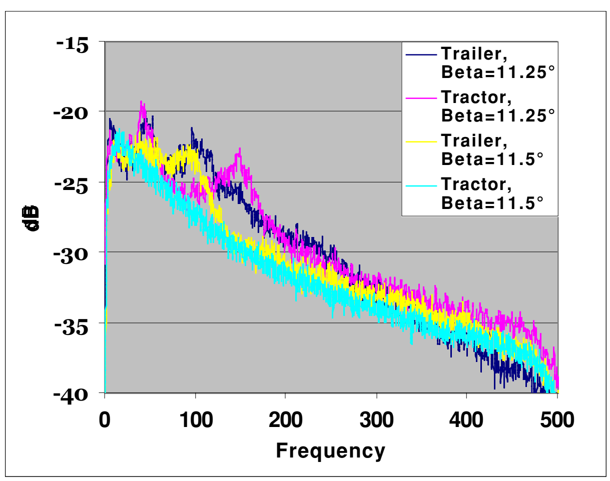
# Basic Model - Spectra Gap Data



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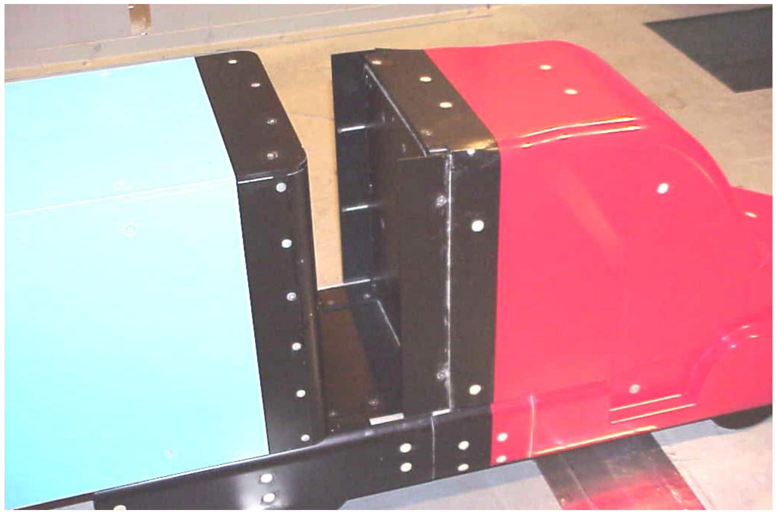
# Side Extenders



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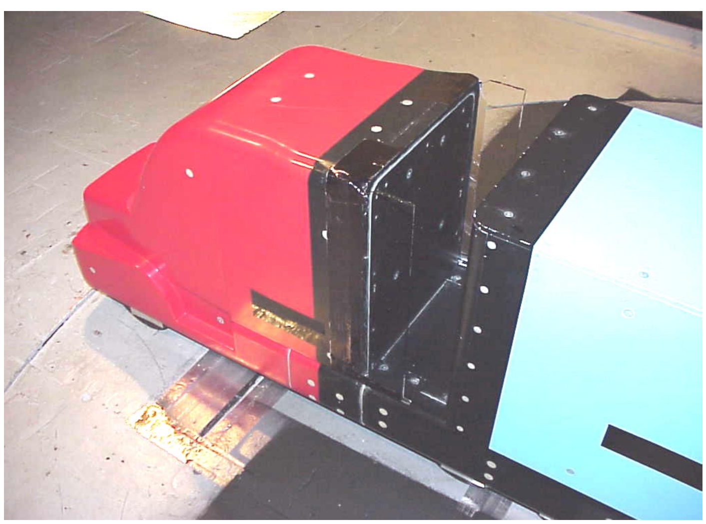
# Glass Side Extenders



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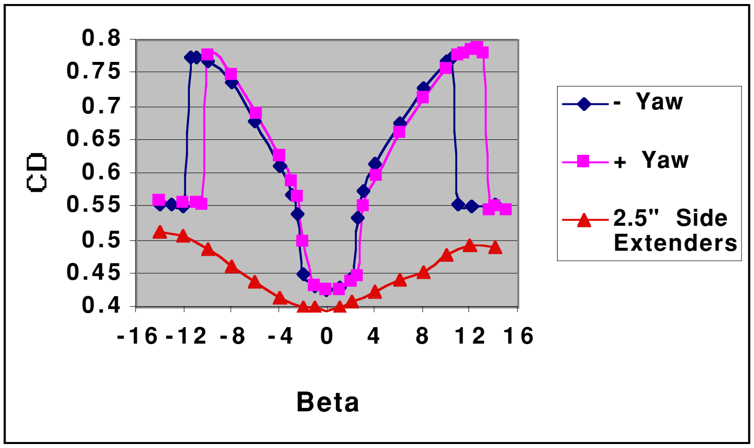
# Side Extenders

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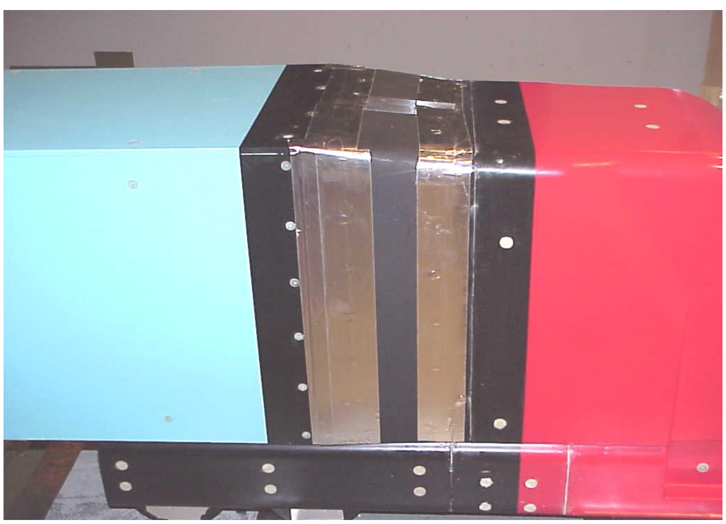
# Filled Gap



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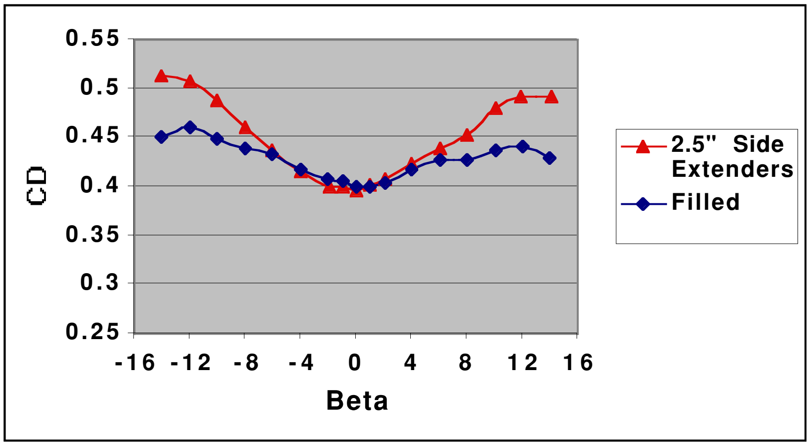
# Filled Gap

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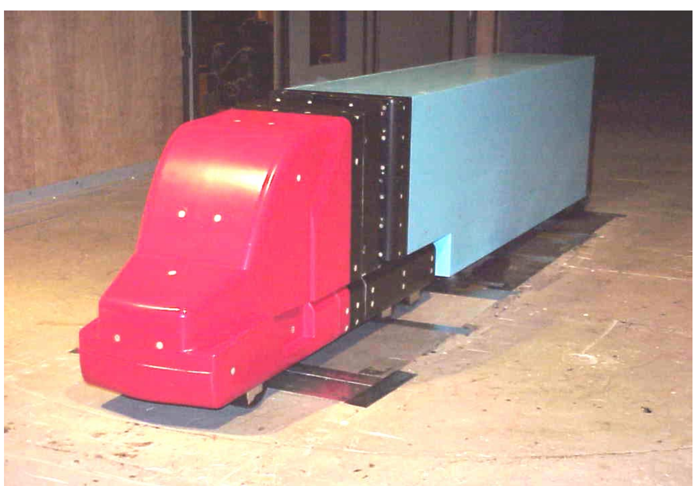
# Lowboy Trailer



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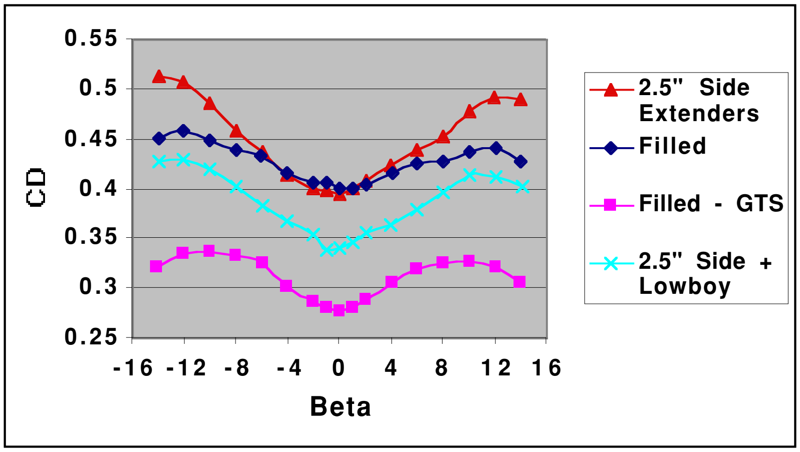
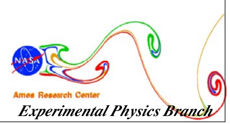
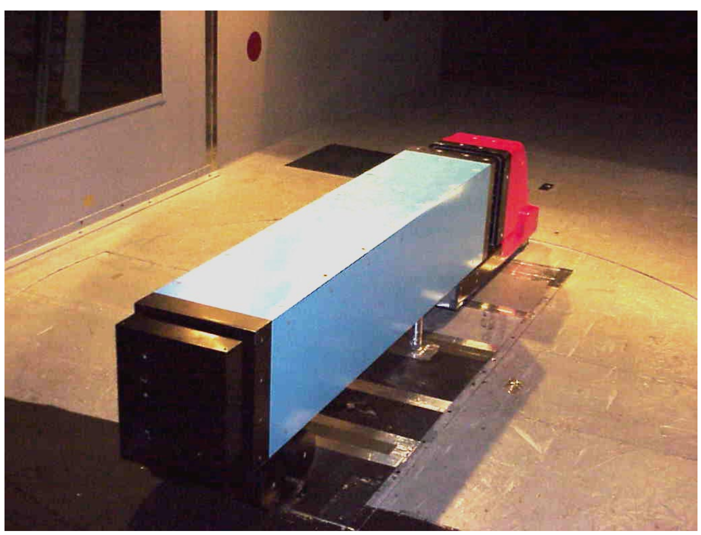
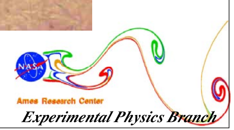
# Lowboy Trailer



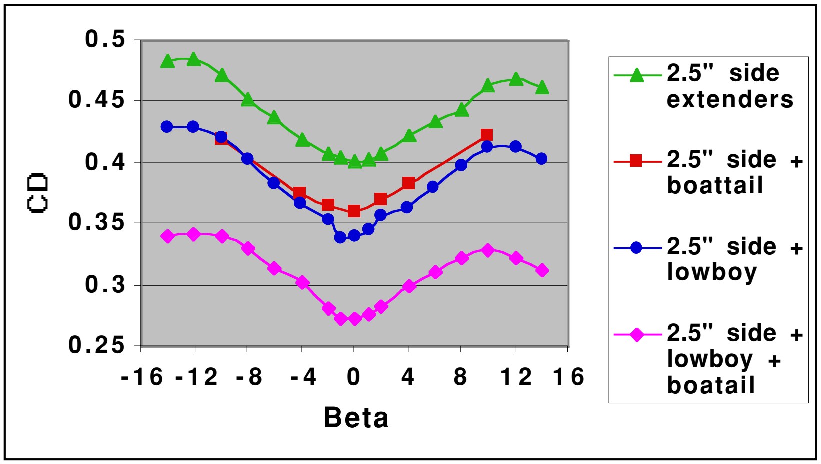




# Boattail





# Boattail

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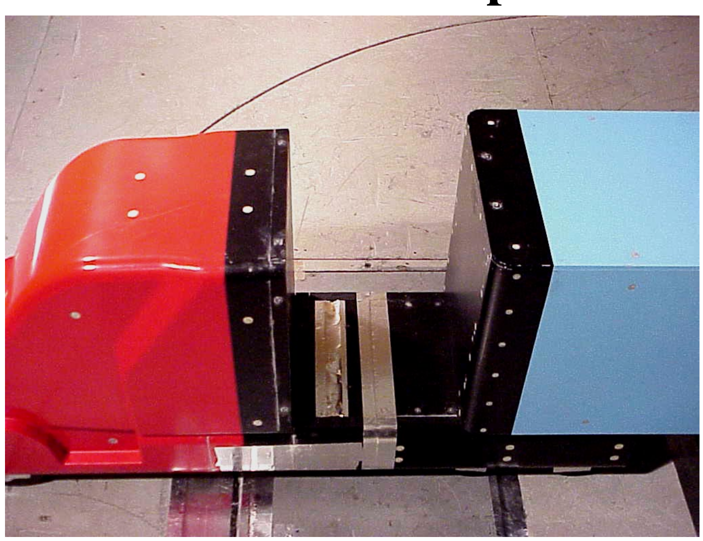
# Extended Gap



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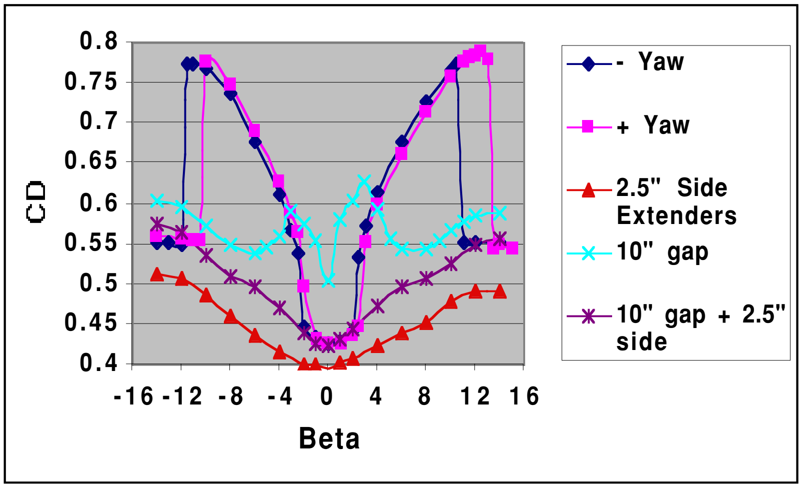
# Extended Gap

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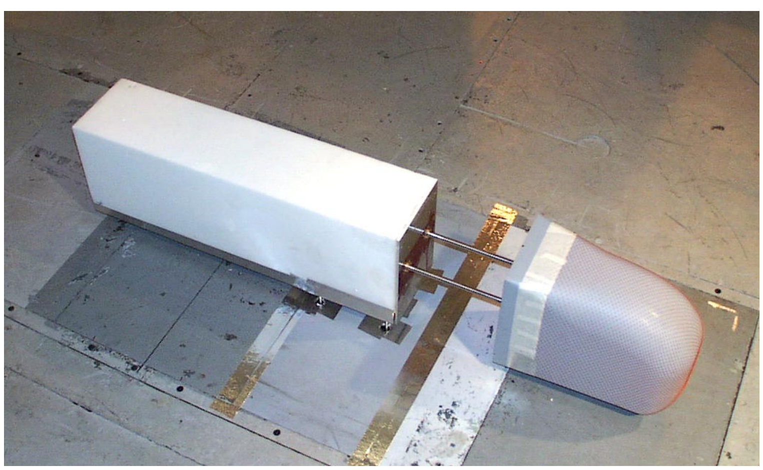
# USC Model



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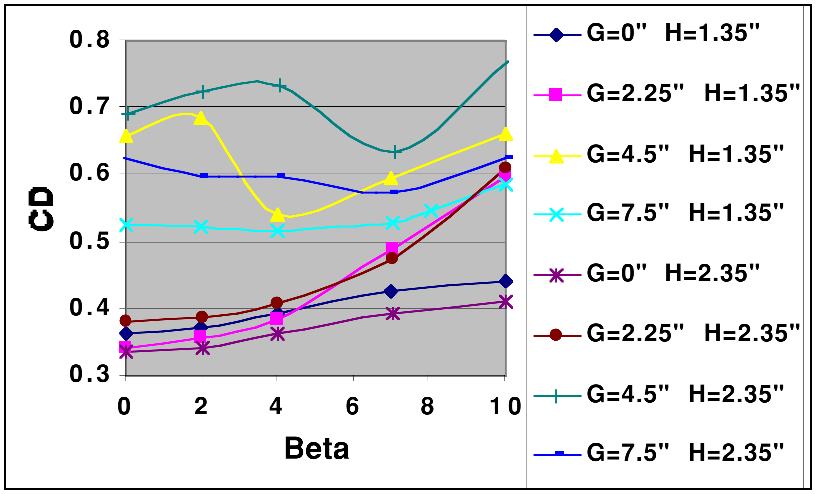
# USC Gap and Height

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# Three-component PIV of Generic Conventional Model (GCM) Truck Test in 7x10 and12-Ft.


JT Heineck, Stephen Walker
jheineck@mail.arc.nasa.gov
650-604-0868



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## Summary of PIV Efforts

1998: Army/NASA 7x10

- GCM Wake flow, with and without boattail device, 0 and 10 deg, 7 planes, 3 Reynolds conditions

2001: Army/NASA 7x10

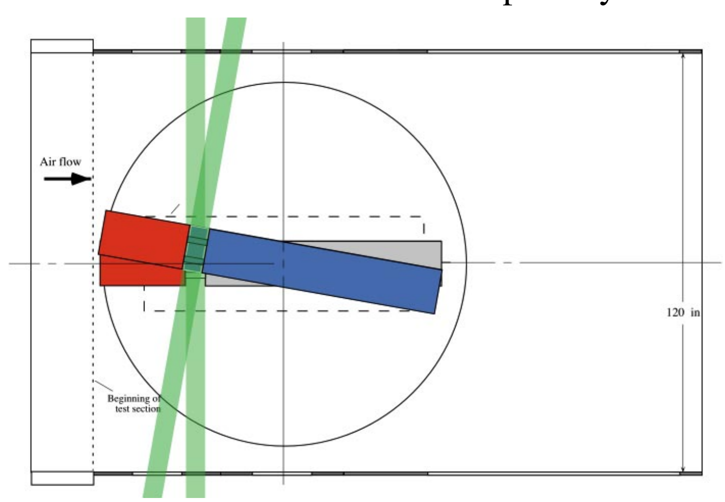
- GCM Gap flow, with and without side extenders, 0 and 10 deg yaw, 3 planes
- GCM Wake flow, with and without boattail device, 0 and 10 deg, 3 planes
- 1 Reynolds Condition

2002: NASA 12-foot Pressure Wind Tunnel

- GCM Gap flow, with and without side extenders, 0, 5 and 10 deg yaw, 3 planes
- GCM Wake flow, with and without boattail device, 0 and 10 deg, 3 planes
- 2 Reynolds Conditions



## PIV in 7 x 10 of GCM – Gap Study

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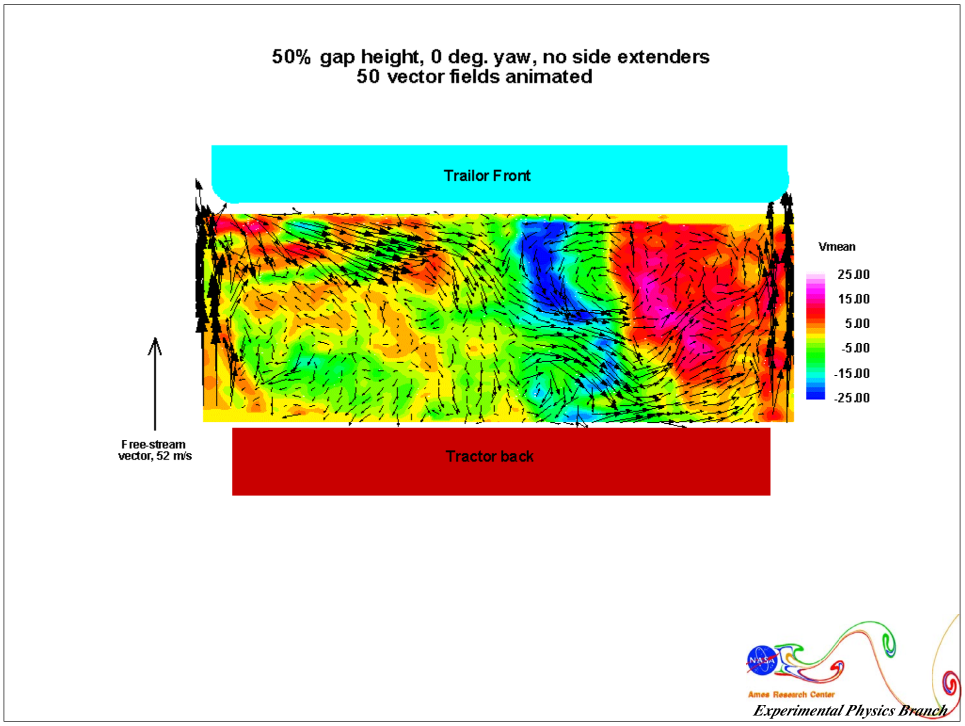
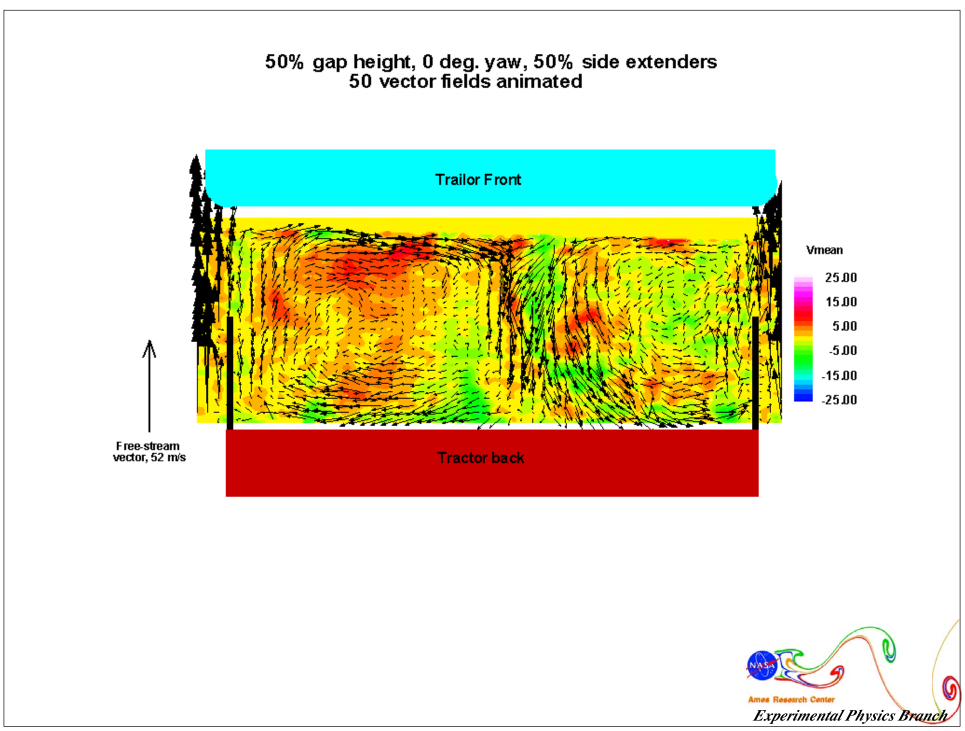

50% gap height, 0 deg. yaw, no side extenders
50 vector fields animated
Trailor Front
Tractor back
Free-stream vector, 52 m/s
Vmean
25.00
15.00
5.00
-5.00
-15.00
-25.00
Ames Research Center
Experimental Physics Branch


50% gap height, 0 deg. yaw, 50% side extenders
50 vector fields animated
Trailor Front
Tractor back
Free-stream vector, 52 m/s
Vmean
25.00
15.00
5.00
-5.00
-15.00
-25.00
Ames Research Center
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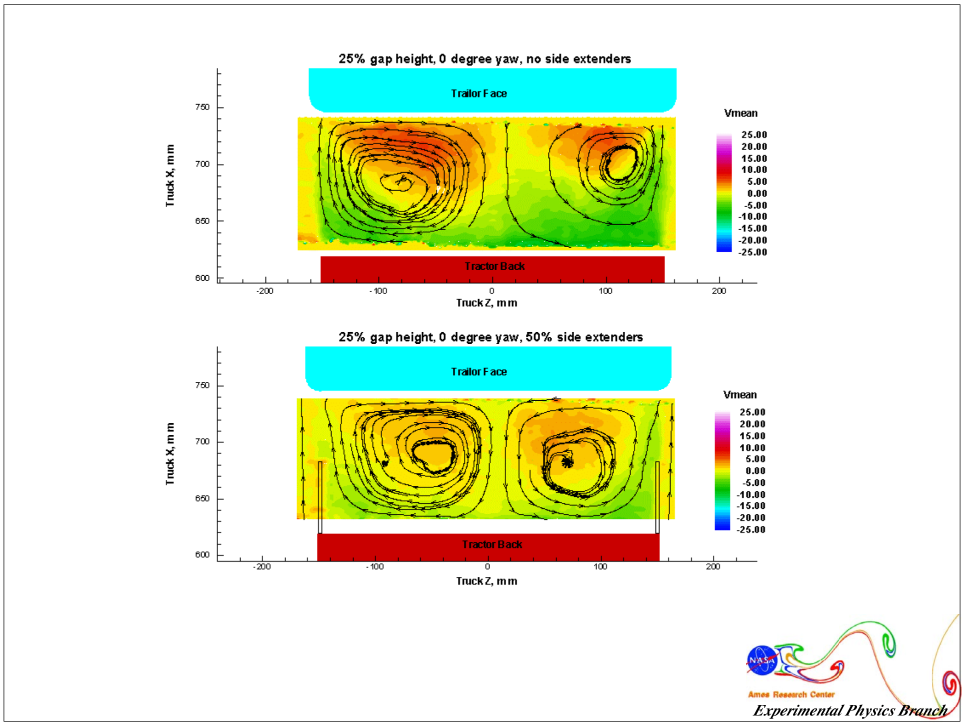
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Trailor Face
Tractor Back
Vmean
Truck X, mm
Truck Z, mm
25% gap height, 0 degree yaw, 50% side extenders
Ames Research Center
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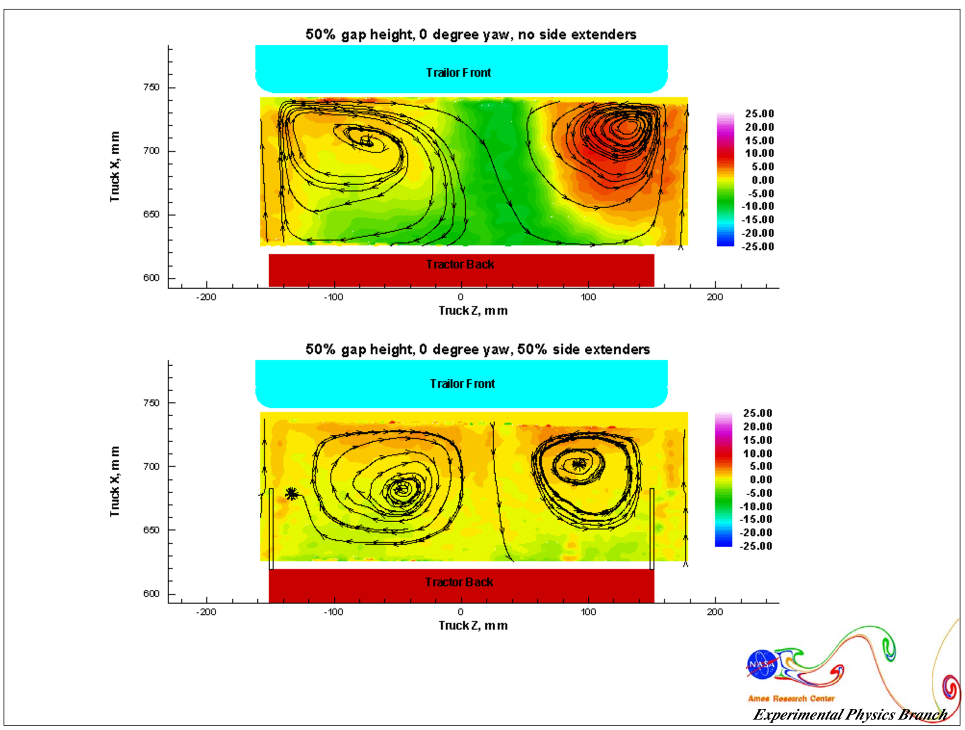
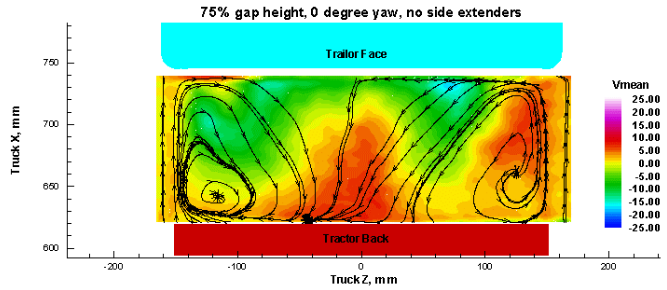
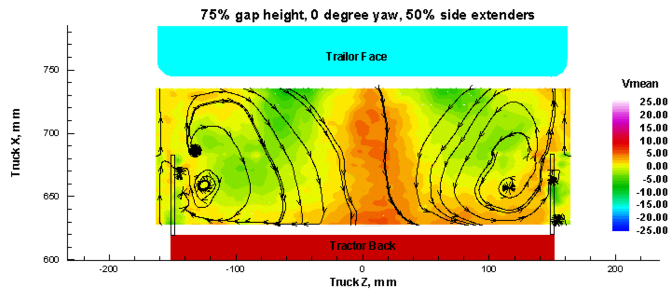

50% gap height, 0 degree yaw, no side extenders
Trailor Front
Tractor Back
Truck X, mm
Truck Z, mm
50% gap height, 0 degree yaw, 50% side extenders
Ames Research Center
Experimental Physics Branch

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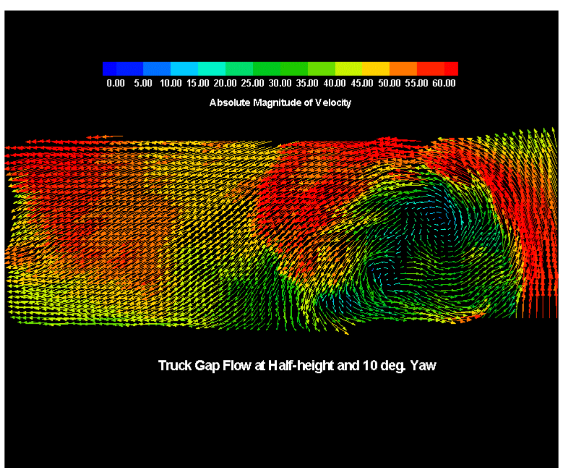
# Animation of 100 Vector Fields, 50% Height in Gap, 10 deg Yaw

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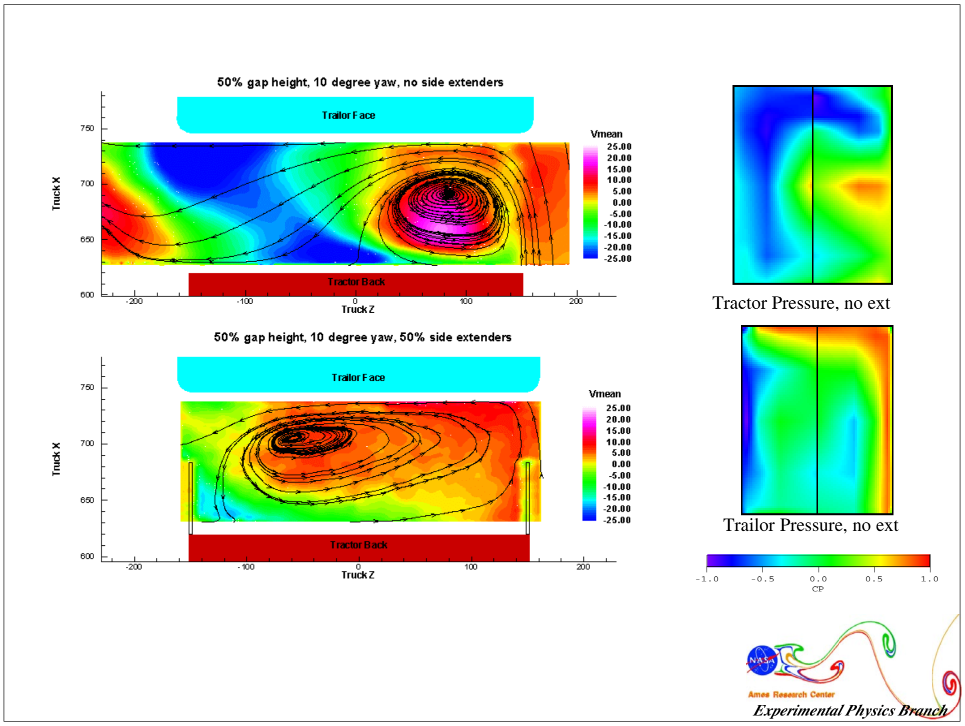
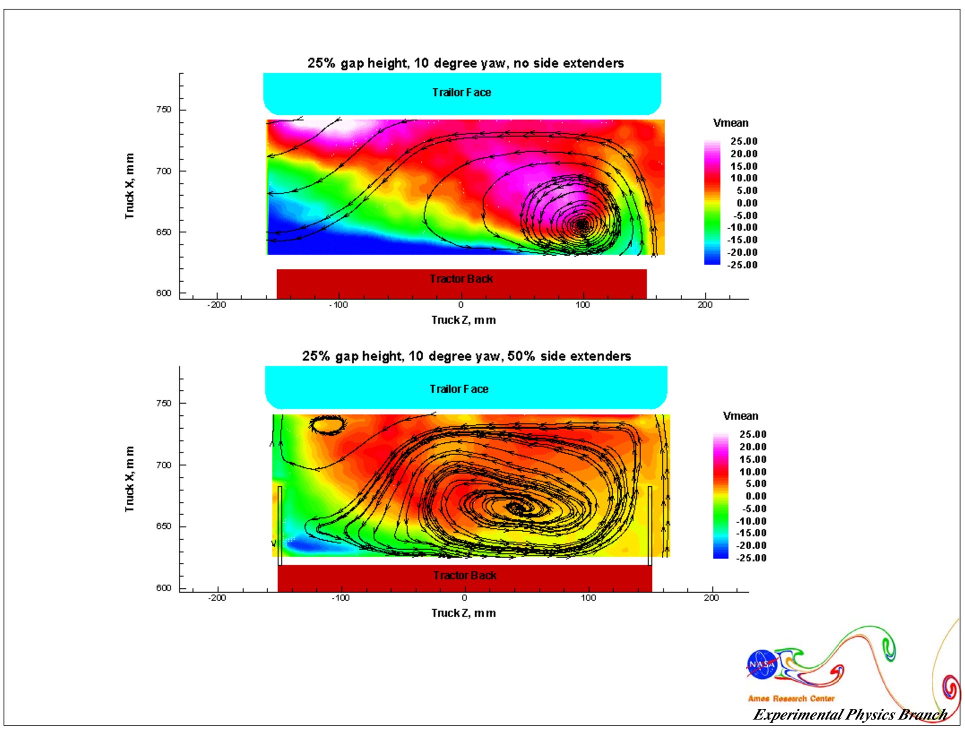

50% gap height, 10 degree yaw, no side extenders
Trailor Face
Tractor Back
Truck X
Truck Z
Vmean
Tractor Pressure, no ext
50% gap height, 10 degree yaw, 50% side extenders
Trailor Pressure, no ext
CP
Ames Research Center
Experimental Physics Branch


25% gap height, 10 degree yaw, no side extenders
Trailor Face
Tractor Back
Truck X, mm
Truck Z, mm
Vmean
25% gap height, 10 degree yaw, 50% side extenders
Ames Research Center
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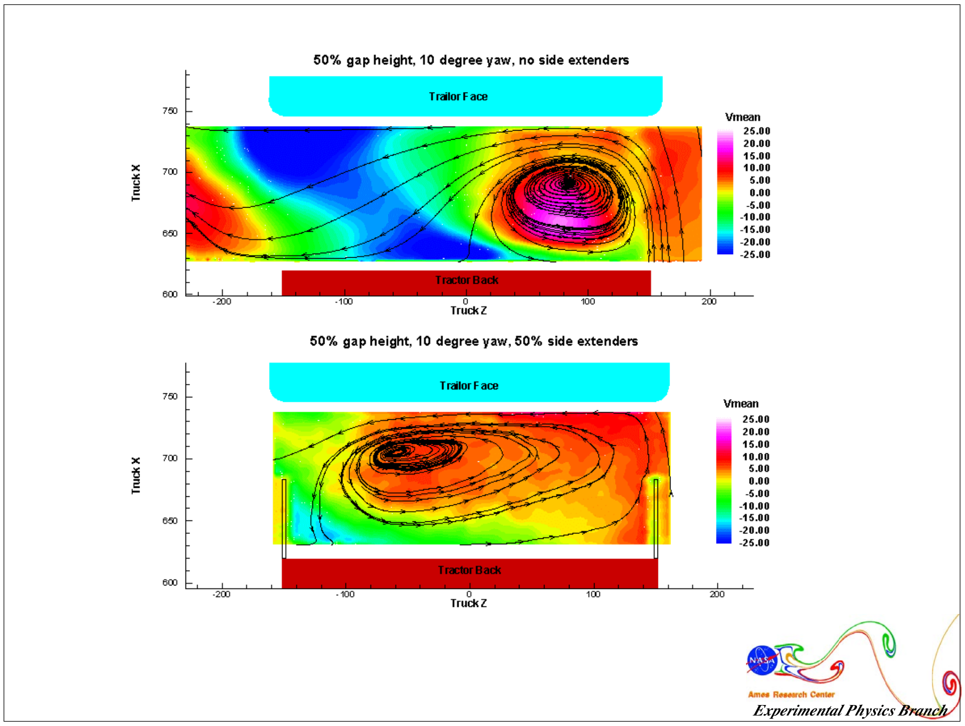

50% gap height, 10 degree yaw, no side extenders
Trailor Face
Tractor Back
Truck X
Truck Z
Vmean
50% gap height, 10 degree yaw, 50% side extenders
Trailor Face
Tractor Back
Truck X
Truck Z
Vmean
Ames Research Center
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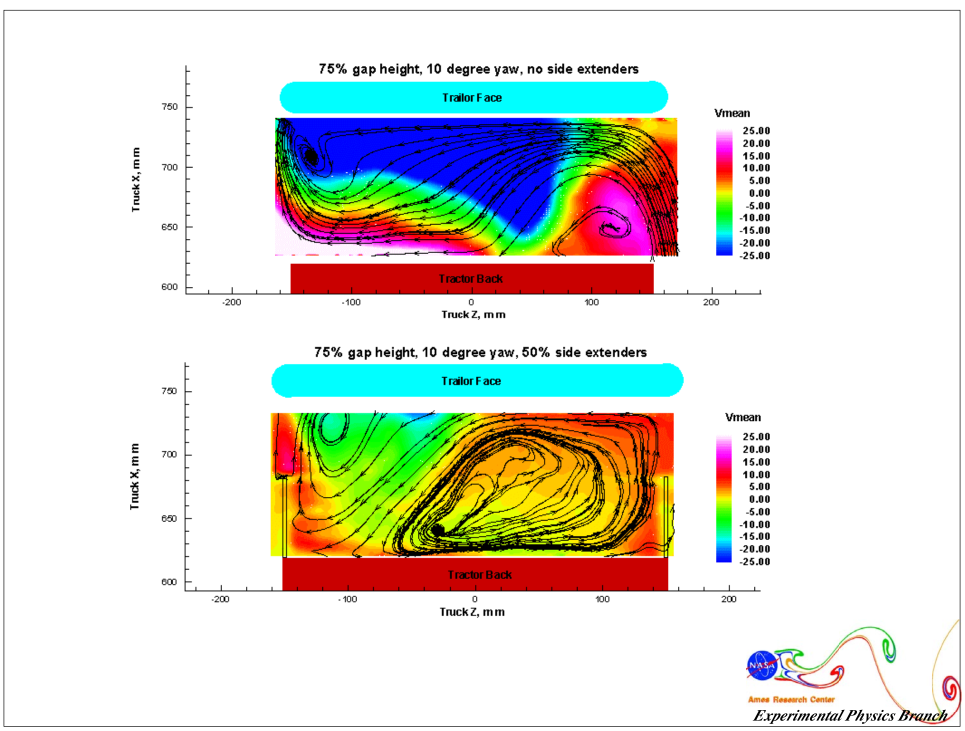
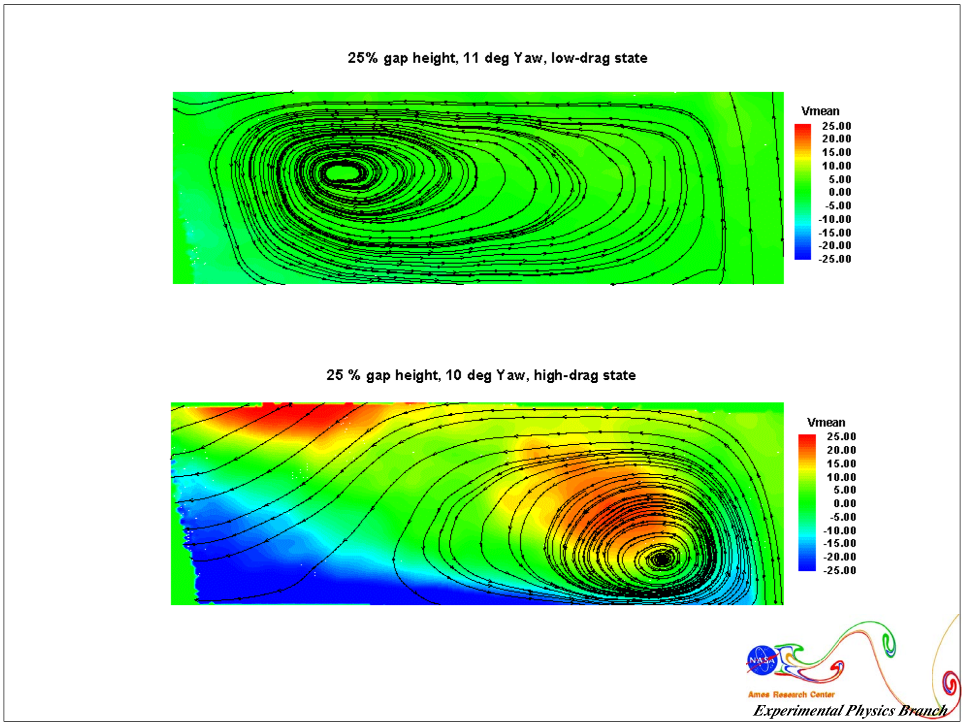
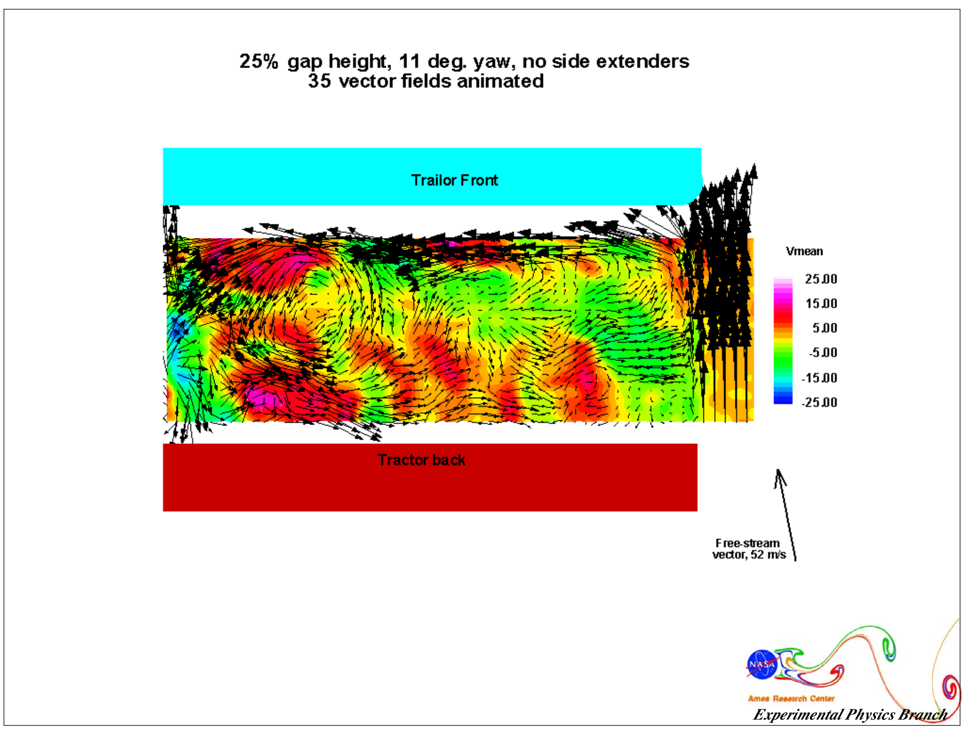
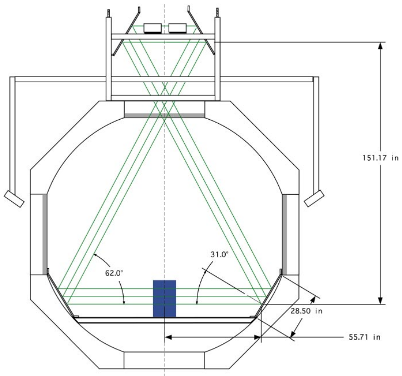


75% gap height, 10 degree yaw, no side extenders
Trailor Face
Tractor Back
Truck X, mm
Truck Z, mm
Vmean
75% gap height, 10 degree yaw, 50% side extenders
Trailor Face
Tractor Back
Truck X, mm
Truck Z, mm
Vmean
Ames Research Center
Experimental Physics Branch

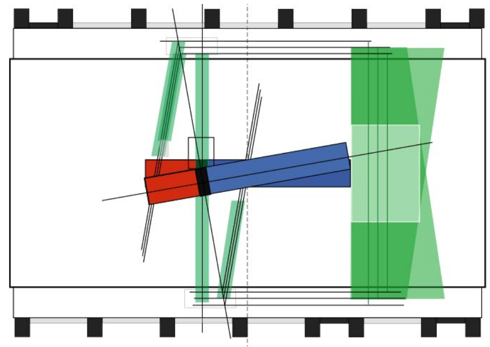

25% gap height, 11 deg Yaw, low-drag state
Vmean
25.00
20.00
15.00
10.00
5.00
0.00
-5.00
-10.00
-15.00
-20.00
-25.00
25 % gap height, 10 deg Yaw, high-drag state
Vmean
25.00
20.00
15.00
10.00
5.00
0.00
-5.00
-10.00
-15.00
-20.00
-25.00
NASA
Ames Research Center
Experimental Physics Branch


25% gap height, 11 deg. yaw, no side extenders
35 vector fields animated
Trailor Front
Vmean
25.00
15.00
5.00
-5.00
-15.00
-25.00
Tractor back
Free-stream
vector, 52 m/s
NASA
Ames Research Center
Experimental Physics Branch

# PIV at the 12 foot Pressure Tunnel




Upwind view



Top view

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Experimental Physics Branch

## USC Presentation for
## DOE Office of Transportation Technology
## Office of Heavy Vehicle Technology


M. Hammache, staff
T.Y. Hsu, staff
D. Arcas, PhD student
D. Monnesinghe, MS student
D. Lazzara, student
C. Radovich, student
R. Blackwelder, staff
F. Browand, staff
P. Lissaman, staff


**Ground Vehicle Aerodynamics Laboratory**

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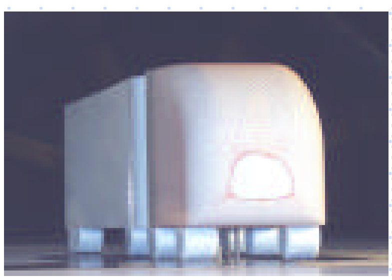
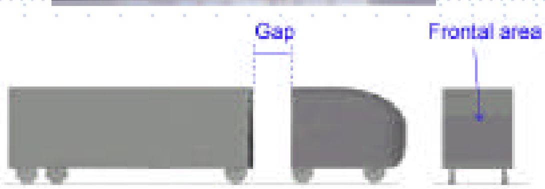
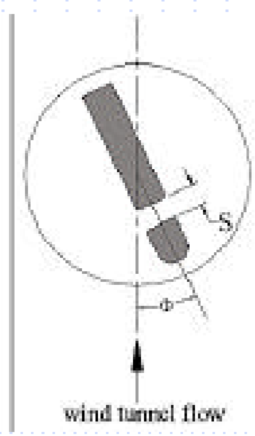
## Aerodynamic Forces on Truck Models, Including Two Trucks in Tandem

Mustapha Hammache, Mark Michaelian, Fred Browand,
SAE paper No. 2002-01-0530 (Force data for tractor-trailer available on CD)





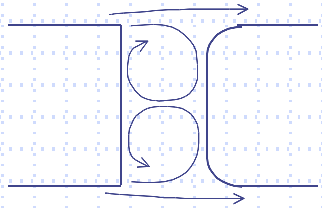



## The Structure of Turbulent flow in the Gap Between Tractor and Trailer

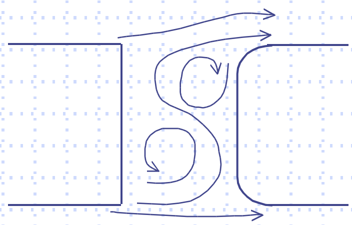
Mustapha Hammache, Fred Browand

# Flow Patterns in the Gap

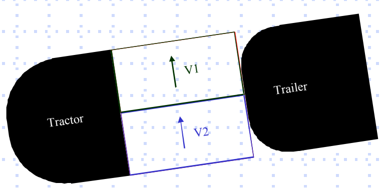
- Small gap

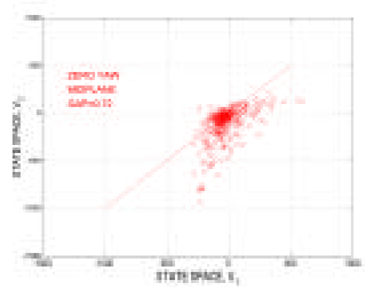


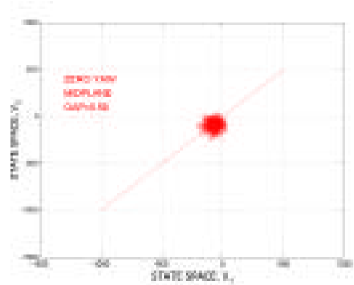
- Critical gap

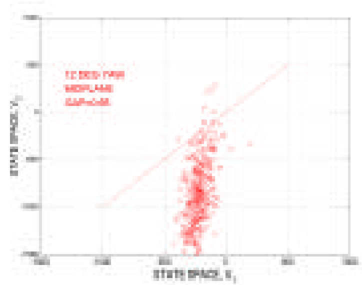


The Structure of Turbulent flow in the Gap Between Tractor and Trailer
Mustapha Hammache, Fred Browand



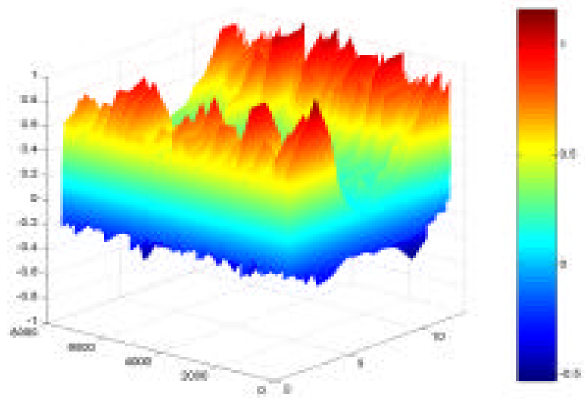







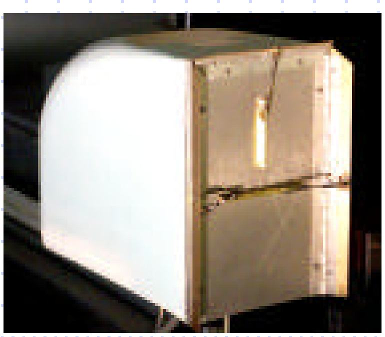
## Instantaneous Pressure Measurements of Turbulent Flow in the Gap of a Tractor-Trailer Vehicle

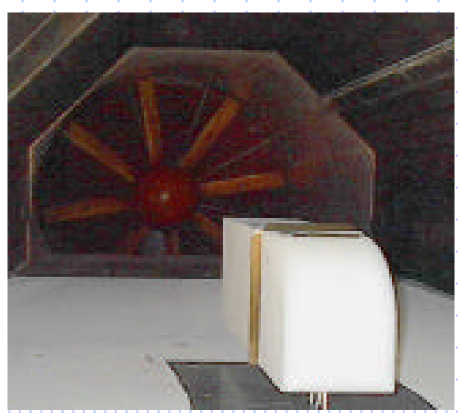
David Lazzara, Submitted, AIAA Student Paper Competition, San Luis Obispo, April 2002



## Effect of Cab Extender Geometries on the Drag of a Model Tractor-Trailer

Devinda Moonesinghe, Charles Radovitch





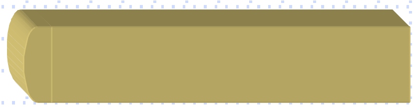
## The Limits of Drag Behavior for Two Bluff Bodies in Tandem

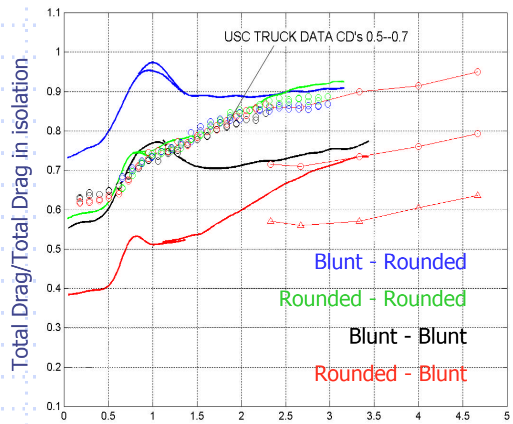
Fred Browand & Mustapha Hammache,

14th USNCTAM, The Roshko symposium on Turbulent Structure and Flow Control, June 23-25, Blacksburg, VA

Blunt

Rounded



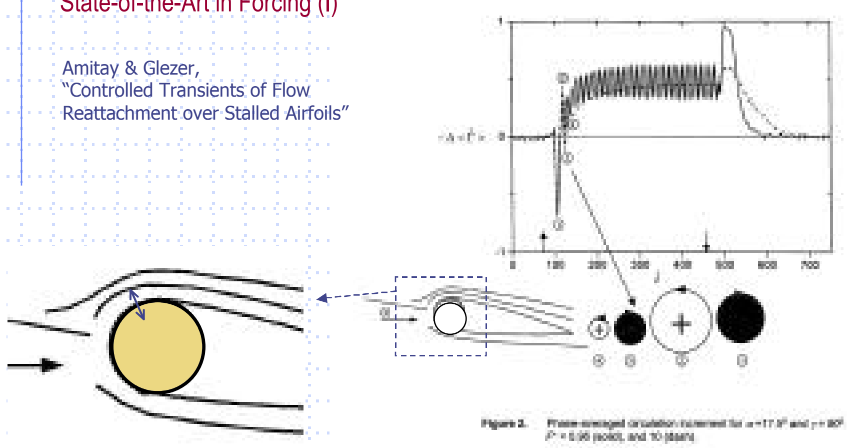



## Base Geometry Modifications and Acoustic Forcing to Reduce Drag

Tsun-Ya Hsu, Mustapha Hammache

### State-of-the-Art in Forcing (I)

Amitay & Glezer,
"Controlled Transients of Flow Reattachment over Stalled Airfoils"



## Base Geometry Modifications and Acoustic Forcing to Reduce Drag

Tsun-Ya Hsu, Mustapha Hammache

### State-of-the-Art in Forcing (II)

Nishri & Wygnanski,
"Effects of Periodic Excitation on Turbulent Flow Separation from a Flap"

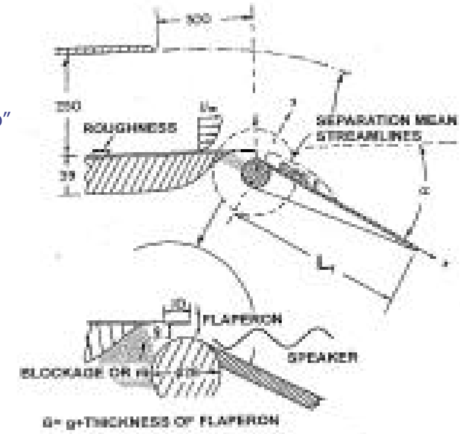
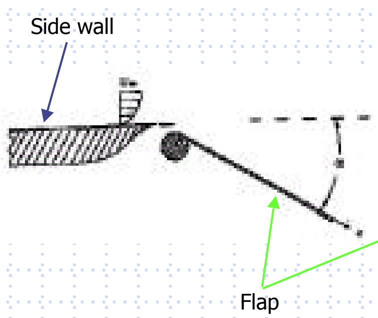


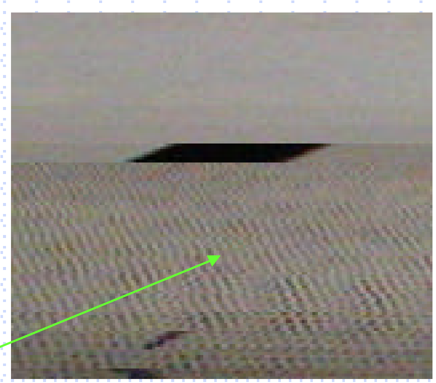

Fig. 1 Sketch of the flap and the region around the flap shoulder.

## Base Geometry Modifications and Acoustic Forcing to Reduce Drag

Tsun-Ya Hsu, Mustapha Hammache

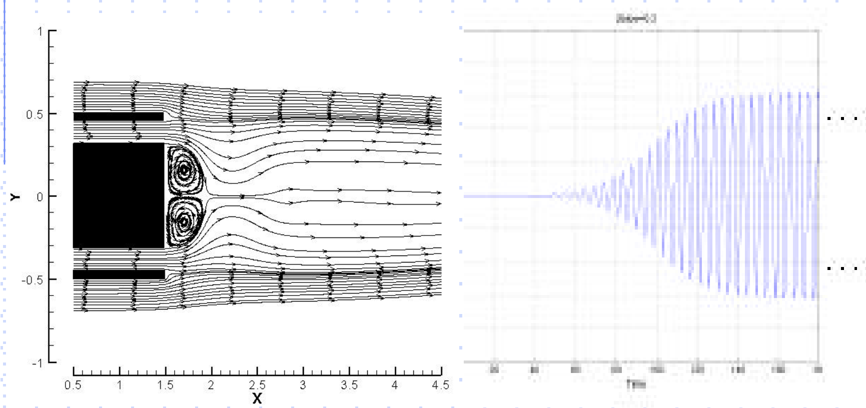
### Current Forcing Design at USC



## 2-D Numerical Models of the Base Flow Region Subjected to Modifications in Geometry or Small Addition or Removal of Mass

Diego Arcas

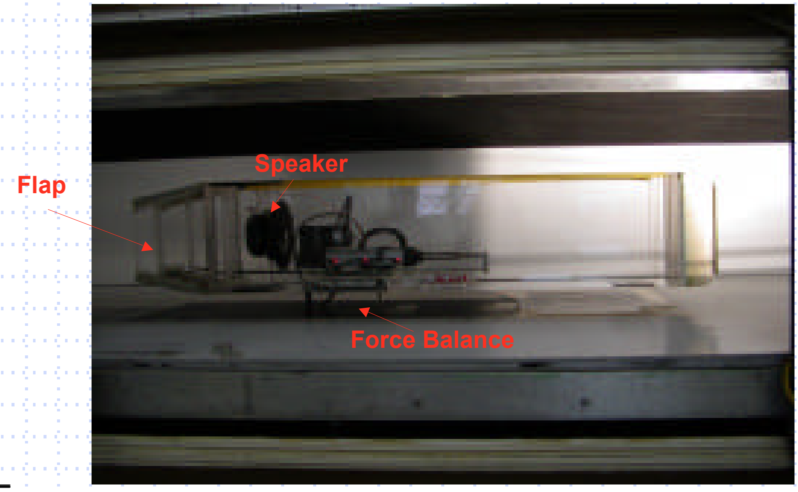



# Contents

- Experimental Apparatus
- Experimental Conditions
- Results
- Summary & Future Study

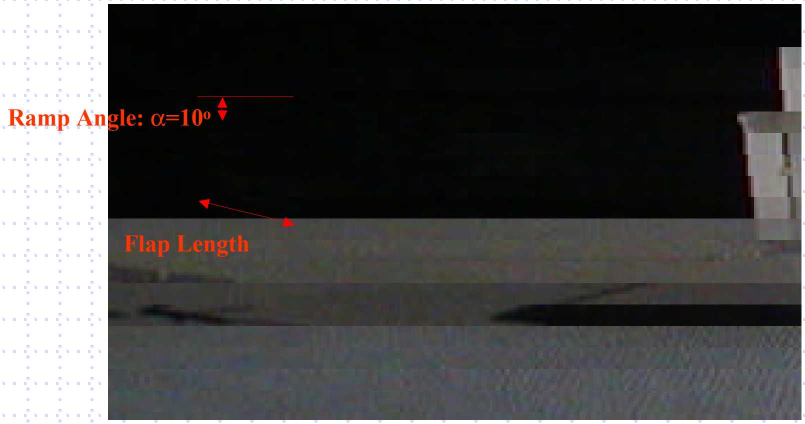
USC GROUND VEHICLE AERODYNAMICS LABORATORY

## Experimental Apparatus






## Experimental Apparatus (Cont'd)



## Experimental Apparatus (Cont’d)




## Experimental Details

- Free Stream Velocity, U = 13 to 20 m/s
- $A = 0.0535\ m^2$
- $Re_{sqrt(A)} = 2.8 \times 10^5$ to $3.2 \times 10^5$
- Flap lengths: 14 to 24 cm
- Ramp angles: 0°, 5°, 10°
- Square wave with frequency, f = 60 to 120 Hz
- Gap width for the jet, g = 1 mm

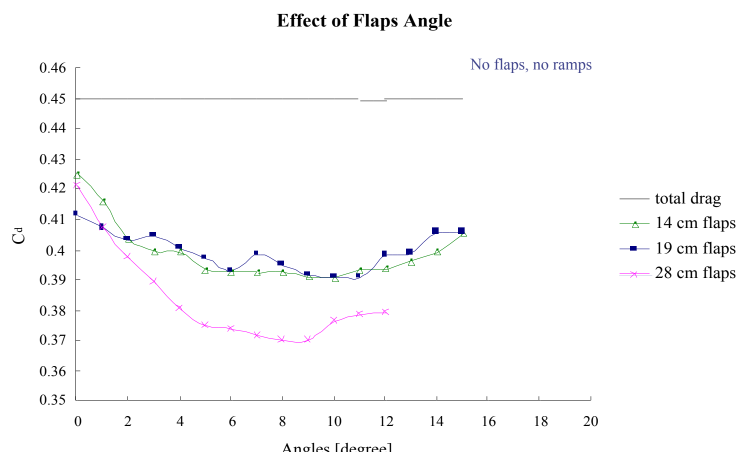
# Experimental Results

- Without Forcing:
  - Drag measurements for varying flap angles
  - Effect of flap lengths on drag coefficients
- With Forcing:
  - Drag measurements for varying forcing frequency



# Without Forcing

## Effect of Flaps Angle on $C_d$: Flaps without Ramps

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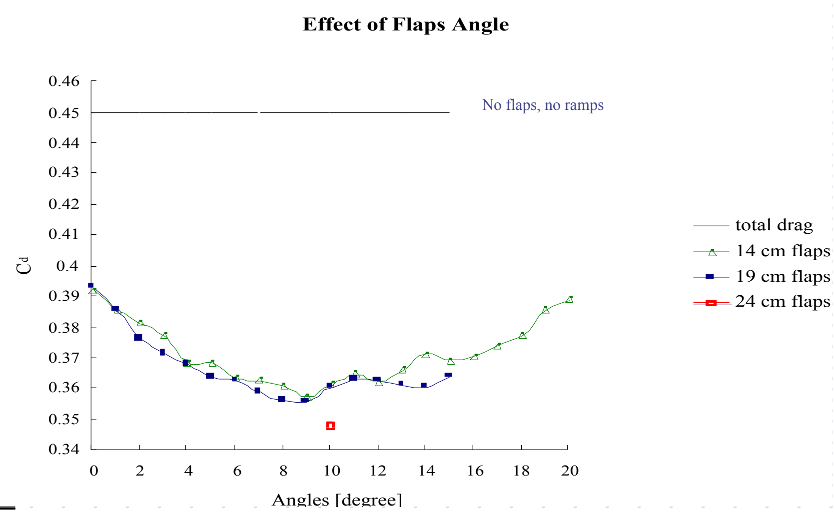
## Effect of Flaps Angle on $C_d$: Flaps with $10^o$ Ramps

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## With Forcing

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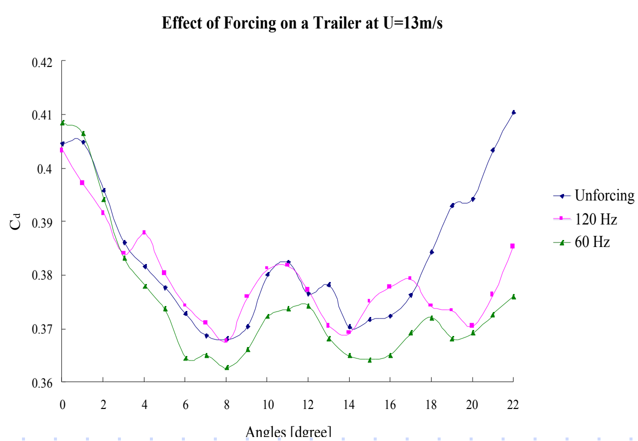
## Effect of Forcing Function on $C_d$: 19 cm Flaps with 15° Ramps , U = 13 m/s

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## Summary & Present Study Continued

- Without forcing: 20% saving based on total drag
- Forcing has effect on drag reduction
- Utilizing DPIV technique to further understand the flow characteristics at flap angle around 10 degrees.
- Develop complex waveforms as a forcing function to decrease drag.
- Use experimental results to develop an enhanced 3D model.

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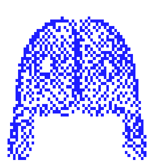
# Flow Structure and Drag Reduction in 2-D Wakes with Boat-Tails



A Direct Numerical Simulation of the Basic Flow

D. R. Arcas, F. K. Browand. and L. G.Redekopp

Dept. of Aerospace Engineering

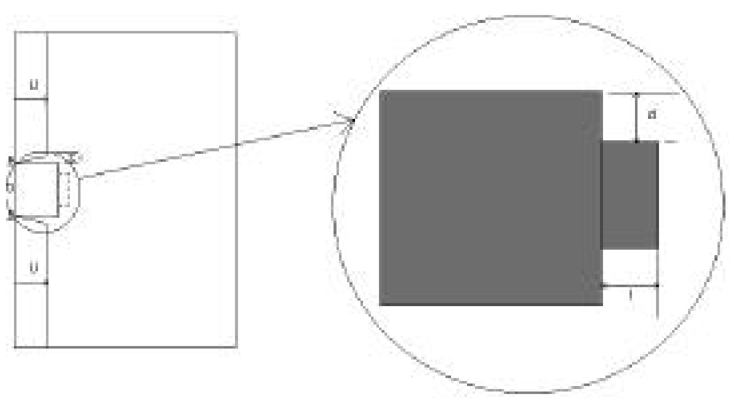
University of Southern California

# Objectives:

- To reach an understanding of the basic flow dynamics associated with geometric configurations of minimum drag.
  - Identification of the minimum drag configurations by means of a parametric study.
  - Study of the velocity and pressure fields.
  - Study the possibility of using suction/blowing for drag reduction purposes.

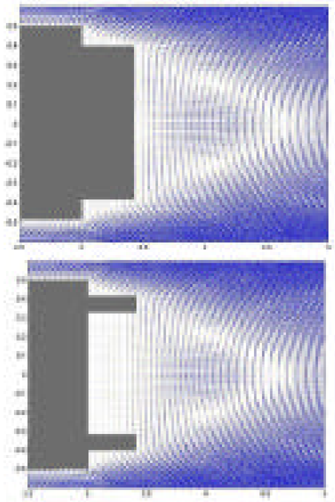
## Bluff-body Wake Geometry

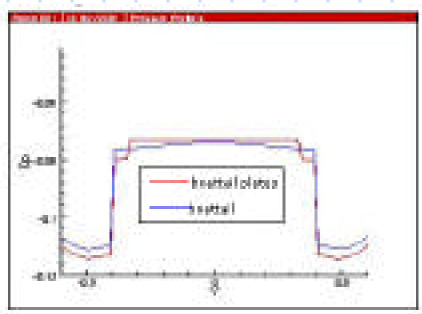
Parameters: $\mathrm{Re} = \frac{U_\infty H}{\nu}$ d, l, δ



# Bluff Body Wake:

-Boattail Configuration



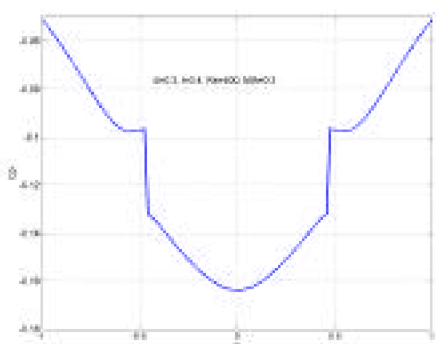


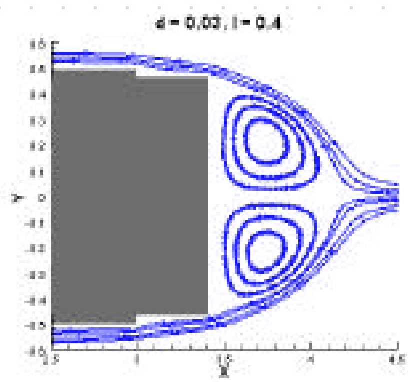

Discrepancies between the *boattail* and the *boattail plates* configurations are minimal in this regime of flow.

# d=0.03, l=0.4

Pressure Profile

Streamline Pattern

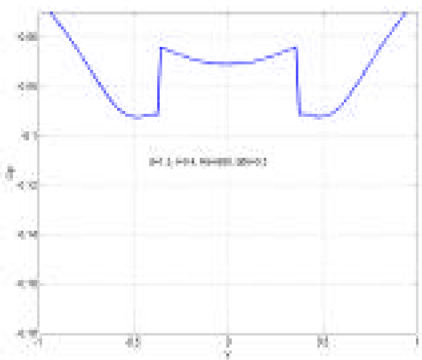


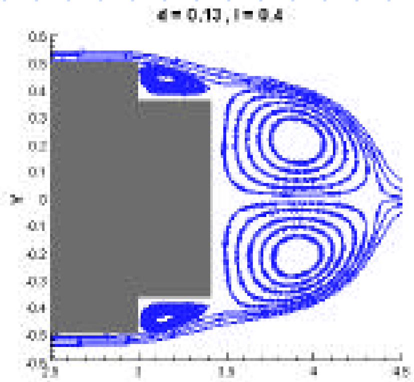


# d=0.13, l=0.4 (Optimum case)

Pressure Profile

Streamline Pattern

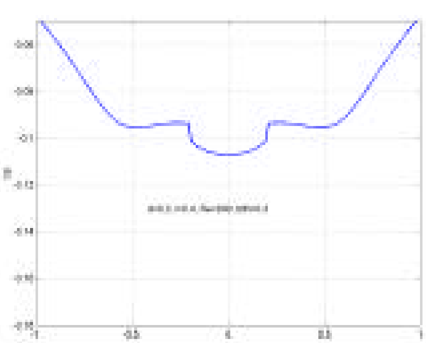


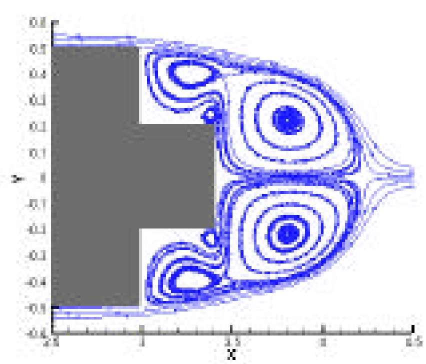



# d=0.3, l=0.4

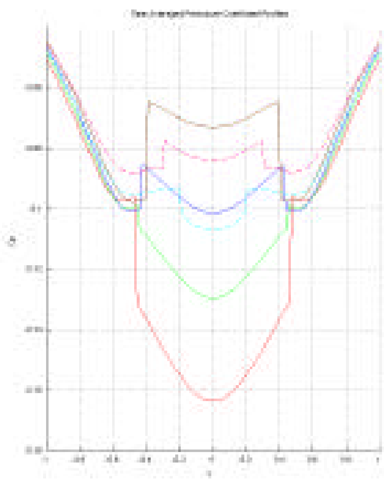
Pressure Profile

Streamline Pattern

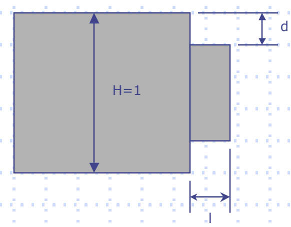




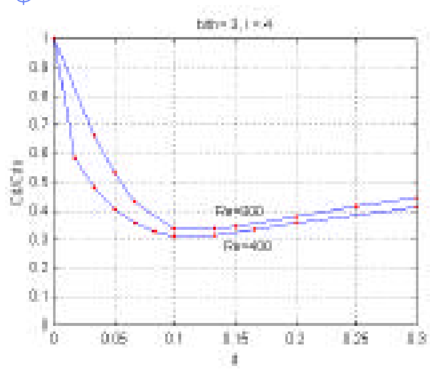
## Pressure Profiles:

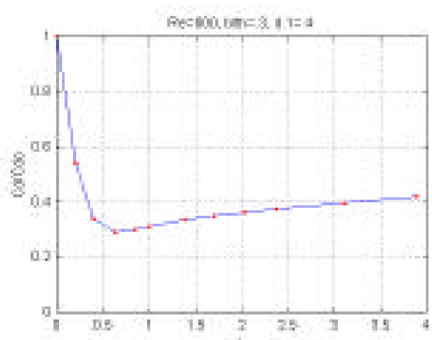


| Rising Pressure | Dropping Pressure |
|---|---|
| d=.03 | d=.10 |
| d=.05 | d=.20 |
| d=.06 | d=.30 |
| d=.10 | |




## Influence of Different Parameters Drag Reduction



Minimum Value of Drag with, d
$d_{min}$~0.12

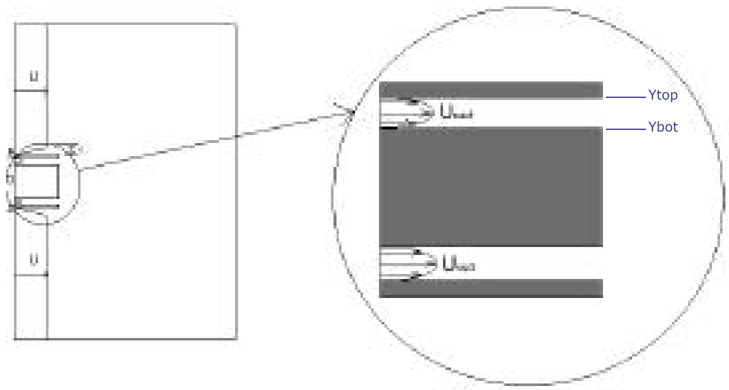
Minimum Value of Drag with , l
$l_{min}$~0.65

$(l/d)_{empirical}$~6

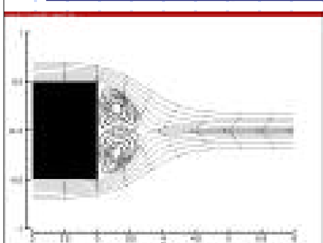
## Bluff-body Wake Geometry

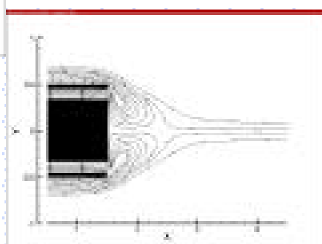
Parameters:

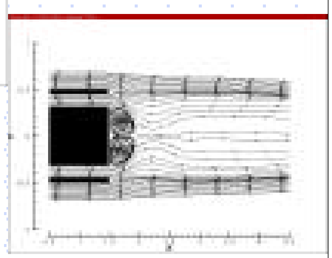
$$Re = \frac{U_j D}{\nu} \qquad R = \frac{U_j}{U_\infty} \qquad q = \int_{y_{bot}}^{y_{top}} U_{suct}(y)\,dy \qquad \delta$$




## Wake Manipulation by Means of Suction/Blowing



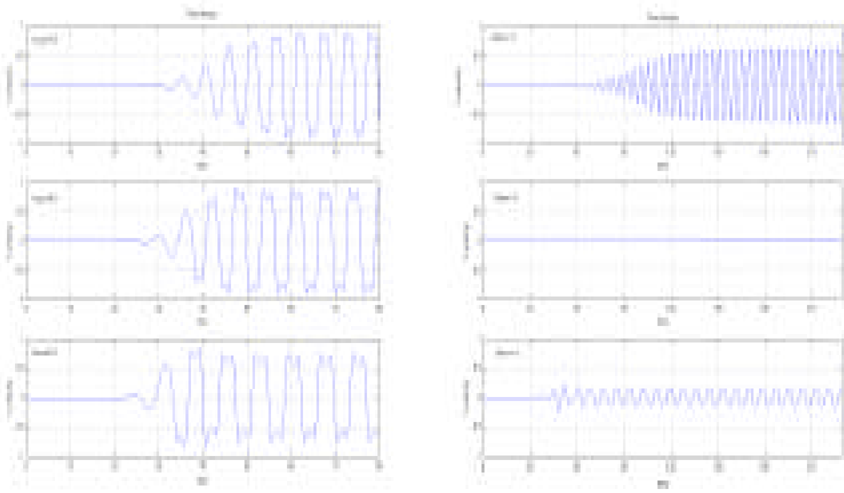




Differences in the Time-Averaged Velocity Field without forcing and with suction/blowing .
(Suction Velocity, Us=0.9, Re=800, δ=.3)

# Suction / Blowing

Suction | Blowing



Time history of the cross-stream velocity signal

# Conclusions

- A significant amount of drag reduction can be achieved by appropriate modification of the base geometry of a blunt body.
- The high pressure region at the trailing edge of the boattail seems to be associated with the change of streamline curvature in the notch-region.
- Suppression of vortex shedding can effectively be achieved by means of blowing fluid into the wake.

## Experimental Summary

• Supplying long flaps (flap length ≈ √A) to the model truck base results in a decrease in drag of about 20%, referenced to the drag of the model having no flaps.

• Referenced to the total drag of a more faithful truck model (wheels, etc.), the drag decrease would be about 10%.

• Referenced to the base drag alone--the most useful reference--the drag decrease is about 40%.

•A preliminary application of acoustic forcing--when added to flap—can produce an additional decrease in drag (referenced to the base drag).

• Acoustic forcing could be made effective with shorter flaps.

## Near-Term Experimental Tasks

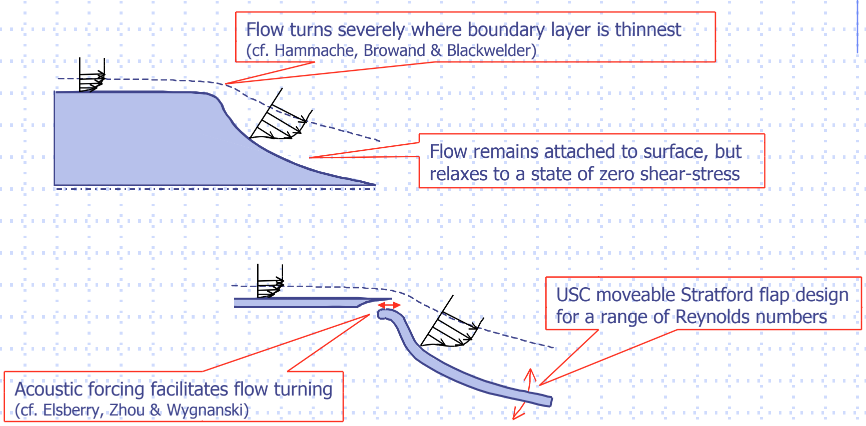
• Pay particular attention to much shortened flap lengths.

• Allow the four flaps to articulate, and allow systematic variation of flap angle, forcing frequency and forcing amplitude.

• Investigate more complex (quasi periodic) wave forms (c.f. Amitay & Glezer, "Controlled Transients of Flow Reattachment over Stalled Airfoils").

• Investigate Stratford-ramp flap shapes (c.f. Hammache, Browand & Blackwelder, "Whole-field velocity measurements around an axisymmetric body with a Stratford-Smith pressure recovery", *JFM*, in press).

## Stratford Ramp Applied to Trailer Base




## Numerical Modeling

- 2-D, low Reynolds number computations predict that boat-tail gives an overall base drag reduction of about 60-70%.

- *Preliminary* results also demonstrate that strong wake oscillations associated with global wake- mode instabilities can be suppressed by the application of blowing and/or suction.

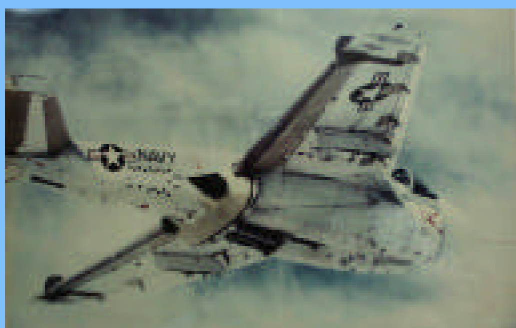
## Near-Term Numerical Tasks

- Perform numerical calculations to include periodic, zero mass flux blowing and more realistic flap geometry so as to make comparisons with our existing experimental results.
- Continue to define the limits of possible base drag reductions.

## Suggested Group Tasks

- Modify LES/DES codes to allow introduction of blowing and suction--including periodic, zero net mass flow perturbations, so as to realize comparisons with our experiments.
- Numerically explore the limits of realistic base drag reduction for high Reynolds number flow and 3-D geometry.
- Provide for experimental verification at high Reynolds numbers.

# DOE/GTRI Pneumatic Heavy Vehicle Aerodynamic Drag Reduction Program & Tuning Test Results
## ~DOE Heavy Vehicle Aerodynamic Drag Workshop~
## April 3, 2002
## by Robert J. Englar, Georgia Tech Research Institute




Application of Advanced Pneumatic Aircraft Technology....

...Through Analytical & Experimental Development ...

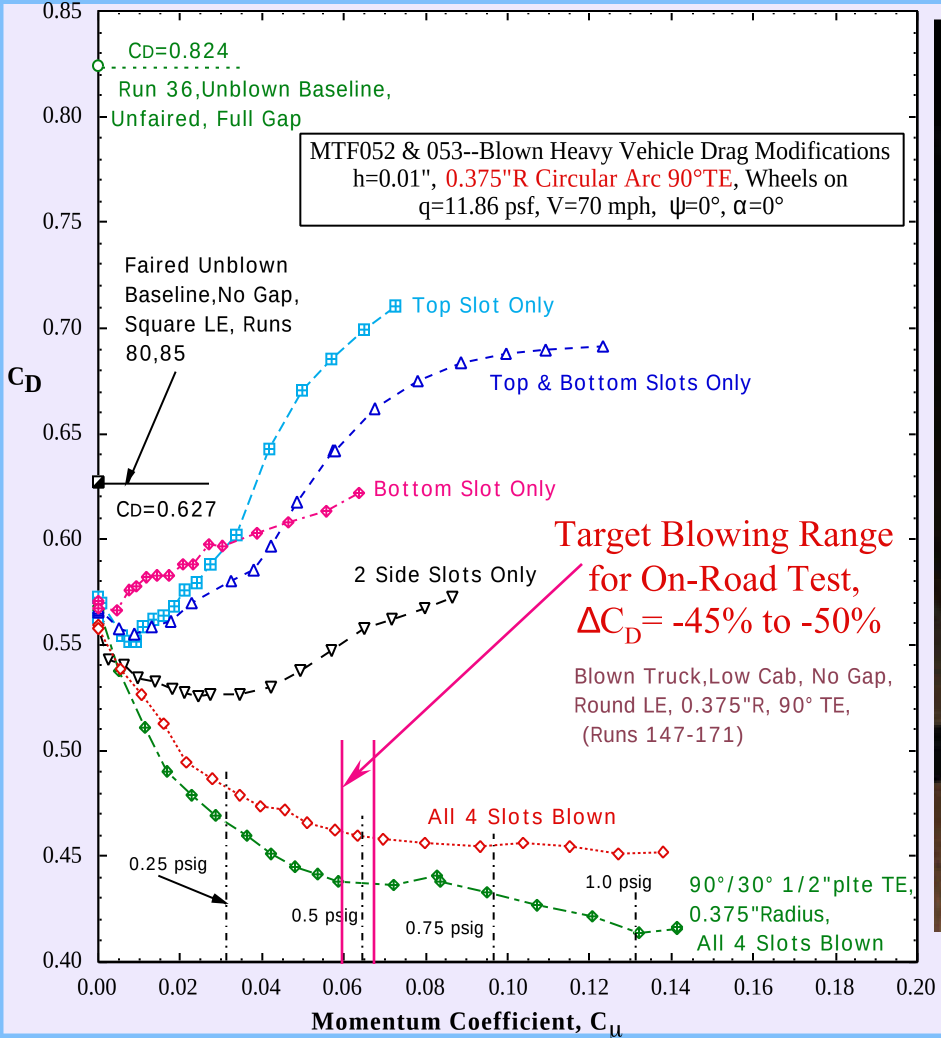
..To On-Road Proof-of-Concept Full-Scale Tests

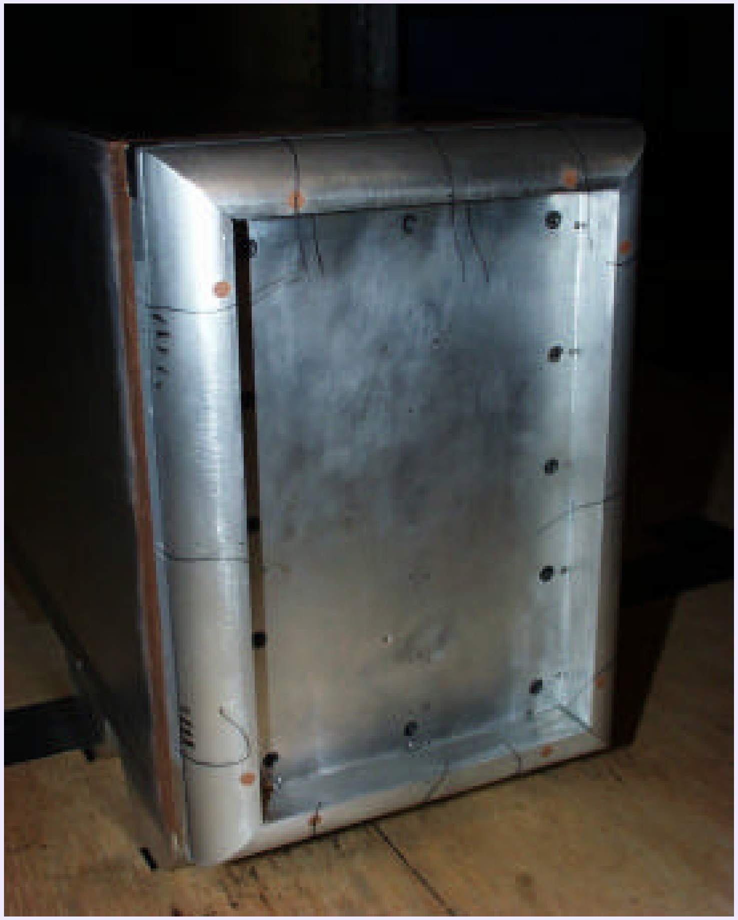
# Outline of Presentation

- Introduction: Pneumatic Heavy Vehicle (PHV) Technology
- Pneumatic Heavy Vehicles….Multi-Purpose Aerodynamic Devices:
  - Force & Moment Reductions or Augmentations
  - Fuel Efficiency & Wear Reduction
  - Improved Safety of Operation
  - Increased Stability (Directional & Lateral)
  - Reduced Splash, Spray Turbulence & Hydroplaning
  - No-Moving-Part Integrated Systems
  - Pneumatic Cooling Systems
- Review of Smaller-Scale Wind-Tunnel Model Test Results
- Full-Scale PHV Test Vehicle Design
- Initial Tuning Test of PHV at Volvo Trucks in N.C.
- Continuing Plans
- Conclusions: So, where do we go from here ?…

  ... Or, how do we PROVE this potential on a real vehicle ??

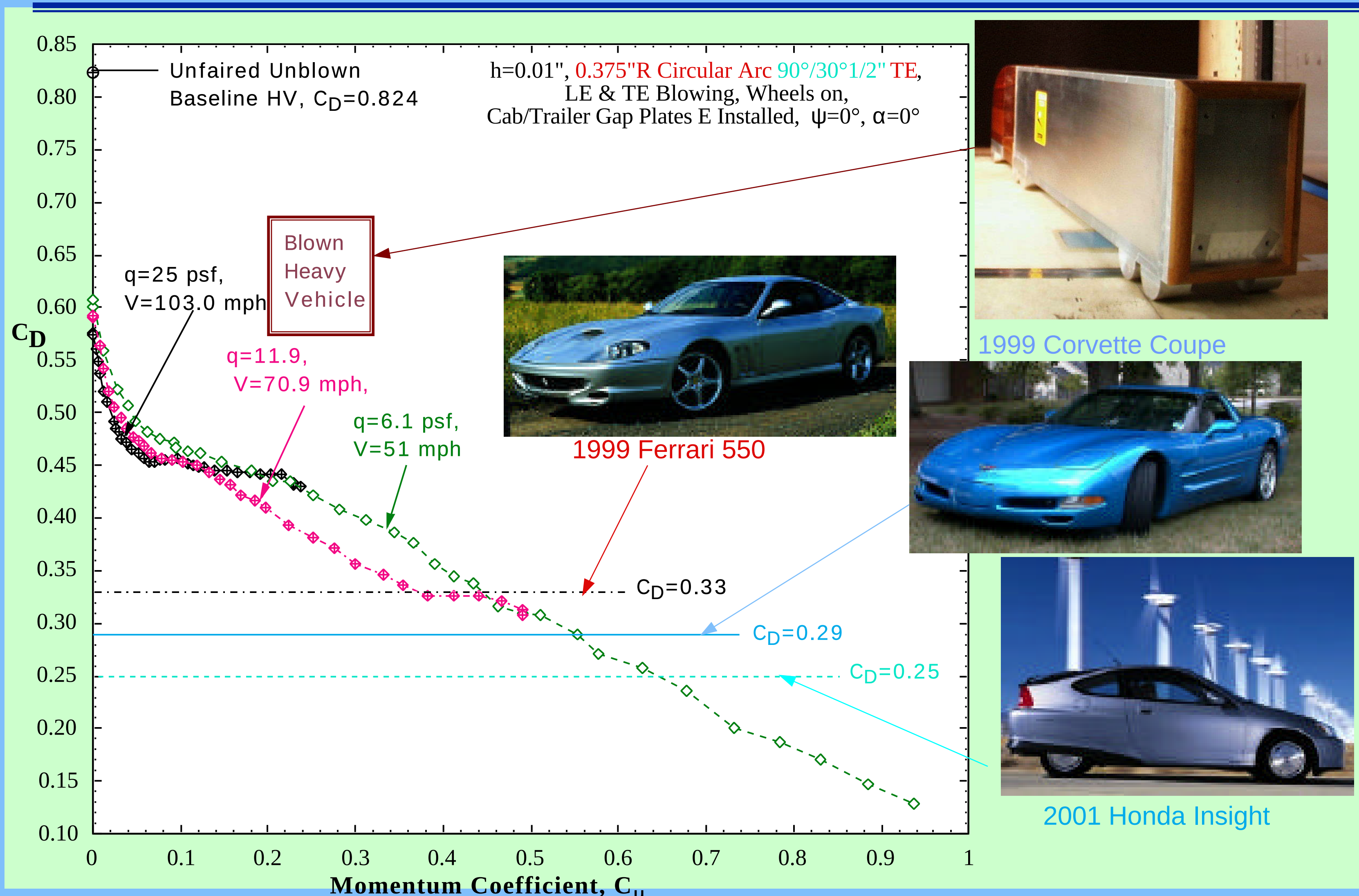
# Background: Aero Development & Tunnel Tests at GTRI Showed 50%(or more) Drag Reduction due to Aft Blowing of Various Slots



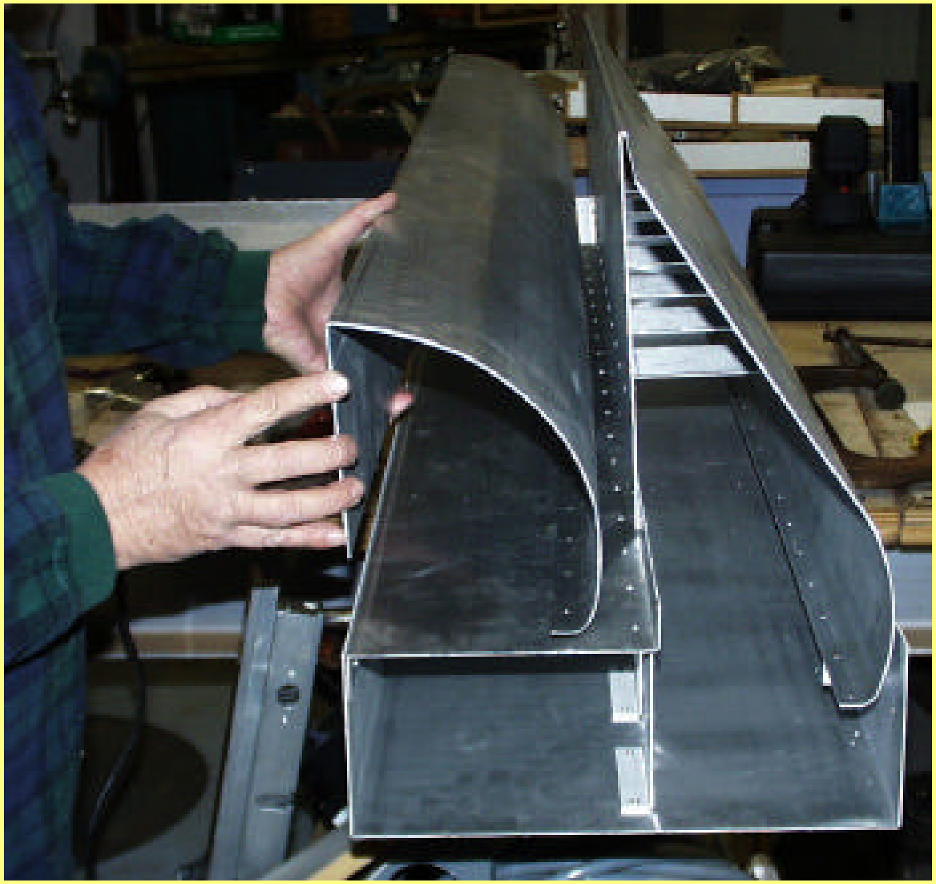



4 Blown Slots on Trailer Rear Doors Of Wind-Tunnel Model

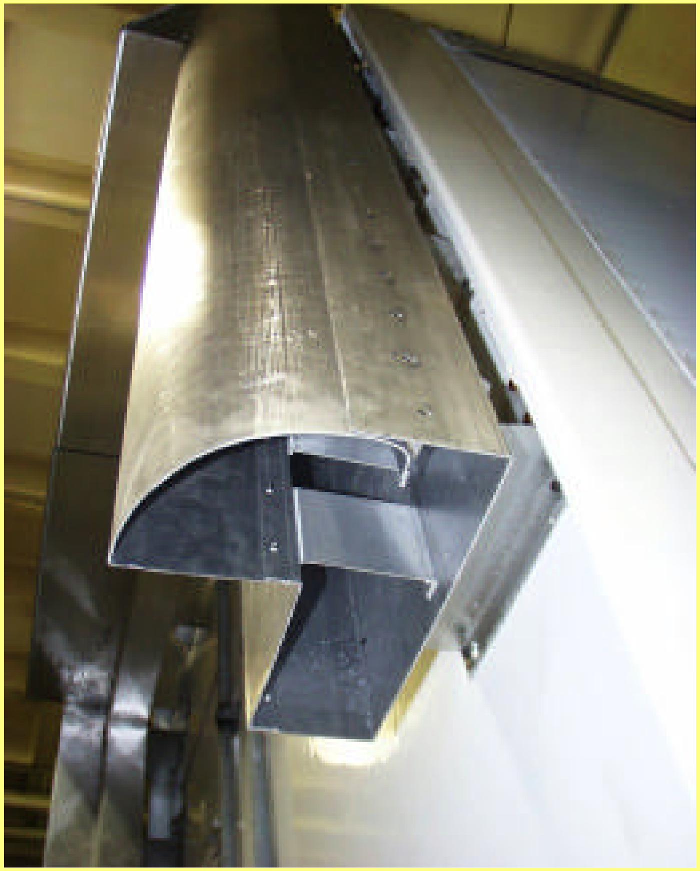
# GTRI Extended Tunnel Tests Showed State-of-the-Art Drag Reduction!!



# Trailing Edge Turning Surface Geometries



Plenums, Slots and Turning Surfaces,
Showing 90° (left) & 30°

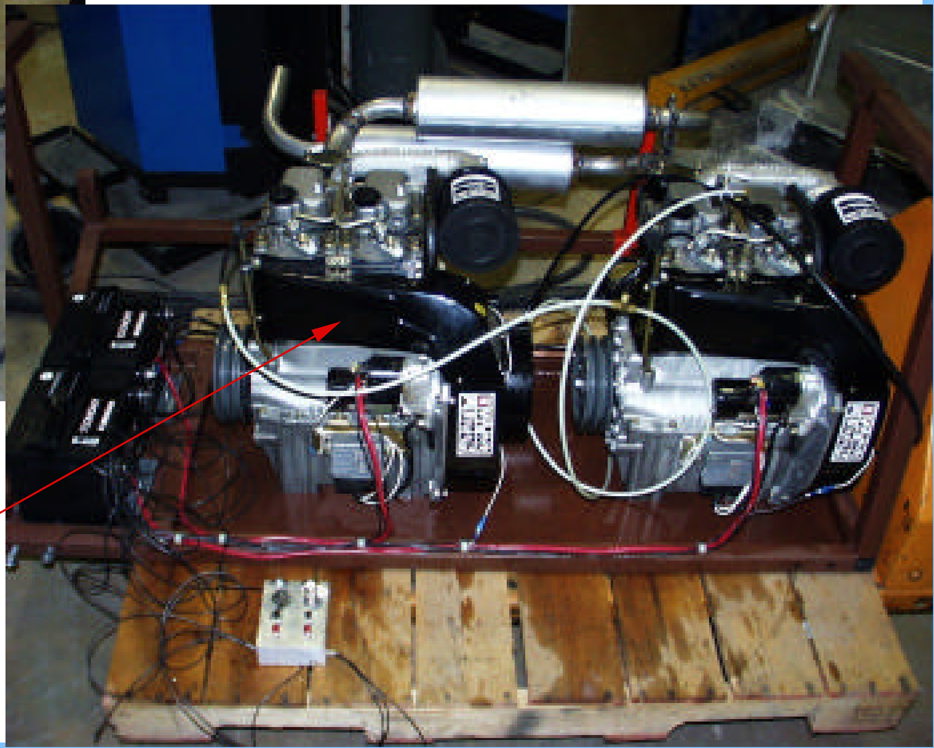


Right Rear Corner, looking up--
90° Side and 30° Top

# Air Source Consists of Blowers, Drive Diesels & Mounting Platform

Air Source = New York Blower Co.
Centrifugal Blowers (2)

Deutz 20 hp Diesels (2)
To Drive Blowers



# Rear View of Assembled Trailing-Edge Blowing System

Doors Open in GTRI High Bay, Showing
Blown Trailing Edges & Personnel Door

Doors Closed, PHV Approaching
Final Assembly Area at GTRI

# Internal Wiring, Structure and Instrumentation

X-wire Bracing, Nat'l Instr. Pressure/Temp Instrumentation , and Data Transmission

Diffuser, Plenum, Duct, Slot, Slot Adjusters, 30° Turning Surface

# Static Testing of Trailing Edge Blowing System

Blower, Screen, Diffuser &
Left Turning Surface (open)

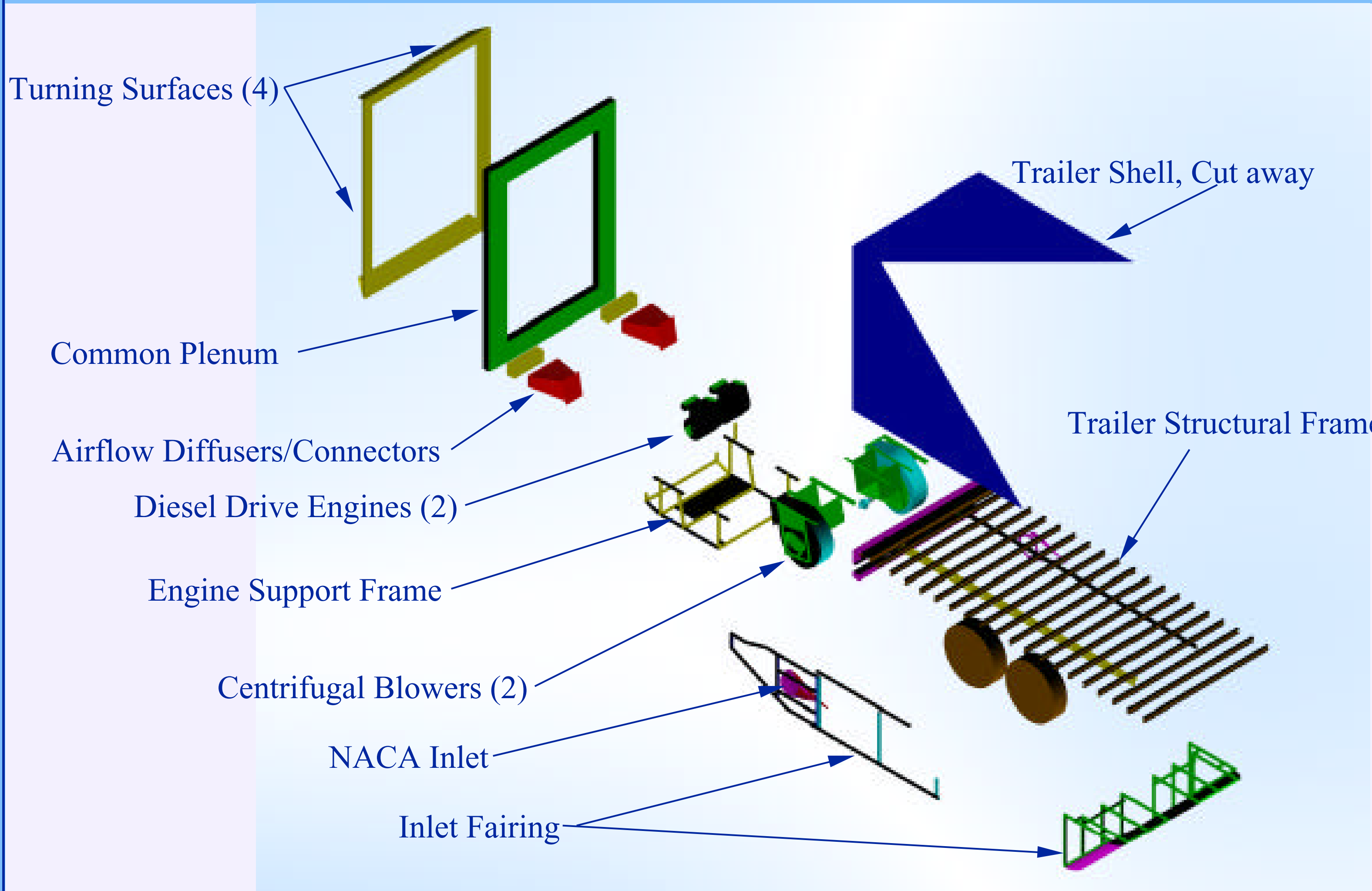
Tuft Showing Flow Exiting the Diffuser
and Entering into Right Plenum

# Final Assembly at GTRI and Departure to N.C.

Installing "Radome"= LE Fairing
and Data Telemetry Antenna Cover

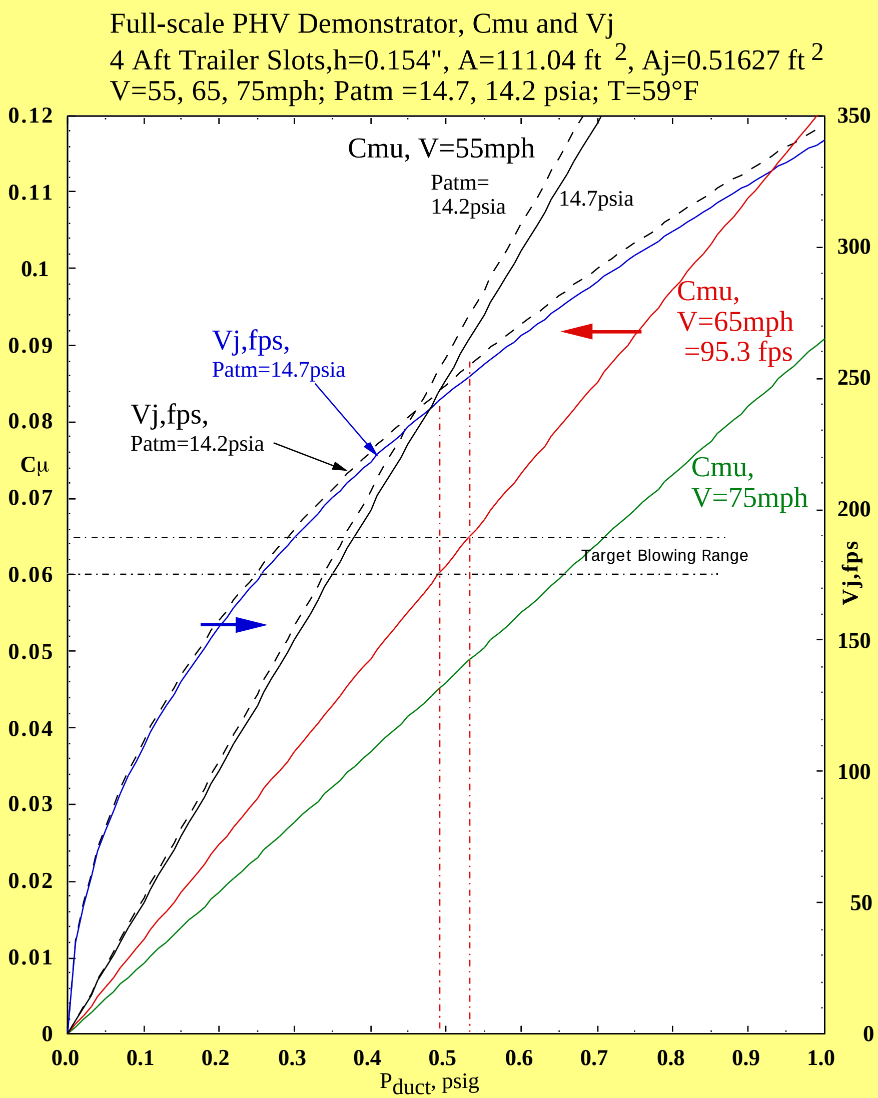
Departure from GTRI to Volvo;
Trailing Edge Still Unsealed

# PHV Trailer Modifications for Blowing Systems




Designed & Modified by Prototype Shop Novatek, Inc

# Blowing Design Parameters



# Pneumatic Heavy Vehicle Trailer Compared to Baseline Reference Trailer from Great Dane

Test PHV Features:
- 4 jet turning surfaces with plenums and blowing slots
- NACA inlet to entrain free-stream total pressure into blowers
- Diesel-driven external blowers feeding diffusers to plenums to slots
- Volvo engine fuel system, GTRI data telemetry of blowing parameters

# Flow Visualization of Blowing Jets

Tuft Showing Flow Uniformity at Diffuser Center

Combined Jet Strength and
Wake Contraction (see Shirt)

# Static Jet Turning Displayed During Run-up Testing

Setting Slot Heights and Confirming
Jet Turning at Low Blowing Rate

Right Rear Corner, looking up--
Tufts Show Jet Turning to Left:
90° on Side and 30° on Top

## First Tuning Test Conducted at Volvo Trucks of North America, February 28-March 1, 2002

Objectives: • Blowing Optimization for Upcoming Fuel-Economy Test at TRC
• Instrumentation, Blowing, Data Reduction, & Control Systems Checkout

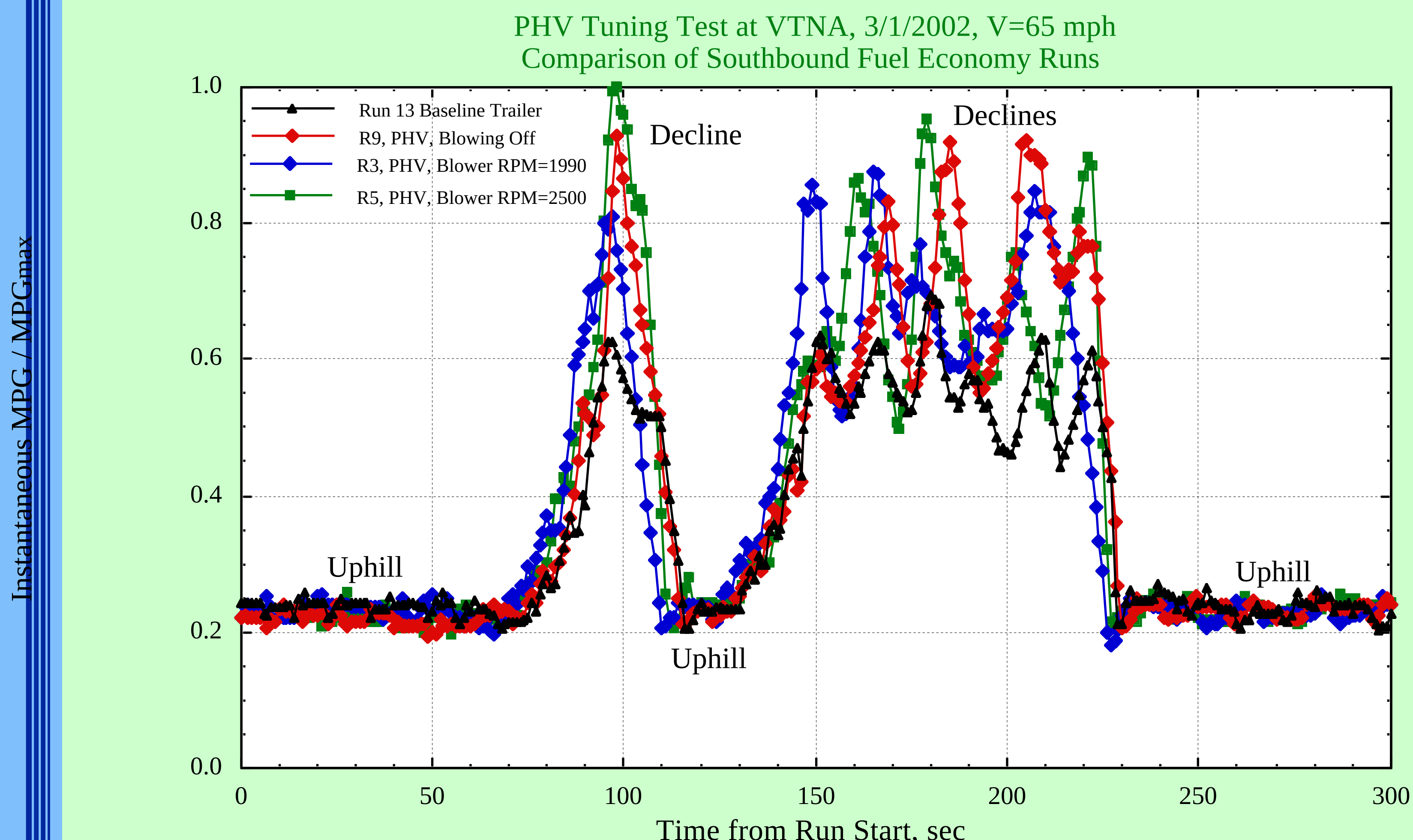
Conducted by : GTRI, Novatek, Volvo

## On-the-Road Operation:Jet Turning Entraining the Flowfield and Reducing Vehicle Drag


Rear View with Jets Blowing


Close-up of Tufts Showing Jet Turning

# Tuning Test Preliminary Results, Southbound MPG




Typical Fuel Consumption Recorded during Blown, Unblown, and Baseline Test Runs at 65 mpg

# Tuning Test Preliminary Results (V=65 mph), Changes in Time-Averaged Fuel Economy, %MPG

| Configuration | Test Runs | Blower RPM | Route Direction | %MPG change | %Config'n MPG change |
|---|---|---|---|---|---|
| Baseline Trailer | 13 | 0 | Southbound | 0.00 | 0 |
| | 14 | " | Northbound | " | |
| PHV, No Blowing | 9 | 0 | SB | 11.37 | 8.39 |
| | 10 | " | NB | 6.04 | |
| PHV, Moderate C µ | 3 | 1980-2000 | SB | 10.80 | 9.36 |
| | 4 | " | NB | 8.22 | |
| PHV, Higher C µ | 5 | 2500 | SB | 15.30 | 14.25 |
| | 6 | " | NB | 13.41 | |

# Tuning Test Preliminary Results (V=65 mph), Comparison to GTRI Wind Tunnel Results, and Conclusions

| Configuration | WindTunnel $C_D$ | % $C_D$ Change | % Equiv. GPM Reduction | Road Test Run No. | % GPM Reduction | % Equiv. $C_D$ Change | % MPG Increase |
|---|---|---|---|---|---|---|---|
| Baseline, No Gap, Sq. LE & TE | 0.627 | 0 | 0.0 | 13 (Gap) | 0.00 | 0.00 | 0 |
| Unblown PHV, Cmu=0 | 0.57 | -9.1 | -4.6 | 9 | -10.21 | -20.42 | 11.37 |
| PHV,4 Slots Cmu=0.05 | 0.44 | -29.8 | -14.9 | 5 | -13.27 | -26.54 | 15.30 |

**CONCLUSIONS:**

- Limited Tuning Runs confirmed up to **15.3% increase in MPG,** or about **26.5% reduction in $C_D$**, due to blown PHV configuration, **but** this first Tuning Test **was not optimized** (Speed, Temps, Blowing rate, etc.)
- Plans to conduct 2nd Tuning Test (TT2) with suggested test procedure and vehicle improvements prior to SAE fuel economy test at TRC

# Initial Tuning Test Problems -- Correct for TT2

**Corrections to be made:**

- Right Diesel stopped (errors in some blowing data); Repair engine
- Change gearing on diesel-to-blower connections
- Bottom and front engine fairings were omitted: Install these
- No fuel flow meters for blower diesels; Install these
- Free stream pitot-static probe in side wall boundary layer; re-locate

**Improvements to be made:**

- Run at higher speed for more Aerodynamic Dominance (75 vs 65 mph)
- Run on warmer day with some sidewinds and gusts
- Reduce blowing slot height for higher Vj
- Run with less effectively faired tractor

# Upcoming SAE Type 2 Fuel Economy Tests on PHV

- At Transportation Research Center (TRC), East Liberty, OH: Summer ,2002
- 1 PHV Test Truck & 1 Control HV, running simultaneously on 8-mile track
- Both HVs Loaded to Typical Operating Weight ( ~60,000 lb.)

- Test Configurations for PHV (each run = 3 speeds, 2-3 days; 450 miles):
  1. Blowing **On** , **C$\mu$ = best**
     1.a, 1.b: Two Optional Blowing-on Runs: **Intermediate C$\mu$'s**
  2. Blowing **Off**, **C$\mu$ = 0**
  3. Blowing **Off,** Round Leading-Edge and Trailing-Edge Aero **Surfaces Off**
  4. Blowing **Off,** Engine, Blower & Fairing Components Off = **Baseline Trailer**
  5. Mirrors **Off**, for DOT

- <u>**Results:**</u> For each Configuration: Fuel Burned / Miles Driven, corrected by Control HV

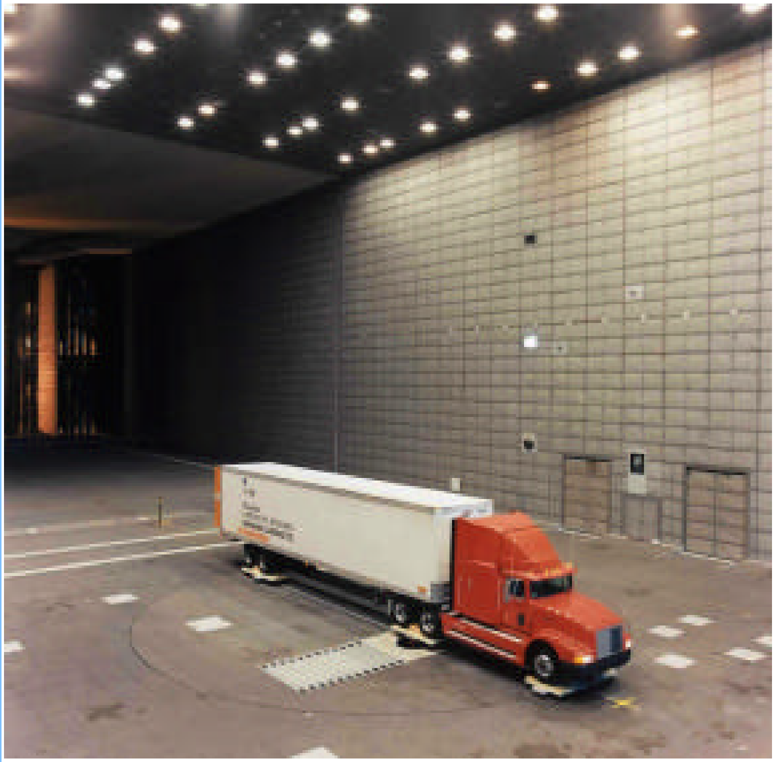
## CONCLUSIONS:Pneumatic Aerodynamic Concepts Now Verified ~Offer Significant Potential For Improvement of Heavy Vehicles; (Green = Confirmed in Tuning Test 1)

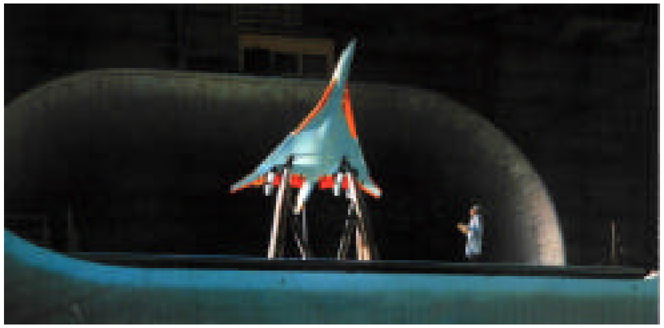
- Pneumatic Devices on trailer, **blowing slots on all sides and/or front top**
- **Separation control & base pressure recovery, LE suction = drag reduction; or Base suction = drag increase** **Latest test results: Blowing-on $\Delta C_D$ = -26%or more**
- Additional lift for **rolling resistance reduction** ($F_{Rolling} = \mu N$, where N=Wt - Lift), **or** Reduced lift (increased download) for **traction and braking**: instantaneously **switchable**
- Partial top/bottom slot blowing for **roll control & lateral stability**
- One-side blowing **(LE or TE)** for **yaw control & directional stability**
- **Aerodynamic control** of all three forces and all three moments
- **No moving parts**, small component drag; **Very short aft addition=no length limitation**
- **Splash, Spray & Turbulence** Reduction; Reduced **Hydroplaning**
- Use of **existing** on-board compressed **air sources** (exhaust,turbocharger,brake tank,electric)
- **Advanced Pneumatic Cooling Systems (Aerodynamic Heat Exchanger)**
- **Safety of Operation**
- **First On-Road Test Now Completed; MORE to Come!!**

GTRI PATENTED CONCEPTS

# Follow-On Large-Scale Wind Tunnel Investigations

**To Investigate:** Full-Scale $C_D$ alone; Lateral /Directional Stability; Side Winds (Yaw); Safety of Operation; Full-Scale Reynolds Number





NASA Ames Full Scale Complex 80’ x 120’, V=115 mph ..Or.. ODU Langley Full Scale Tunnel, 30’ x 60’, V=80-120 mph

# Computational Prediction for a Simplified Truck Geometry


**Walter H. Rutledge**

**Mary McWherter-Payne, Chris Roy,**

**Dave Kuntz and Jeff Payne**

**Aerosciences and Compressible Fluid Mechanics Department**

**Sandia National Laboratories**



**Heavy Vehicle Aerodynamic Drag: Working Group Meeting**

**Lawrence Livermore National Laboratory**

**April 3rd and 4th, 2002**

# Outline

- Introduction – SNL Role
- FY02 Tasks and Budget
  - Status
  - Results from 2D GTS grid studies
  - New 3D GTS grid
- Additional Tasks (unfunded)
  - Dissection of 10 Degree Yaw GTS Solution
  - GCM
    - 2D
    - 3D
- Leveraging (additional money, ESRF)
- Conclusions

# Introduction

- Overall SNL Role: To provide technical insight to industry relative to:
  - the role of current and future (advanced) computational methods for truck/trailer aerodynamic design
  - Aerodynamic drag reduction for truck/trailer systems
- At end of FY00, SNL moved from just RANS to hybrid RANS/LES
- FY02:
  - The focus is on better y+ resolution for turbulence modeling (New 2D and 3D grids)
  - New SNL participants (Chris Roy, Dave Kuntz, Jeff Payne) in addition to Mary McWherter-Payne

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# Sandia Computational Approach

**Steady RANS**

- Spalart-Allmaras
- k-epsilon
- k-omega Wilcox

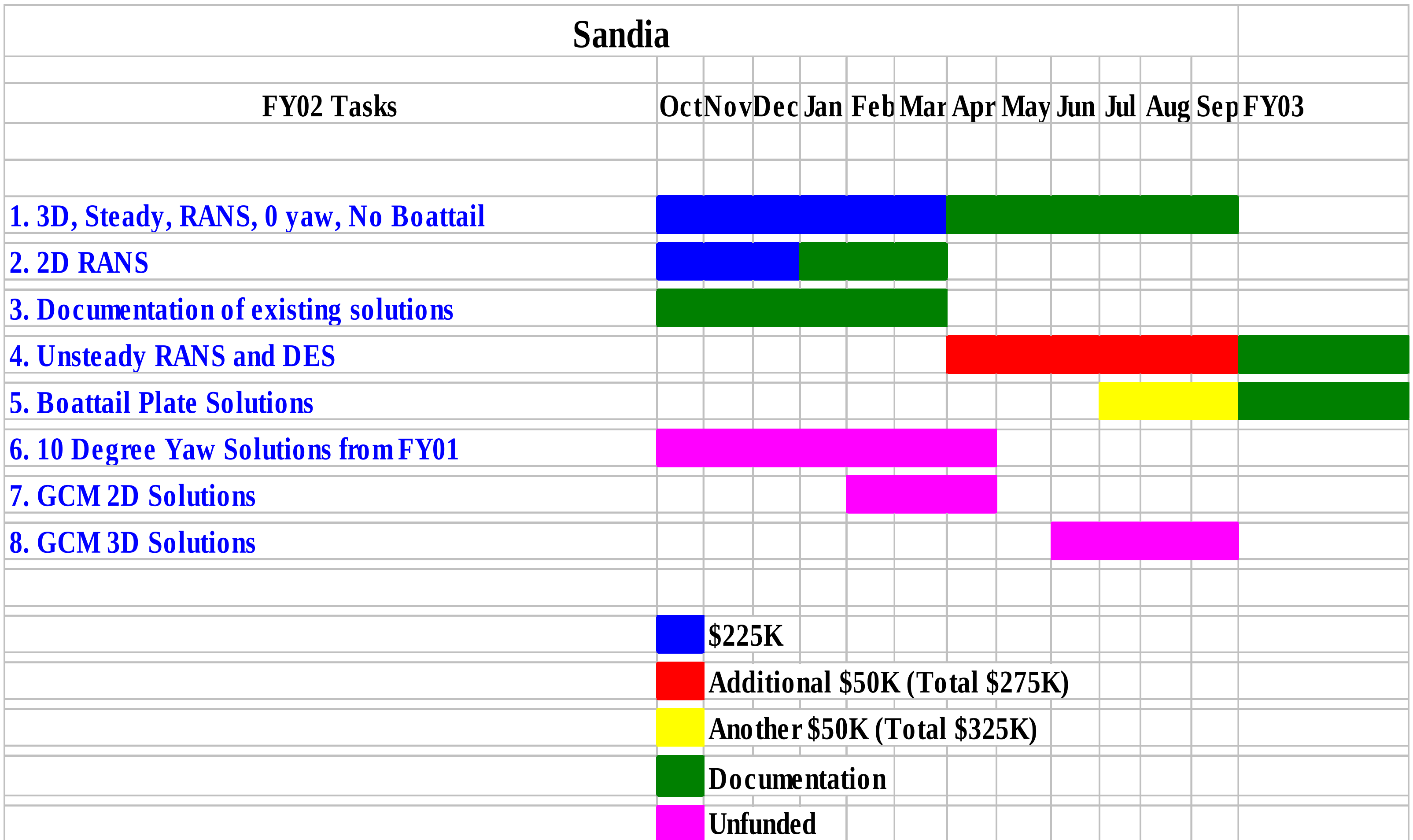
**Unsteady RANS**

- Spalart-Allmaras
- k-omega Wilcox
- Durbin's $v^2$f

**Hybrid RANS/LES**

- Detached Eddy Simulation
- Hybrid RANS/LES

# Sandia FY02 Tasks and Budget

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# Status of FY02 Tasks

- **Task 2: GTS, 2D, RANS** Documentation in progress
  - y+ grid studies completed
  - 2D solutions completed for three turbulence models
- **Task 1: GTS, 3D, Steady RANS**
  - New 3D mesh completed
    - Coarse (300,000 cells)
    - Medium (2.5 million cells)
    - Fine (20 million cells)
  - Grid needs to be decomposed for parallel processing
  - k-omega/Wilcox ready to run
  - Other models to be run (time/funds permitting)
    - Spalart-Allmaras, k-epsilon

# Status of FY02 Tasks

- Task 3: Documentation of existing solutions
  - SNL memo submitted for review (April 2002)
  - Working with LLNL on documentation of previous SNL activities (through FY01)
    - Salari and McWherter-Payne
- Task 4: Unsteady RANS and DES no activity
- Task 5: Boattail with RANS: no activity

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# Additional FY02 Tasks (Unfunded)

- Task 6: GTS, 10 Degree Yaw (FY01 medium mesh, S-A)
  - Flow field plots
  - Comparisons with experiment
    - Drag
    - Skin friction
    - Pressure Coefficient
- Task 7: 2D, GCM
  - Generated multiple meshes
  - k-omega/Wilcox medium mesh solution obtained
  - Appropriate y+ values determined
- Task 8: 3D, GCM
  - Obtained NASA ProE file, but surfaces are missing

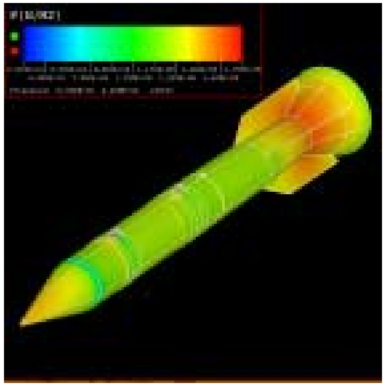
# The  Budget, The Team

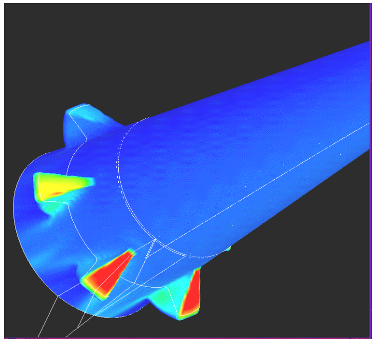
- The Budget: $225K ($50K less than anticipated)
- The Team:
  - Walt Rutledge (Manager)
  - Mary McWherter-Payne
  - Chris Roy
  - Dave Kuntz
  - Jeff Payne (consulting)

# SACCARA Code Capabilities

**Sandia Advanced Code for Compressible Aerothermodynamics Research and Analysis**

- **Multi-block, structured grids for 2-D, Axisymmetric, and 3-D flows**
- **Solution of the Full Navier-Stokes equations for compressible Flows**
- **Finite volume spatial discretization (steady and unsteady)**
- **MP implementation on a variety of distrubuted parallel architectures (IBM, Intel, etc.)**
- **Implicit time advancement schemes**
- **Subsonic → Hypersonic flows**
- **Zero-, one-, and two-equation turbulence models**
- **Ideal, equilibrium, and thermo-chemical nonequilibrium finite-rate gas chemistry**
- **Rotating coordinate system**





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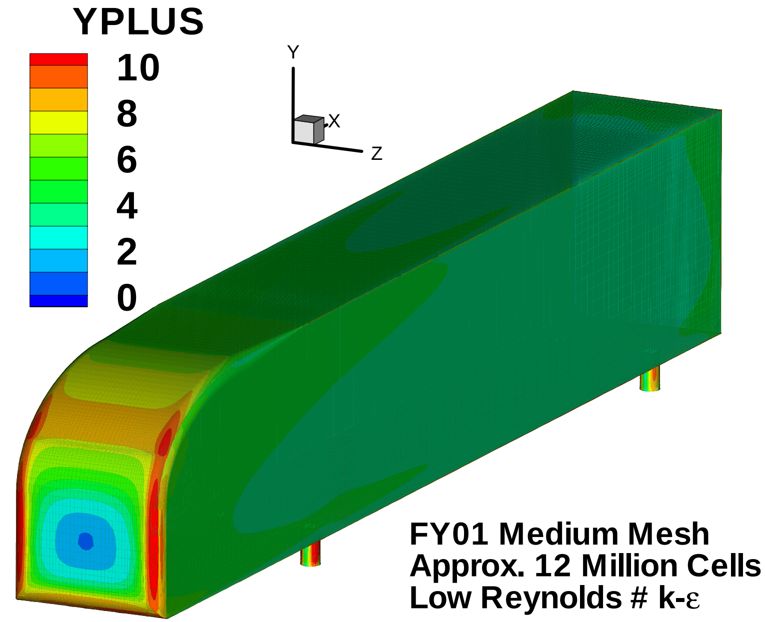
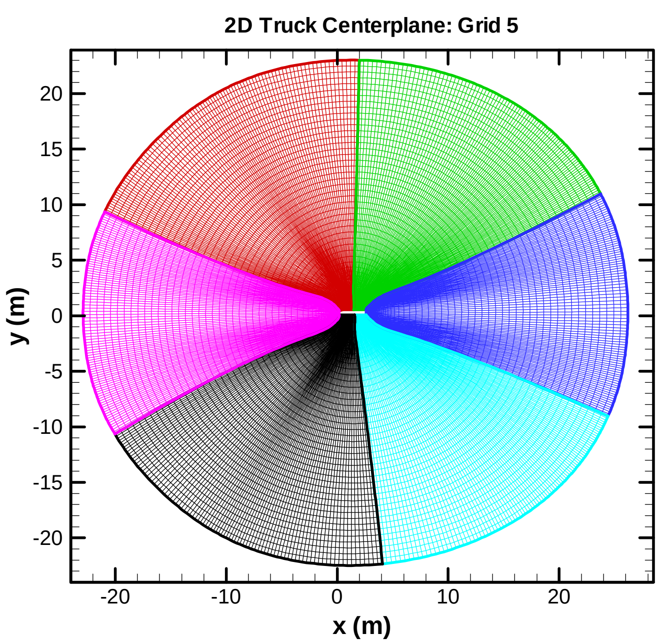
# Task 2: GTS 2D Grid Studies

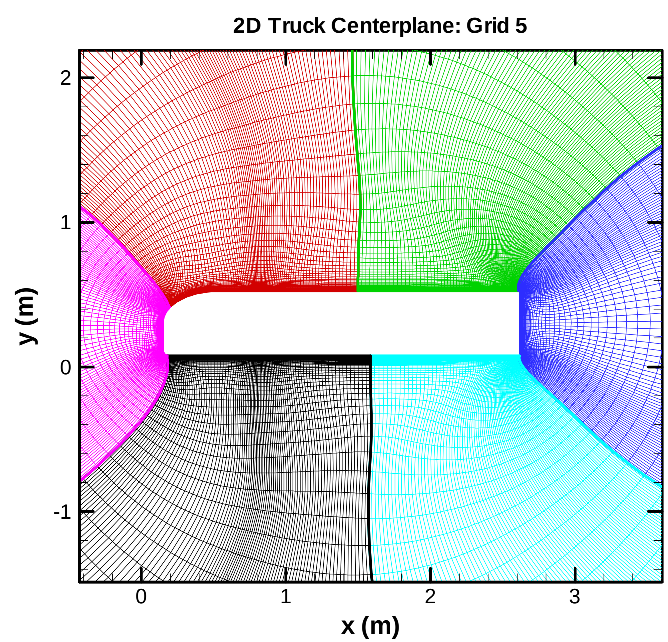
*Want to understand strengths/weaknesses of RANS models*

- **Motivation:** During FY01, it was determined that k-omega/Wilcox would not run on FY01 medium mesh (12 million cells)
  - suspected that wall y+ values were too large
- 5 new 2D meshes completed with max y+ of 0.5, 1, 2, 5, 10
- Ran k-omega/Wilcox, k-epsilon and Spalart-Allmaras on all five meshes to determine:
  - Required y+ to obtain solution
  - Effect of y+ on accuracy of solution

- Previous 3D mesh: $y^+$ too large
- New 2D meshes for $y^+$ study
  - hyperbolic meshes (no tunnel)
  - retain FY01 axial spacing
  - grid1: FY01 normal spacing
  - grid2 through grid5: refine in wall normal direction only

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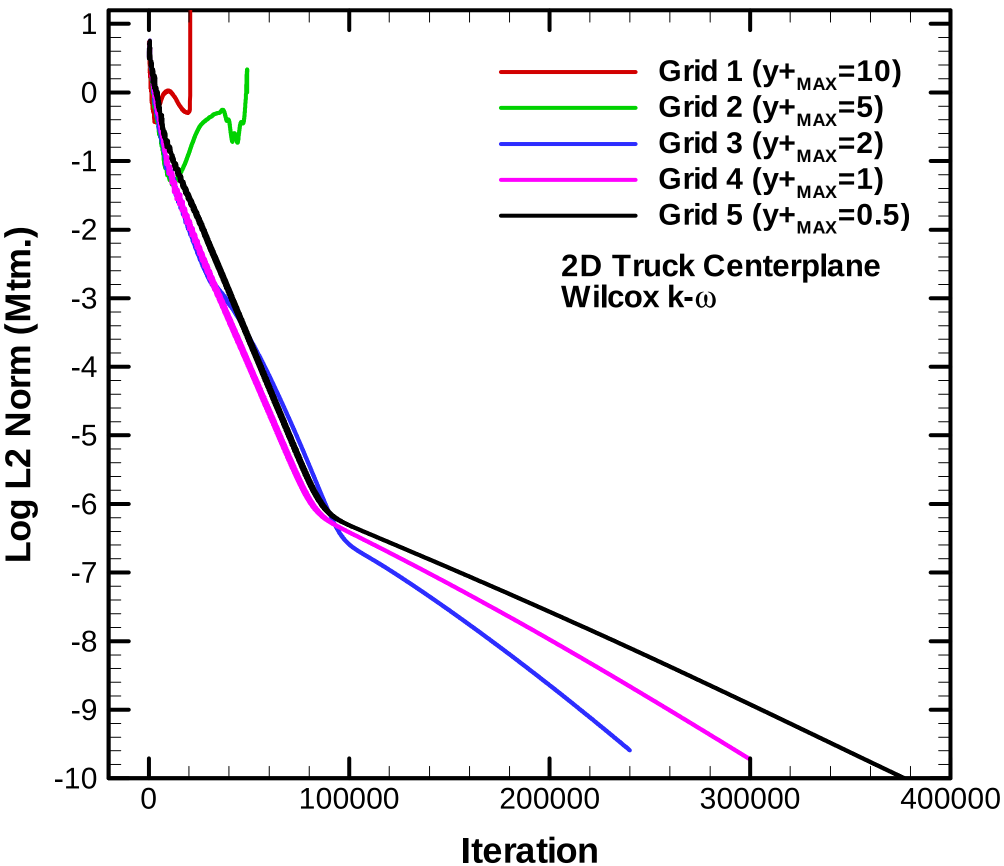
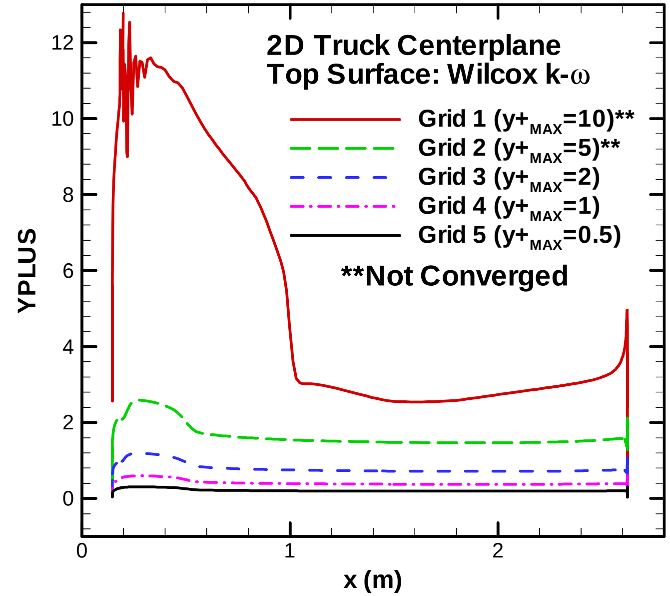
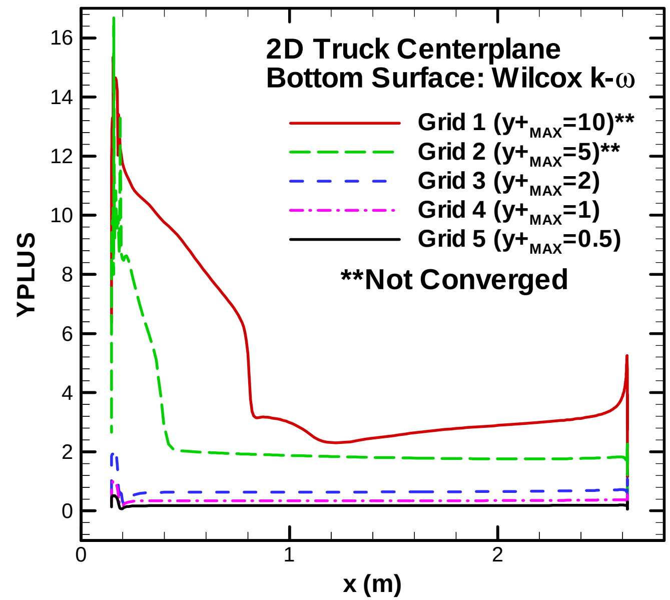
# Wilcox (1998) k-ω




## Convergence History




## Wall y+ Values

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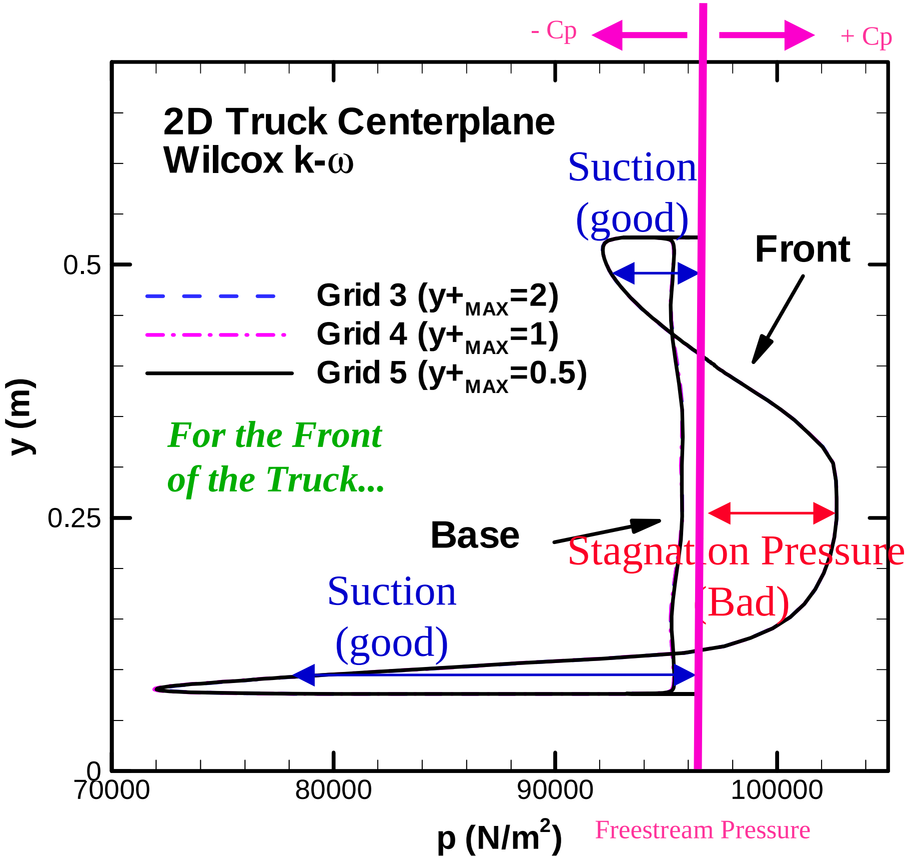
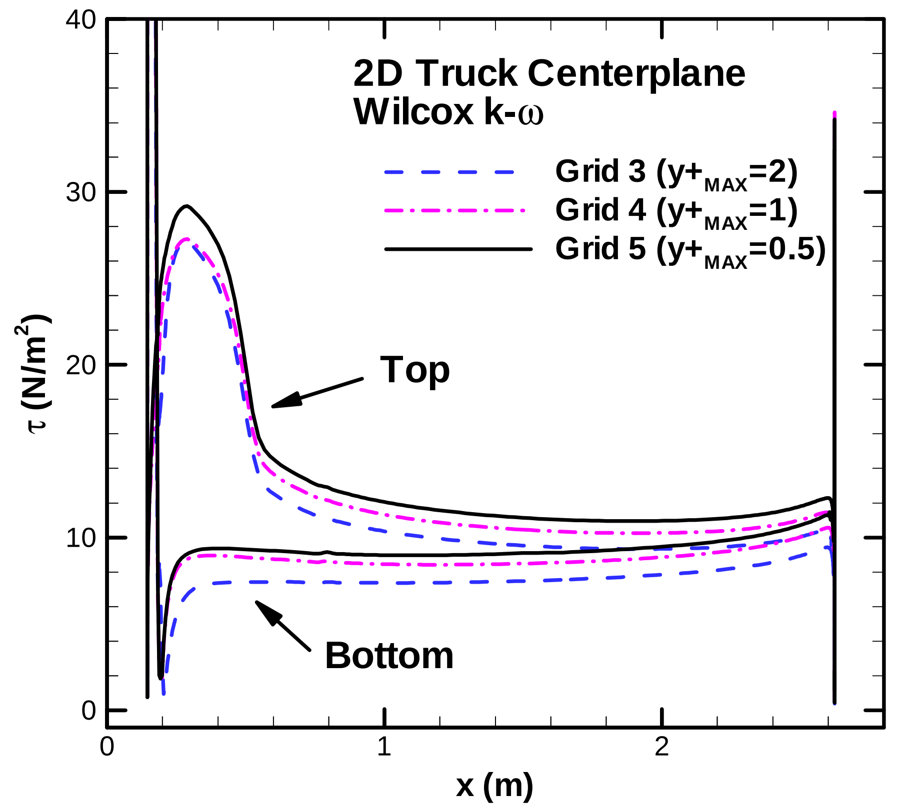
# Wilcox (1998) k-ω




**Surface Pressure**




**Shear Stress**

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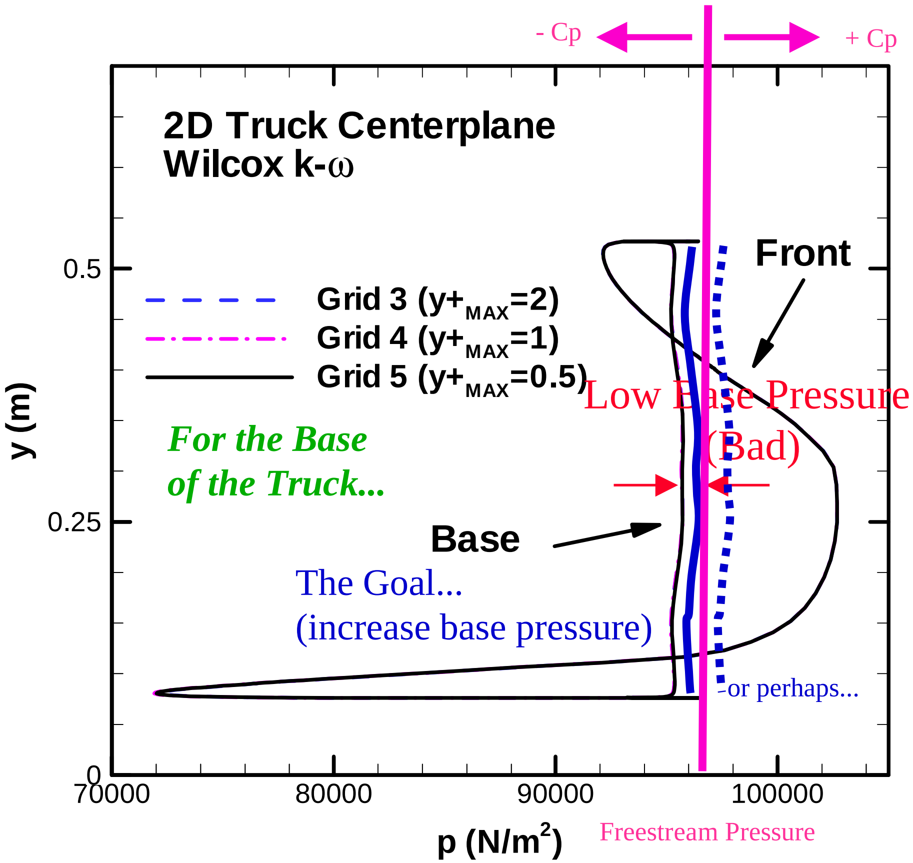
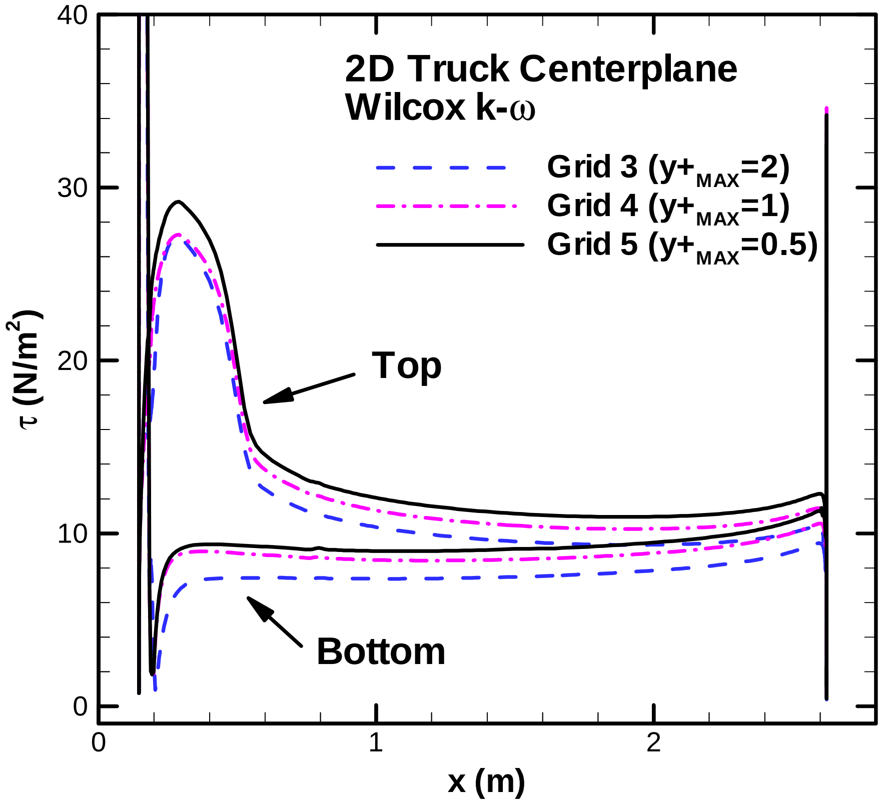
# Wilcox (1998) k-ω

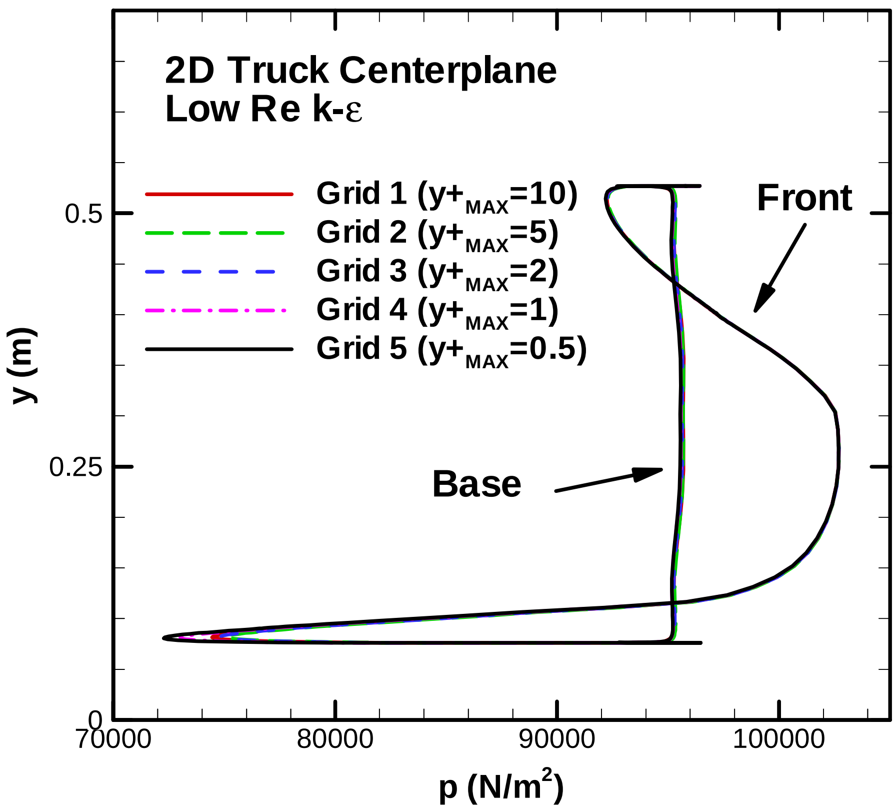



## Surface Pressure

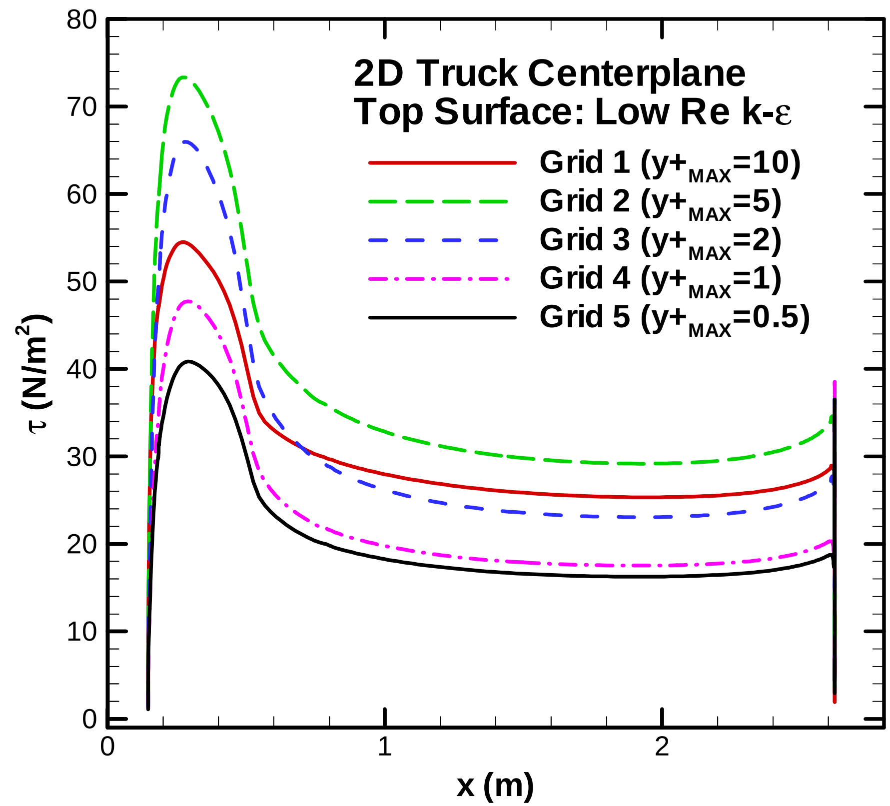



## Shear Stress

# Low Reynolds Number k-ε




**Surface Pressure**




**Shear Stress**

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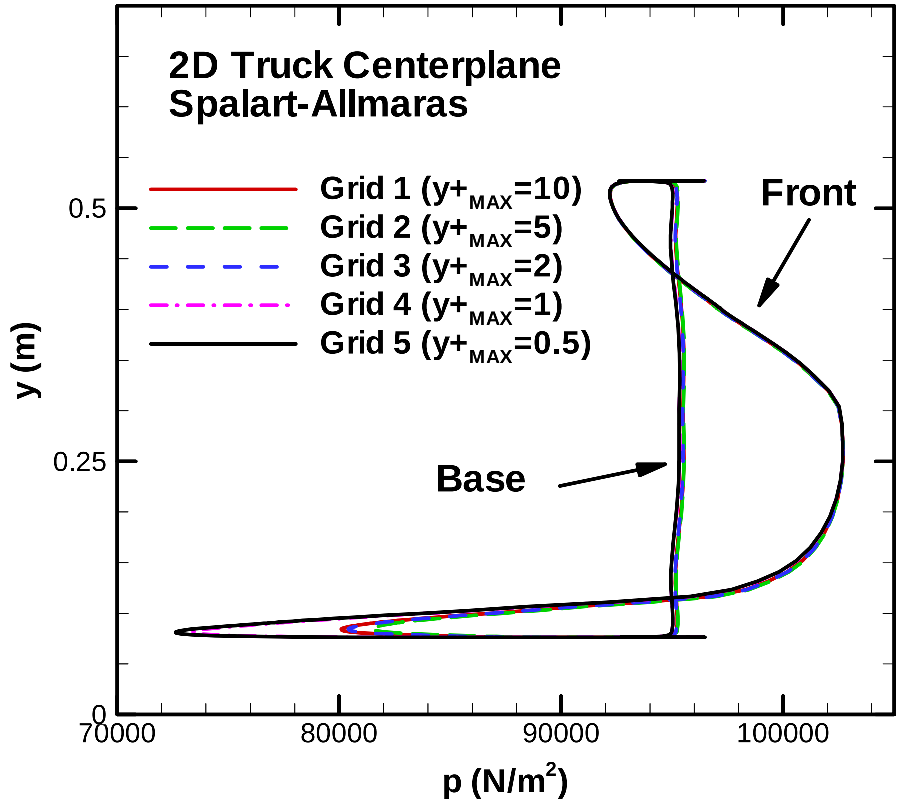
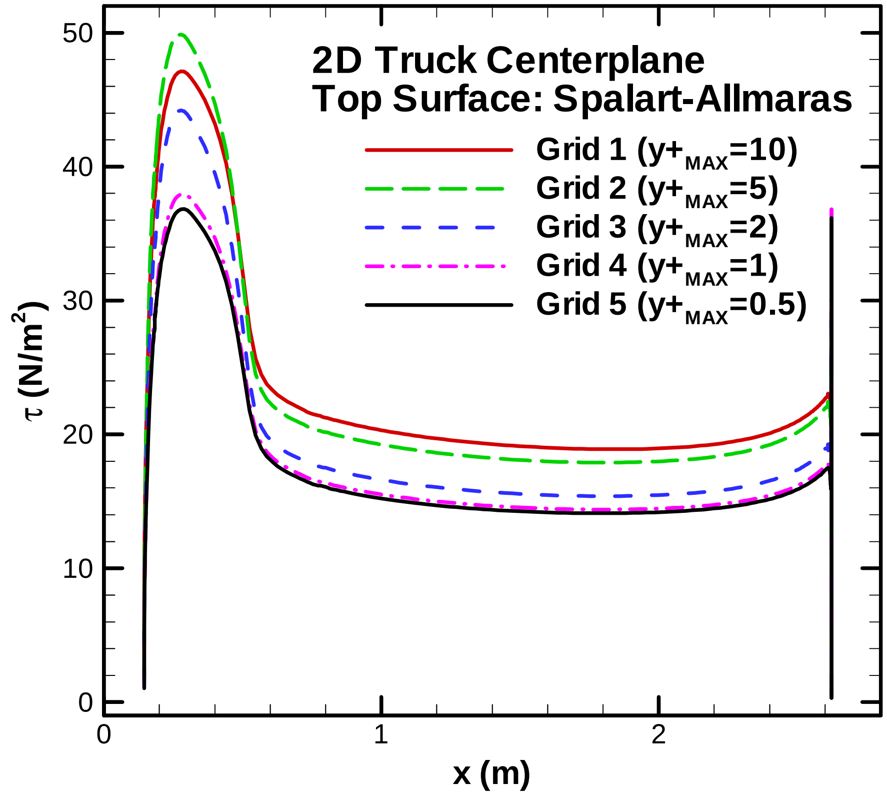
# Spalart-Allmaras

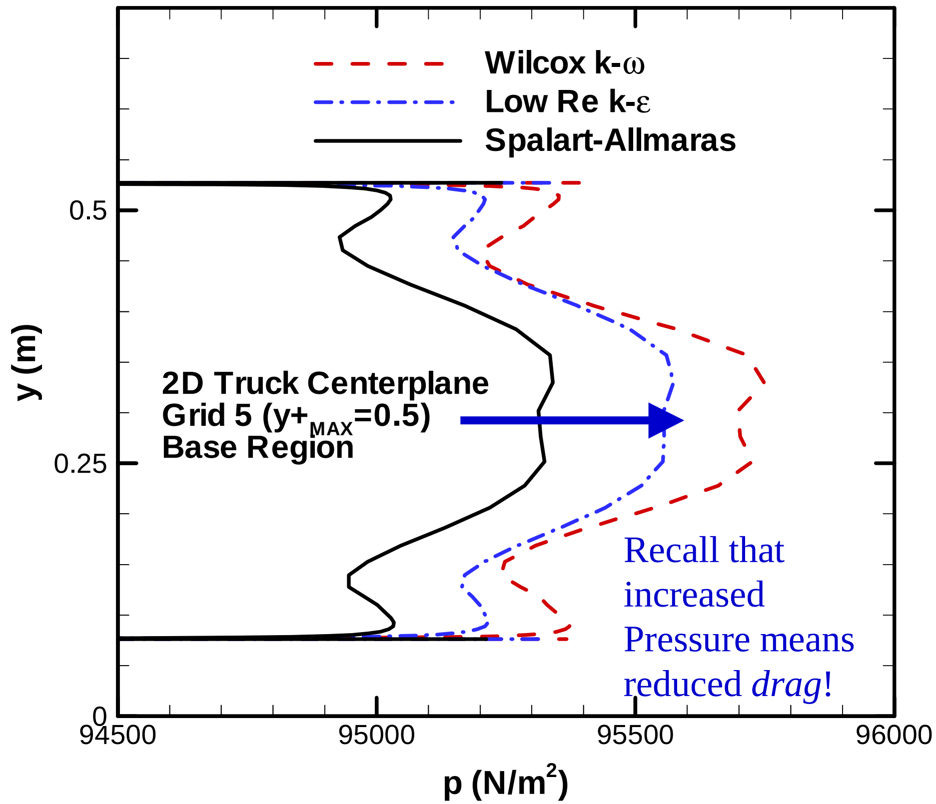



**Surface Pressure**

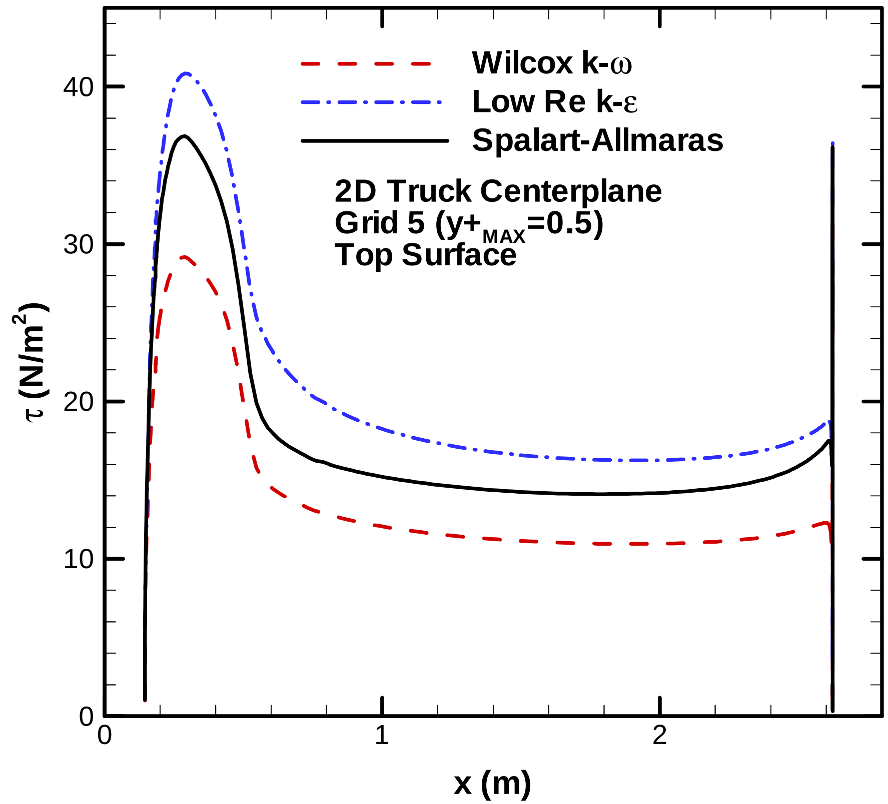



**Shear Stress**

# Model Comparison: Grid5



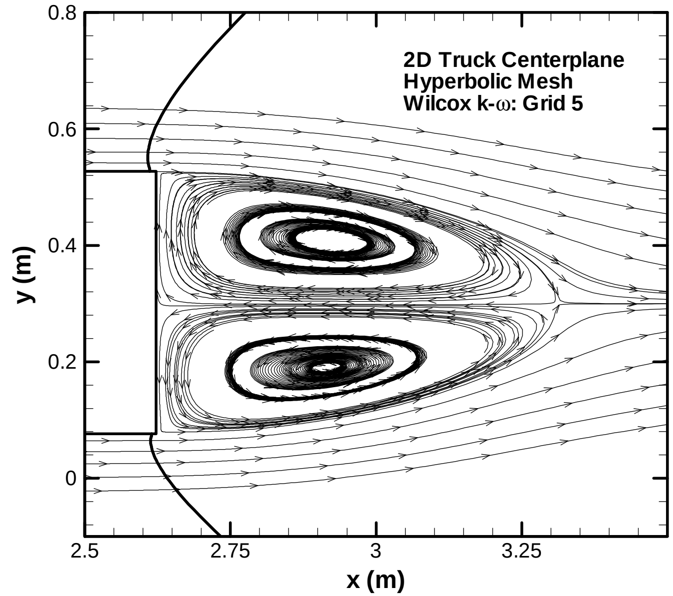

**Surface Pressure**

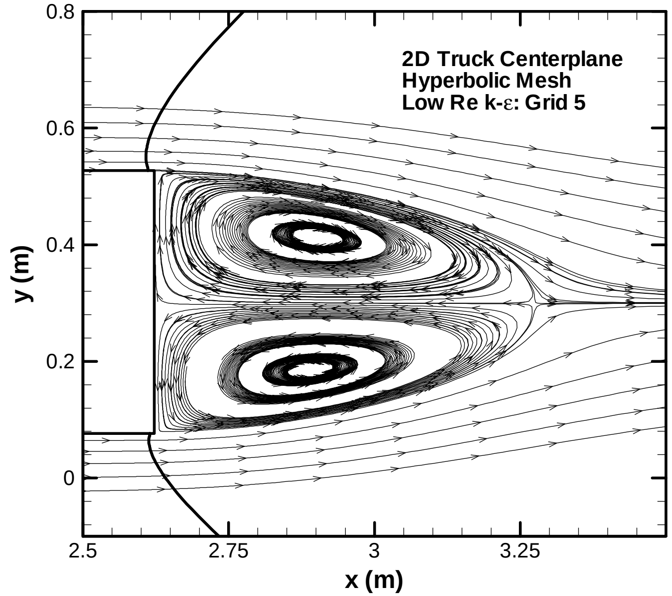


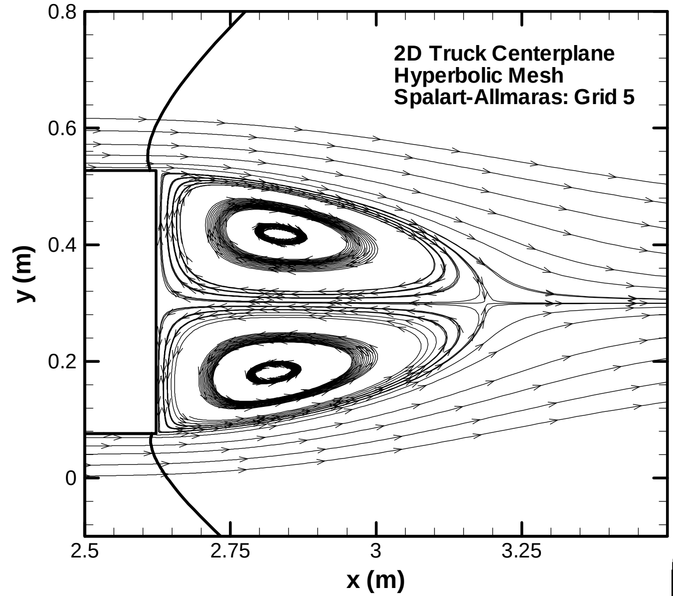

**Shear Stress**

# Model Comparison: Grid5

| | Recirculation Zone Length | Drag Force |
|---|---|---|
| Wilcox | 0.69 m | 932 N/m |
| k-ε | 0.65 m | 1027 N/m |
| Spalart | 0.56 m | 1112 N/m |



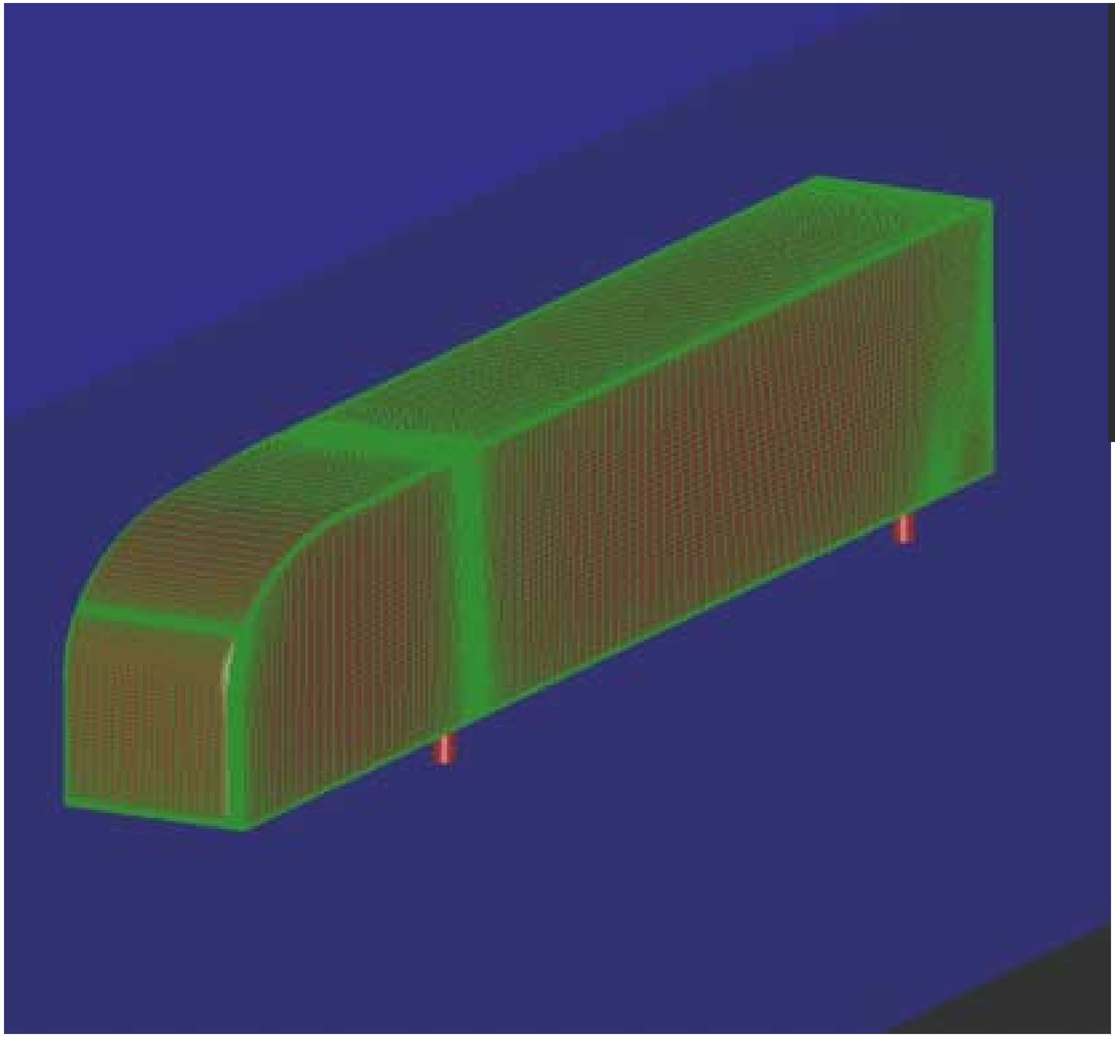



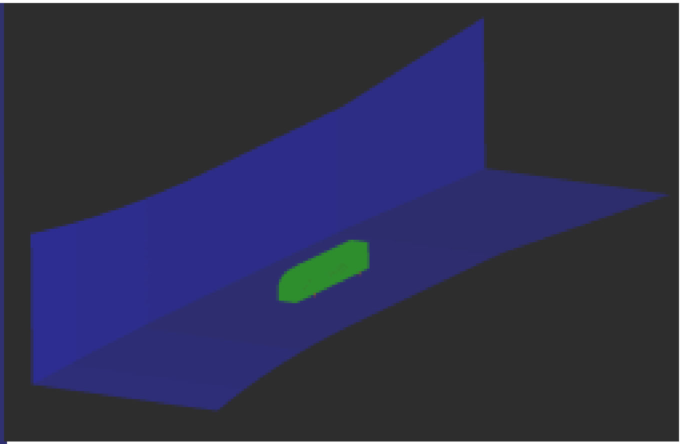



# Conclusions from GTS 2D grid studies

- Previous 3D mesh had $y^+$ too large
  - medium mesh: $y^+$ max = 10
  - coarse mesh: $y^+$ max = 20?
- New 2D hyperbolic mesh for $y^+$ study (no tunnel)
  - Wilcox k-$\omega$ will not run with $y^+ > 2$
  - k-$\varepsilon$ and S-A will run with $y^+ > 1$, but accuracy suffers
  - pressure not as sensitive to $y^+$ as shear stress
- Spalart-Allmaras predicts:
  - shorter recirculation zone
  - higher drag
- Wilcox k-$\omega$ predicts:
  - longer recirculation zone
  - lower drag

# Task 1: New 3D Grid for GTS





- New 3D meshes complete
  - Coarse (300,000 cells)
  - Medium (2.5 million cells)
  - Fine (20 million cells)
- Grid needs to be decomposed (parallel)
- Will run:
  - k-omega/Wilcox
  - Spalart-Allmaras
  - k-epsilon (time permitting)

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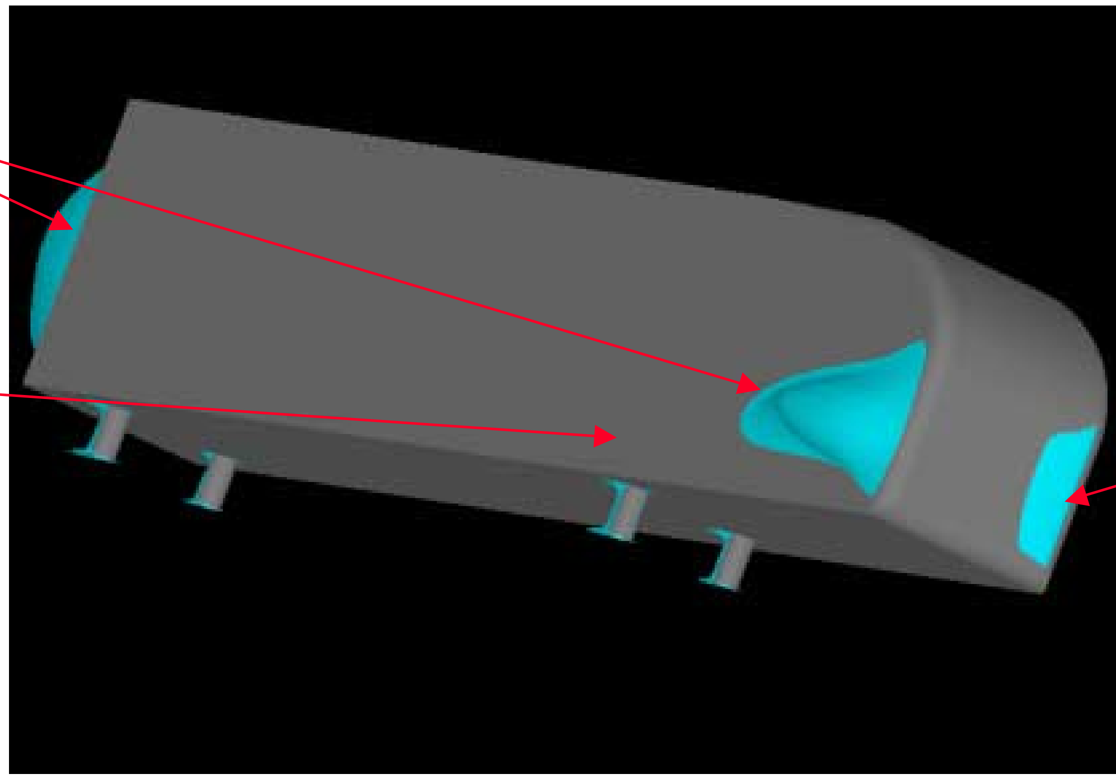
# Task 6: GTS, 10 Degree Yaw Solution Spalart-Allmaras, FY01 Medium Mesh

FY01 Medium Mesh is 12 million grid point mesh that Kambiz and Mary completed last year

Negative u-component of Velocity

Recirculation Zones

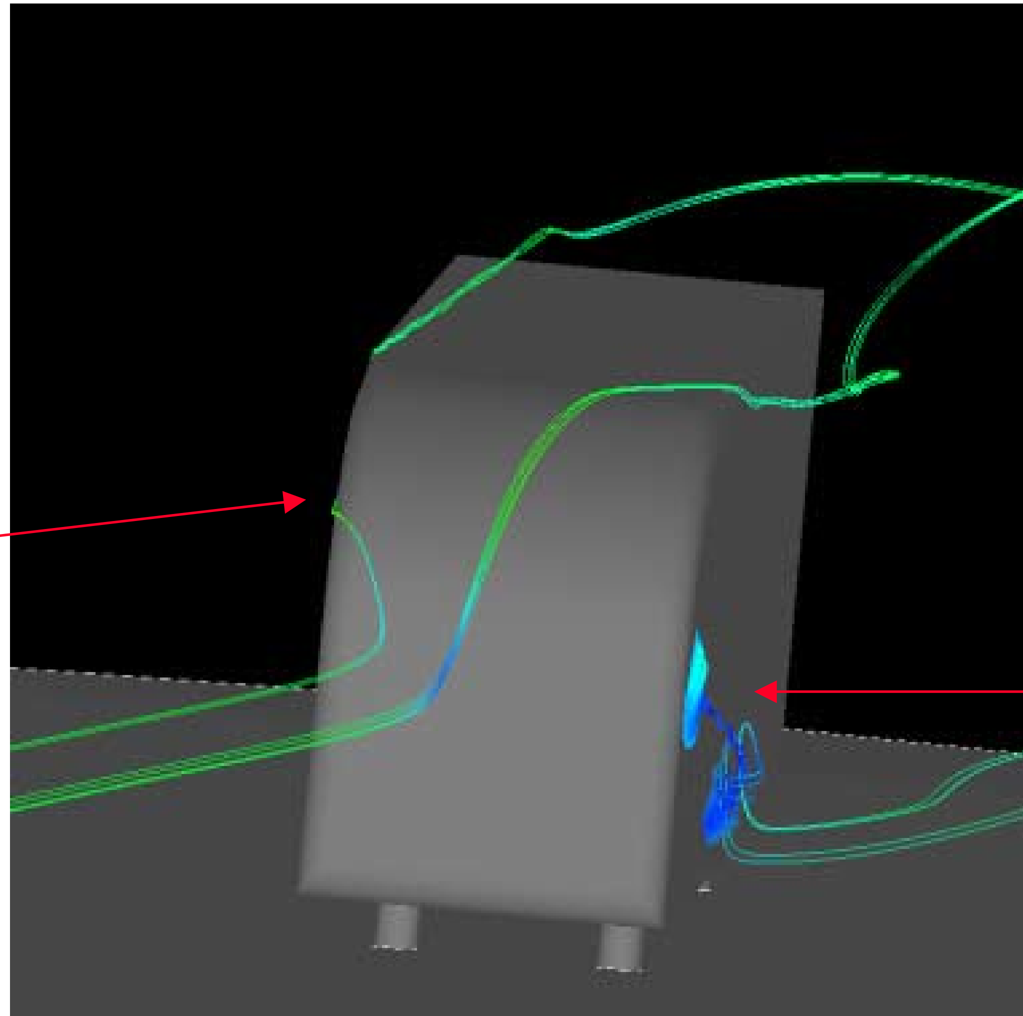
Leeside



u<0, but not a Recirculation Zone (attached flow)

# GTS: 10 Degree Yaw Solution
# Spalart-Allmaras, FY01 Medium Mesh

Windward side



Leeward side

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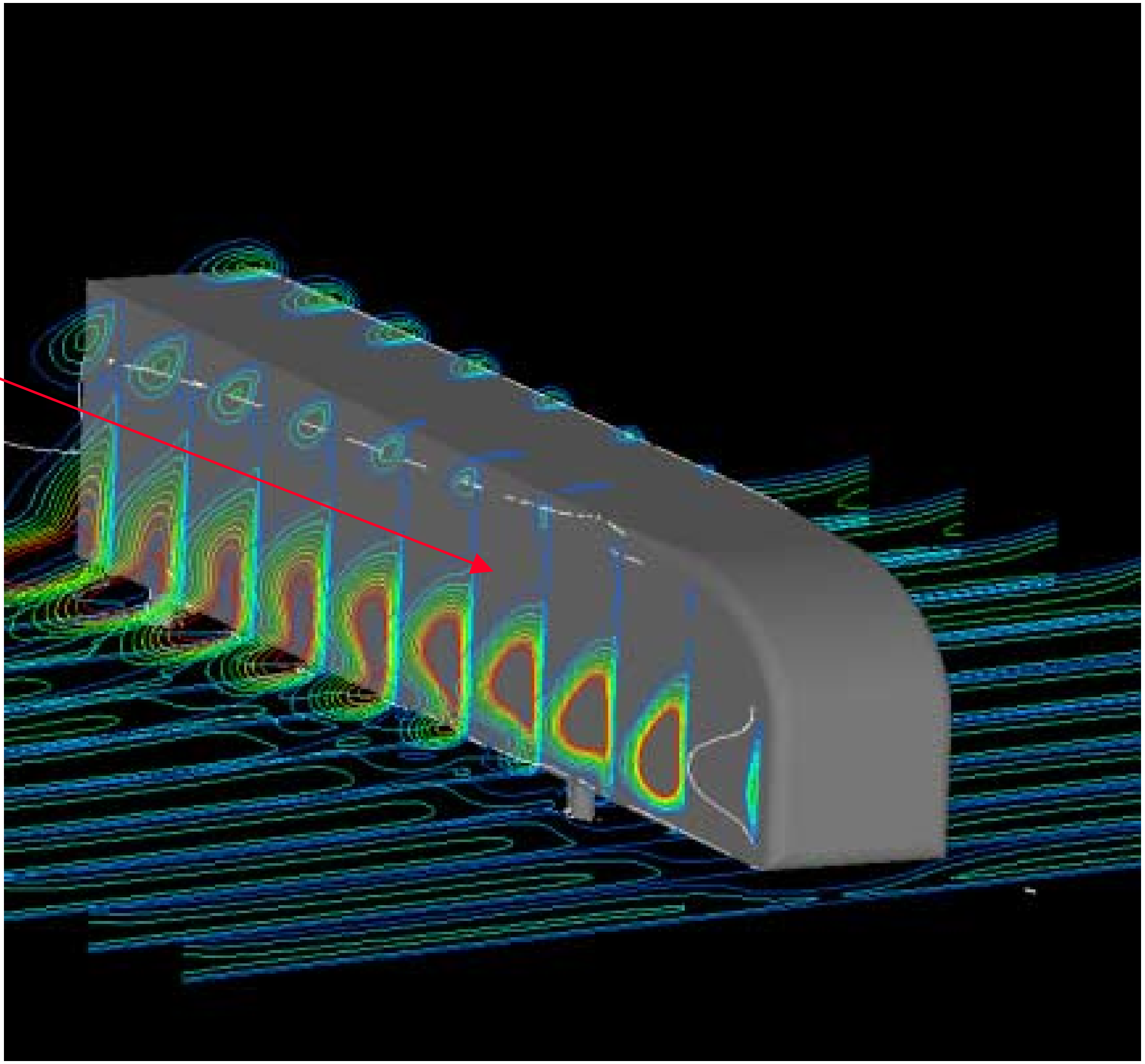
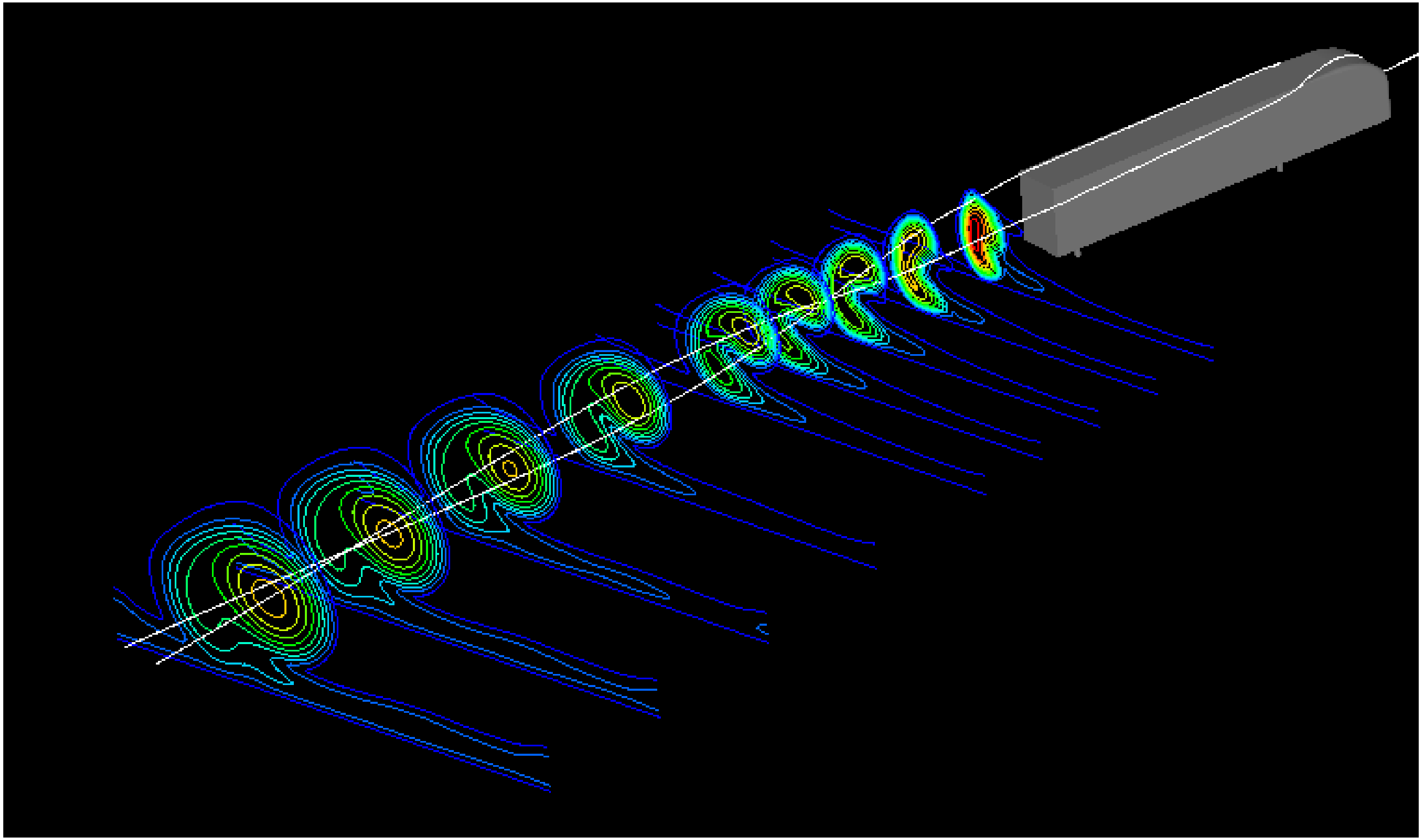
# Spalart Allmaras, 10 Yaw, FY01 Medium Mesh
# Total Viscosity and Vortex Cores

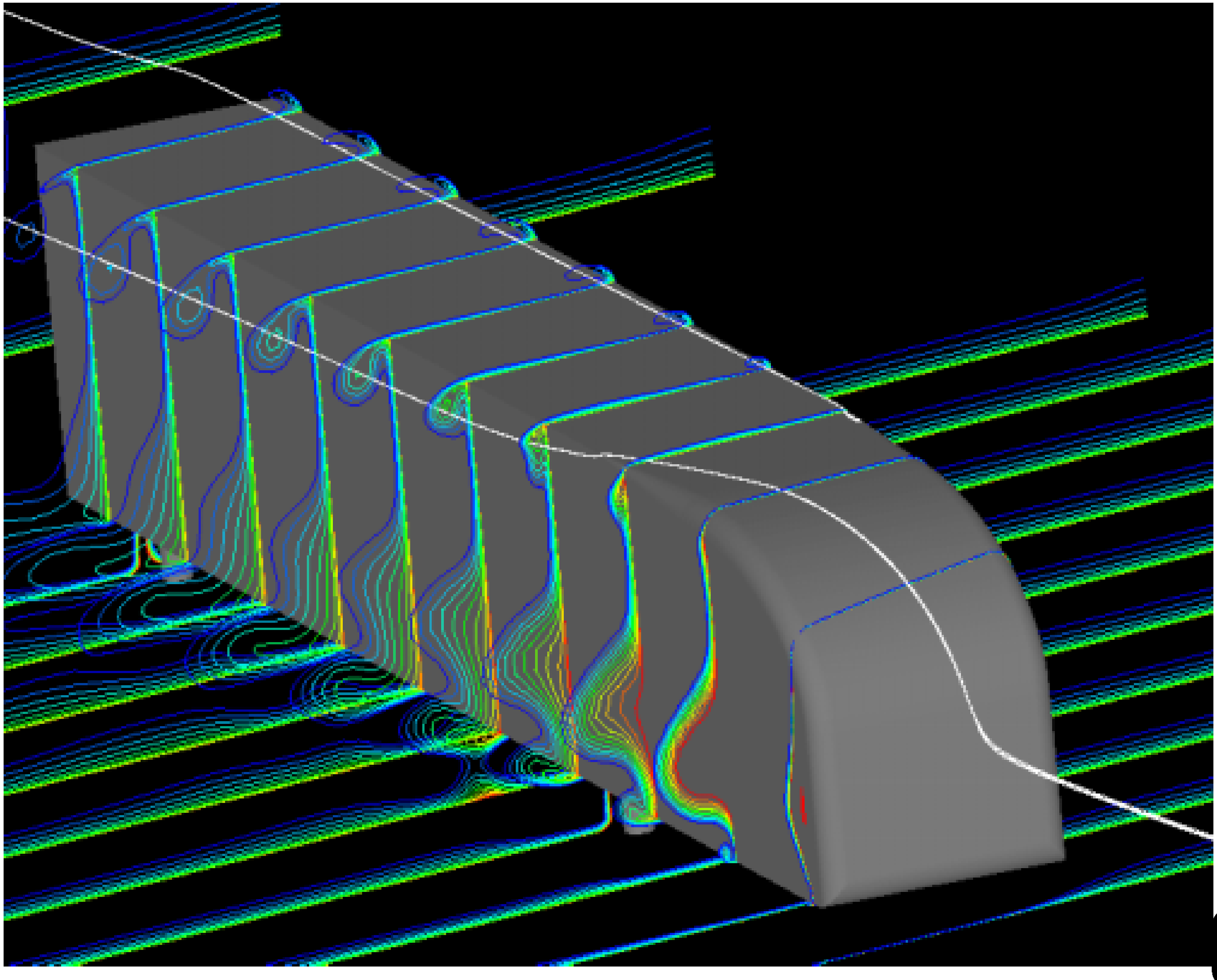
Leeside



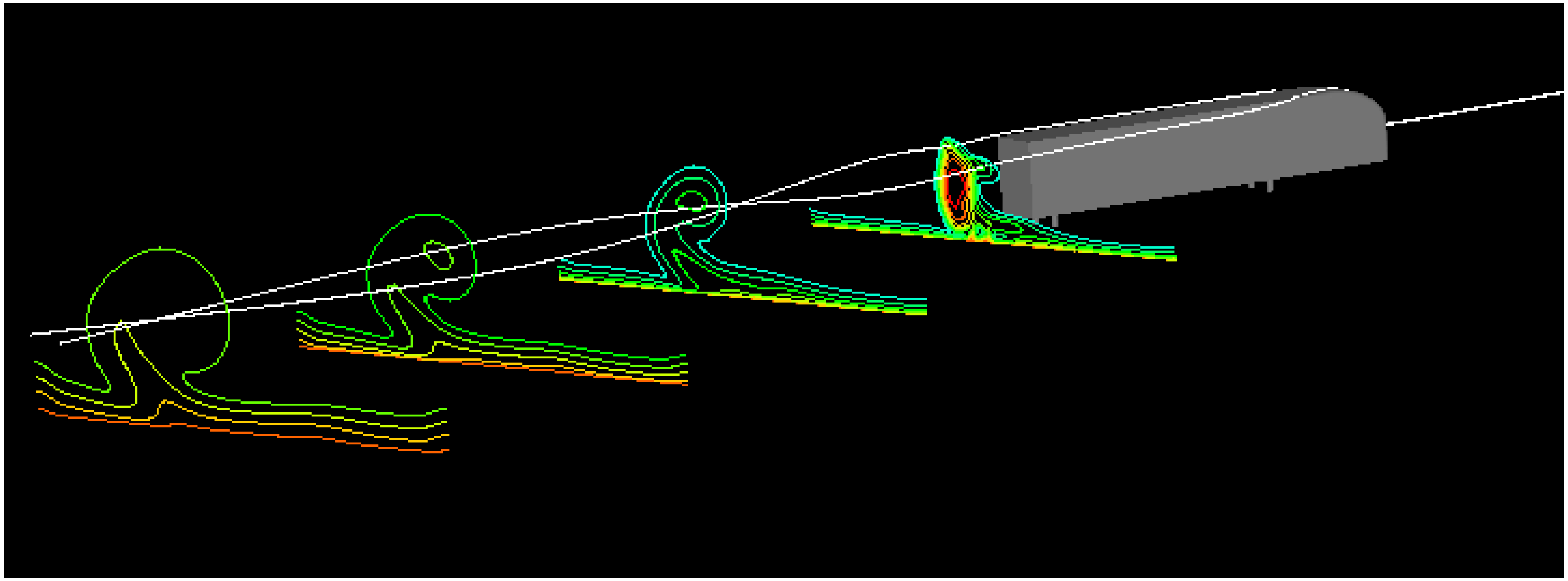
# Spalart Allmaras, 10 Yaw, FY01 Medium Mesh
# Total Viscosity and Streamlines



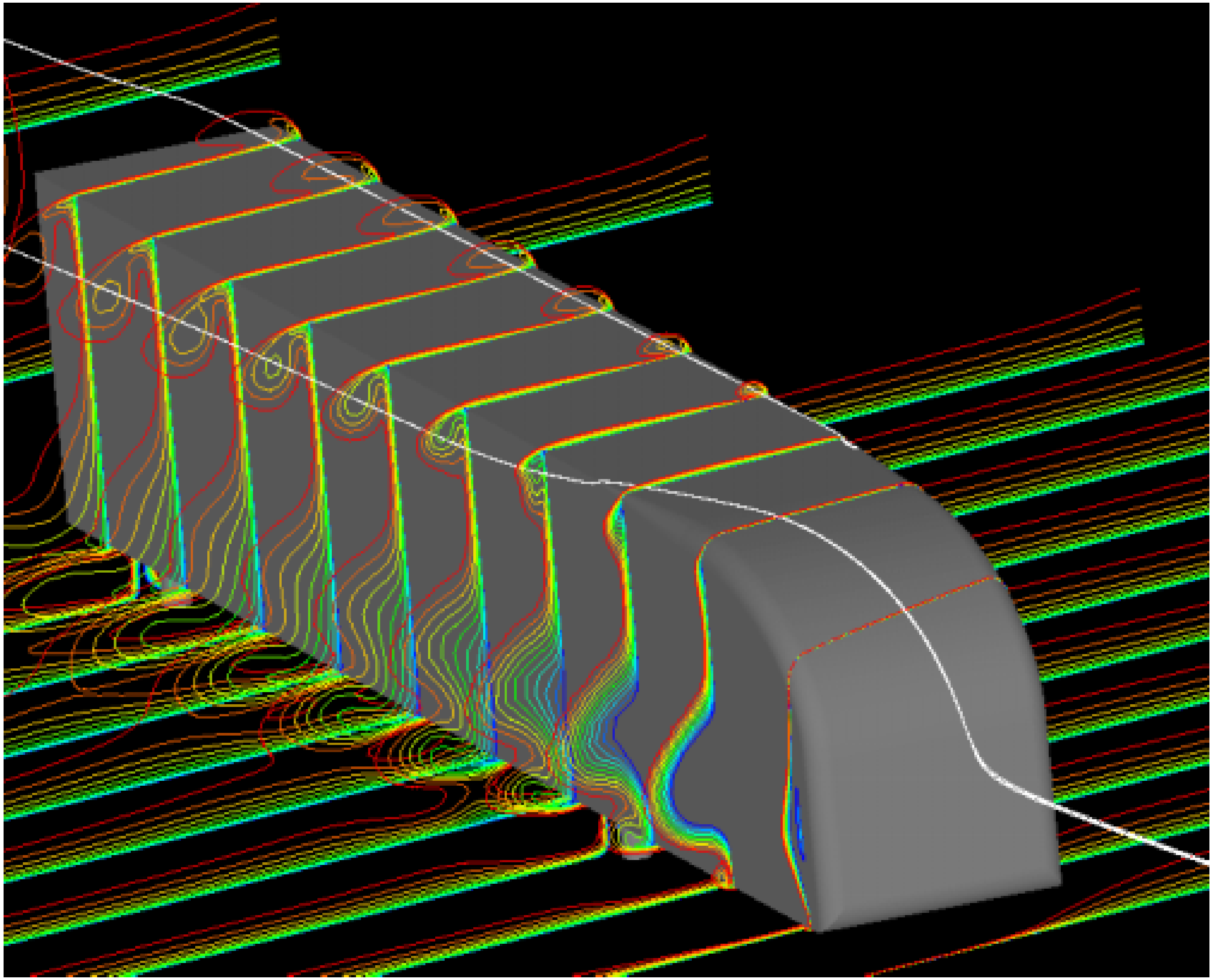
# Spalart Allmaras, 10 Yaw, FY01 Medium Mesh
# Temperature and Streamlines



# Spalart Allmaras, 10 Yaw, FY01 Medium Mesh
# Temperature and Vortex Cores



# Spalart Allmaras, 10 Yaw, FY01 Medium Mesh
# Mach Number and Streamlines



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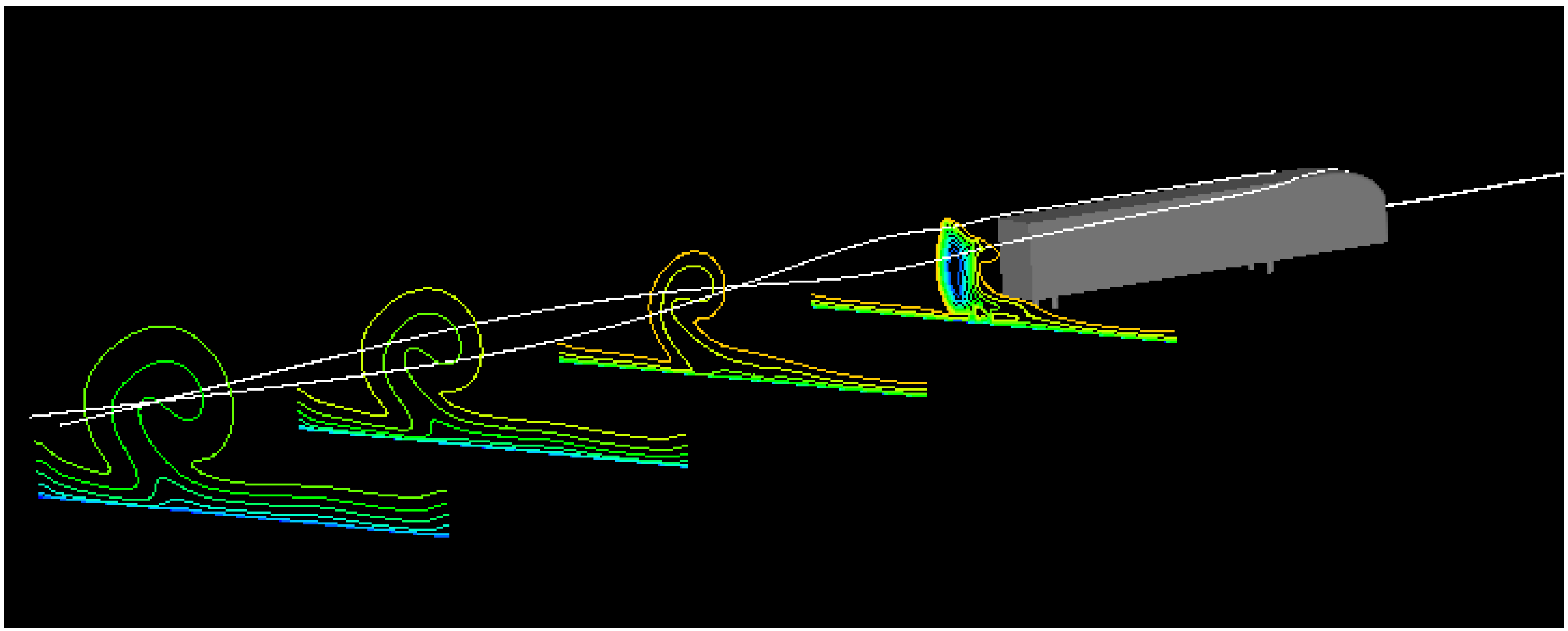
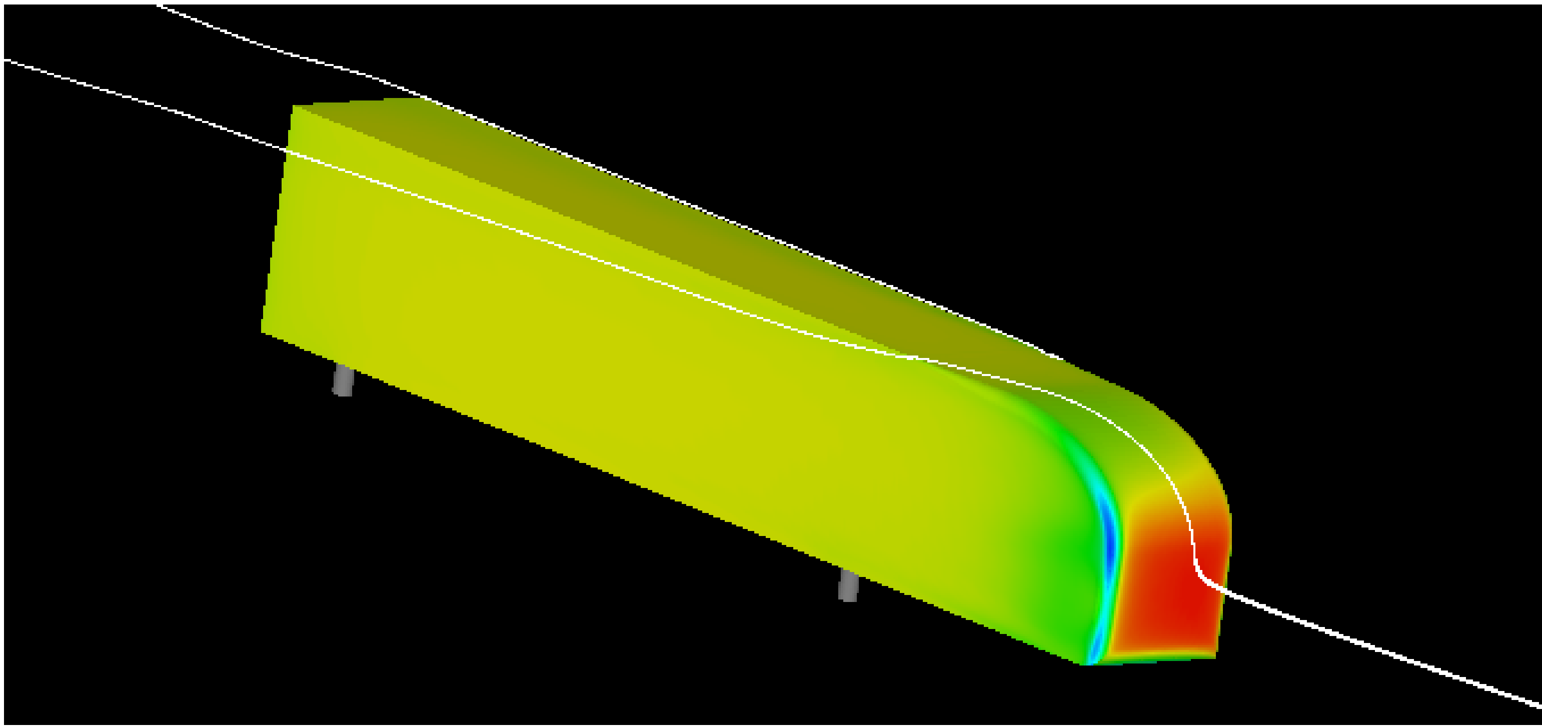
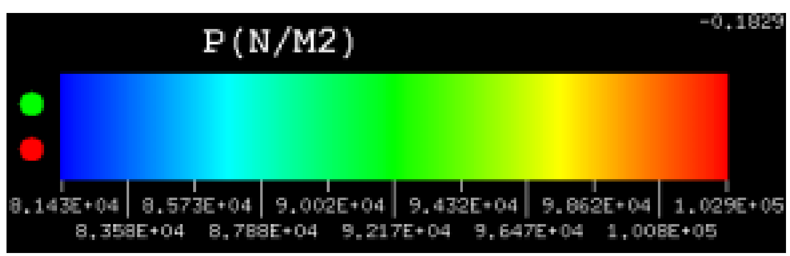
# Spalart Allmaras, 10 yaw, FY01 Medium Mesh
# Mach Number and Streamlines



# Surface Pressure



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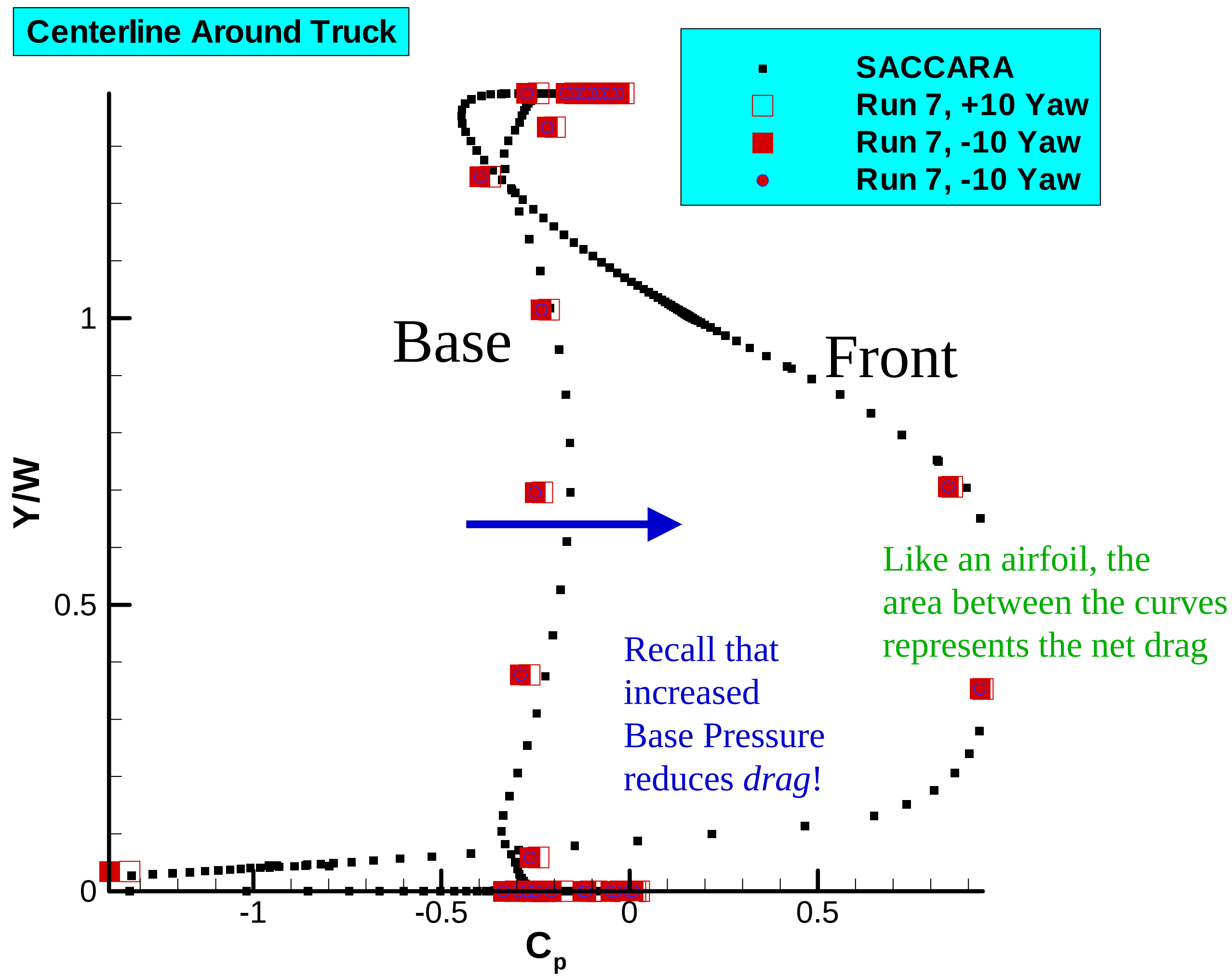
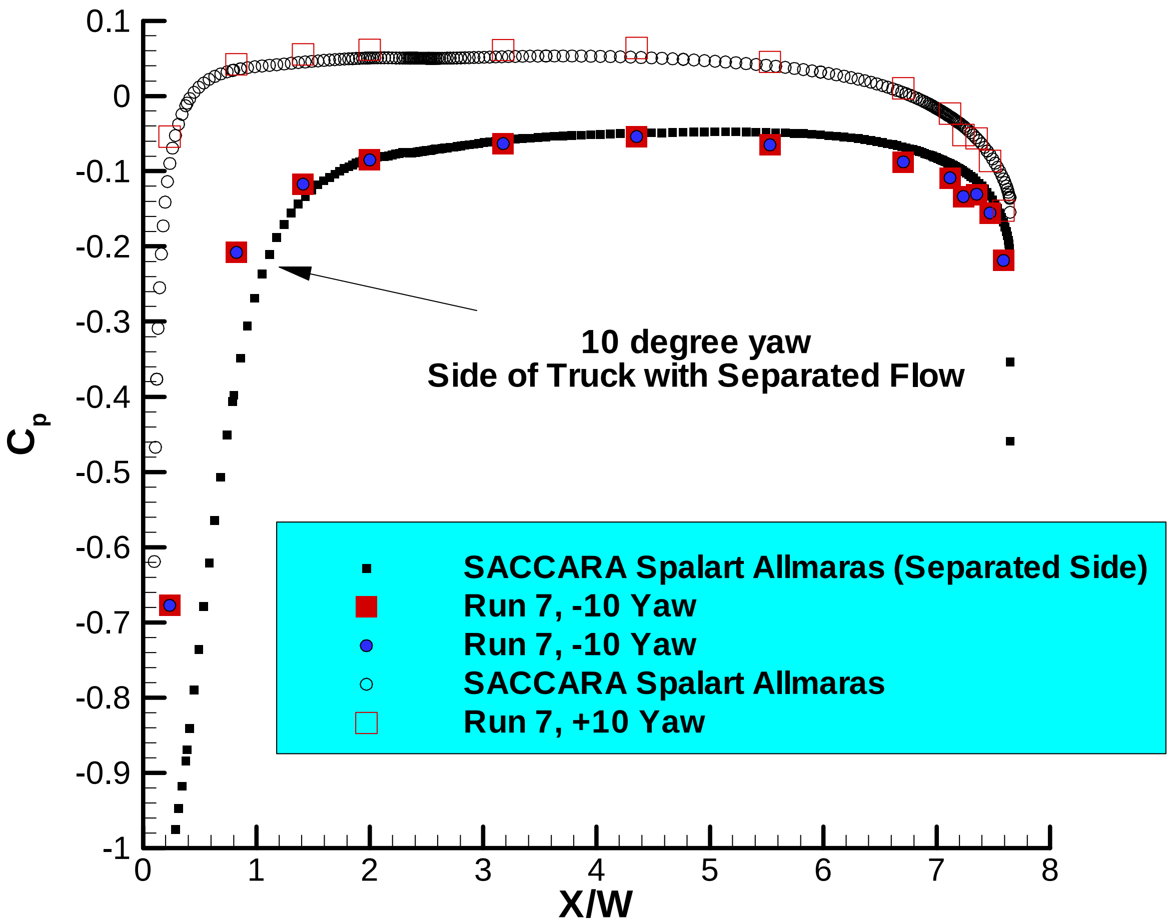
# Spalart-Allmaras, 10 Yaw, FY01 Medium Mesh, Vertical Cut



# Spalart-Allmaras, 10 Yaw, FY01 Medium Mesh, Horizontal Cut

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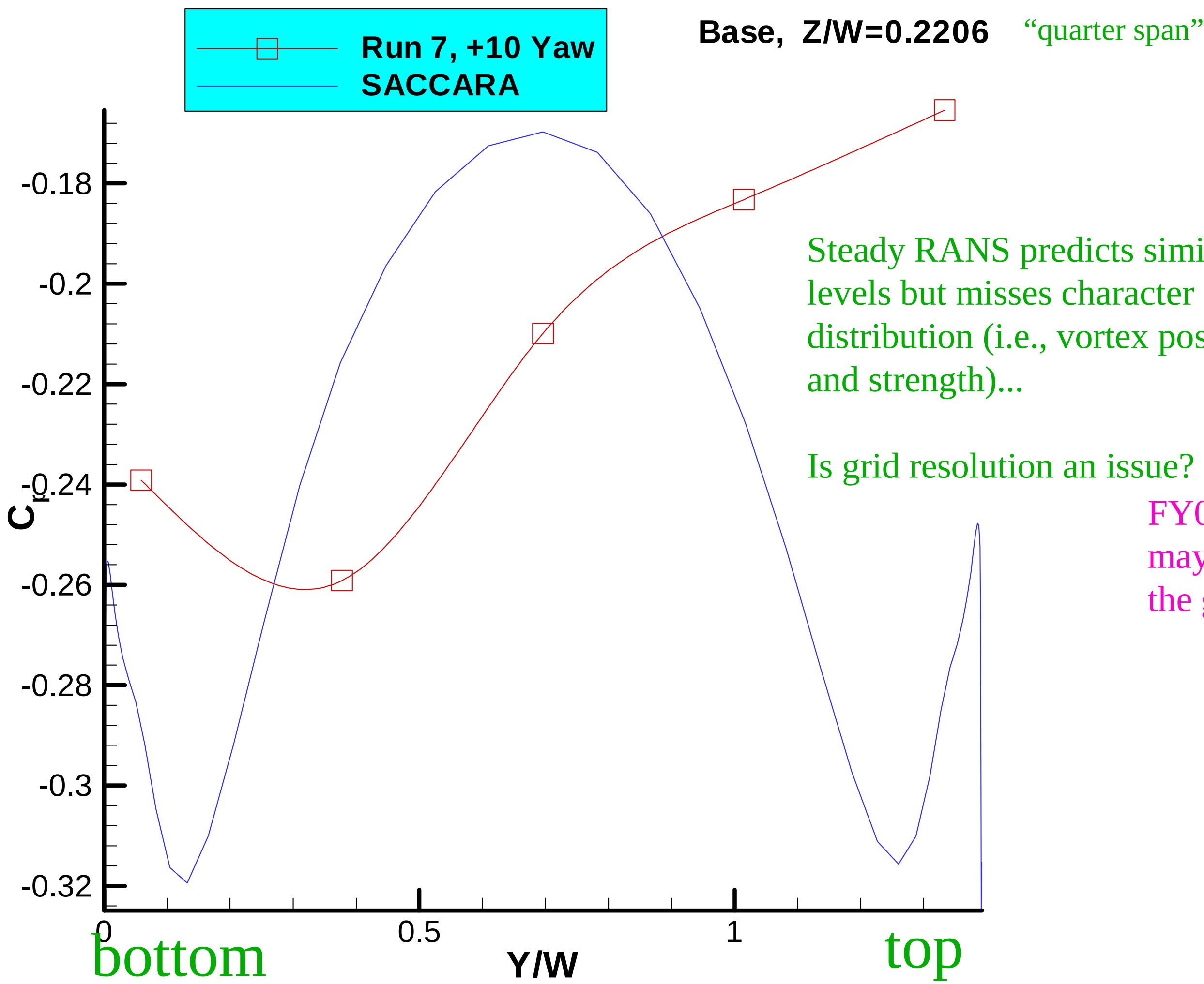
# Spalart-Allmaras, 10 Degree Yaw, FY01 Medium Mesh, Vertical Cut




Highly Expanded Scale

Steady RANS predicts similar pressure levels but misses character of distribution (i.e., vortex position and strength)...

Is grid resolution an issue?

FY02 grids may answer the grid question

# Spalart Allmaras, 10 Degree Yaw, FY01 Medium Mesh

**10 Degree Yaw**
**Wind Axis Force Coefficients**

| | | $C_D$ | $C_S$ |
|---|---|---|---|
| SACCARA | $10^o$ yaw | 0.6679 | 1.2210 |
| Experiment (Run 7) | $10^o$ yaw | 0.5055 | 1.1833 |
| Wall Reference[a] | $-10^o$ yaw | 0.5197 | -1.0865 |
| | $-10^o$ yaw | 0.5202 | -1.1039 |
| Experiment (Run 7) | $10^o$ yaw | 0.54 | 1.2640 |
| Upstream Reference[b] | $-10^o$ yaw | 0.5543 | -1.1360 |

a. Static pressure reference is measured at wall pressure tap.

b. Static pressure reference is measured upstream of test section.

Numerical $C_D$ used wall reference pressure to compute freestream dynamic pressure

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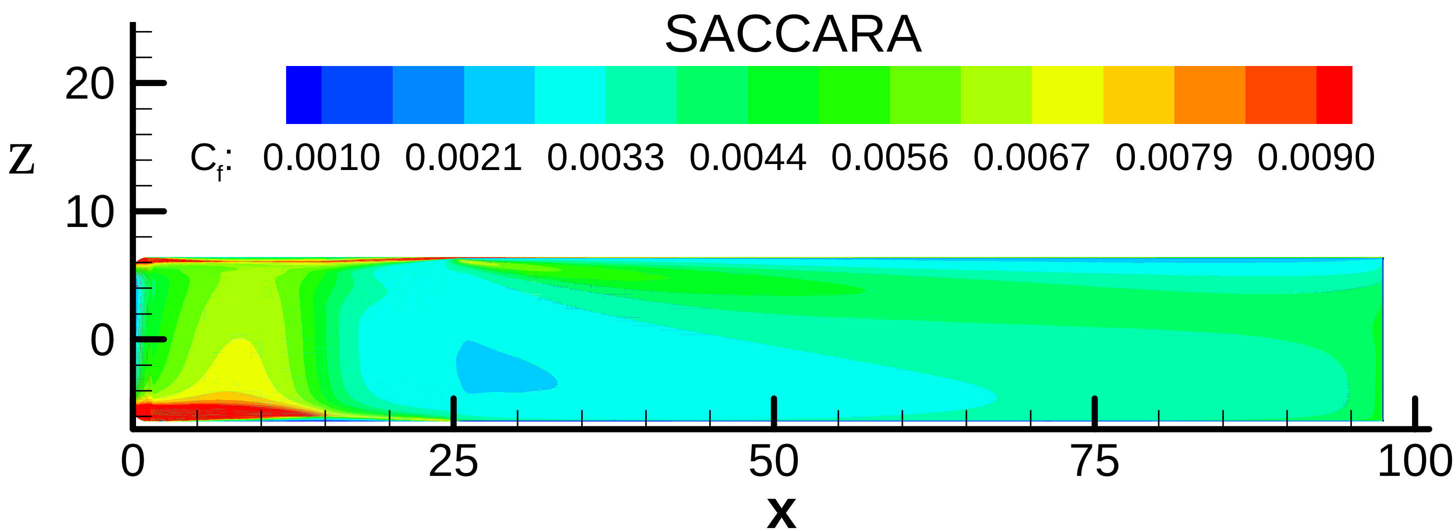
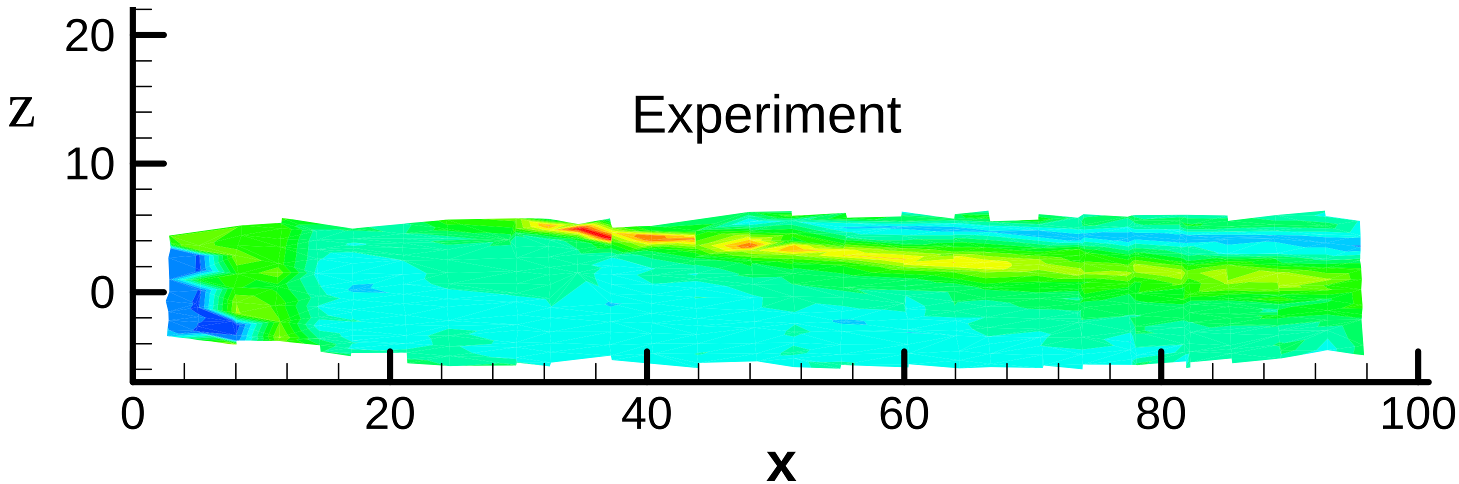
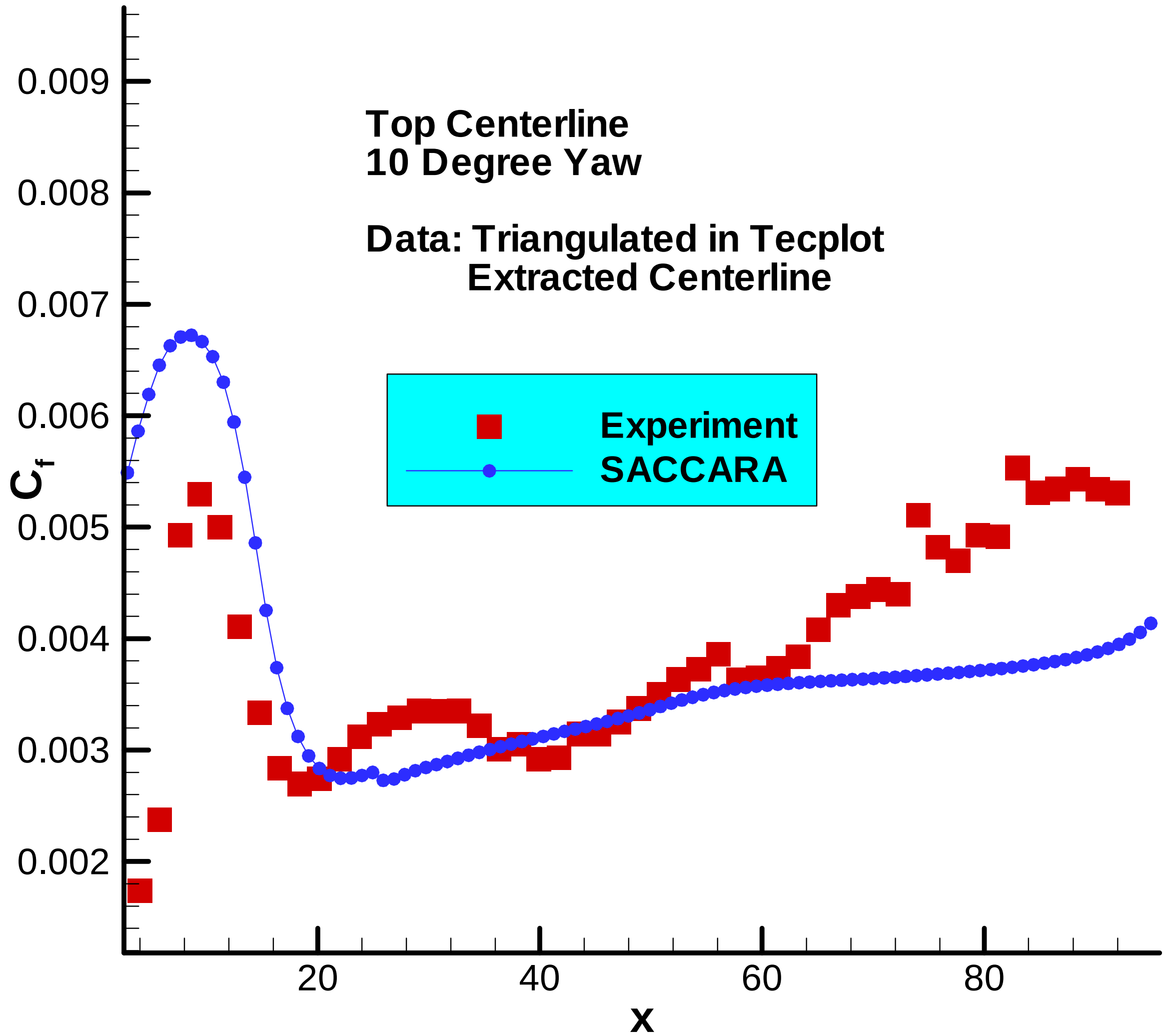
# Skin Friction on Top, 10 Degree Yaw Spalart-Allmaras Compared with Experiment



# Skin Friction on Top, 10 Degree Yaw
# Spalart-Allmaras Compared with Experiment

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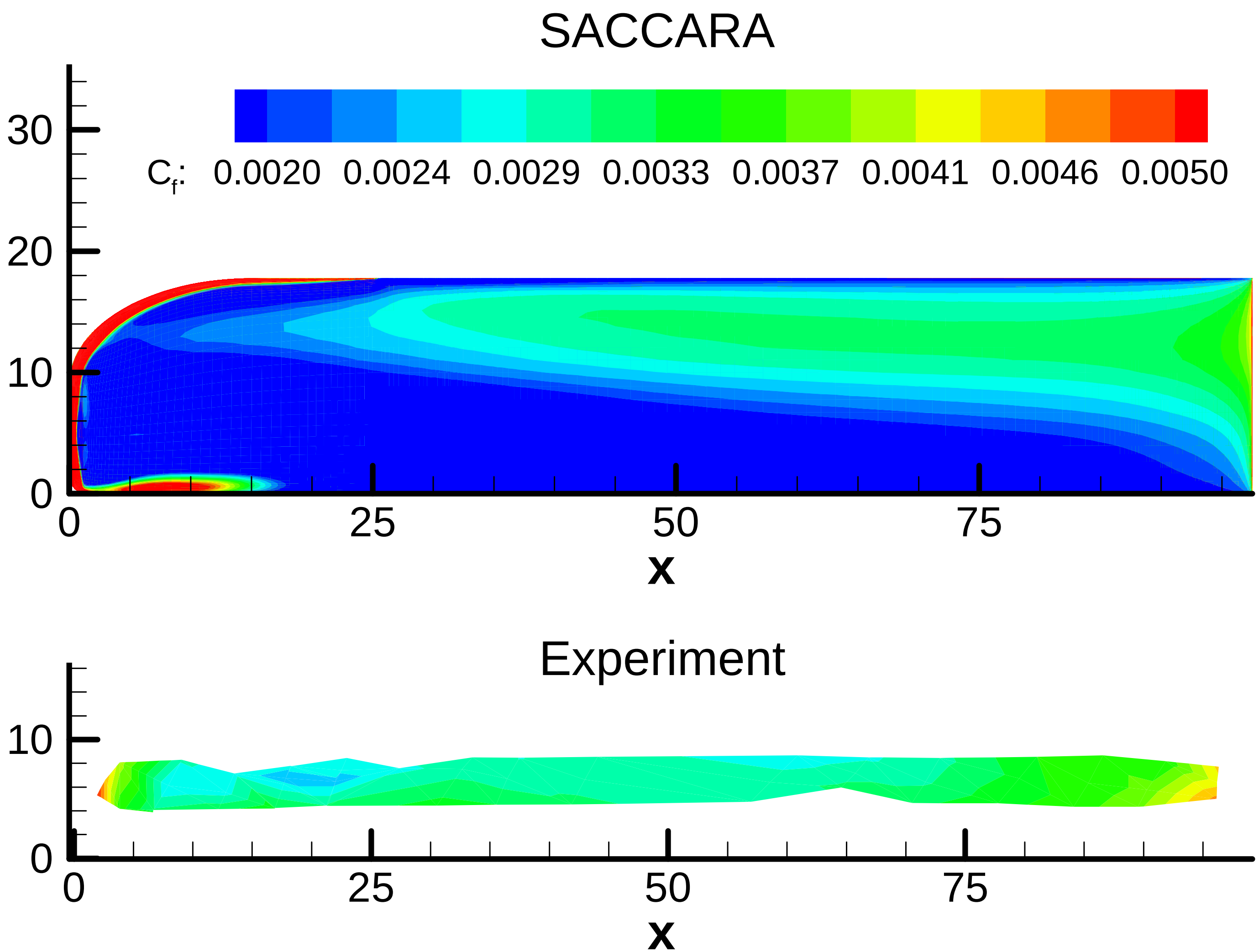
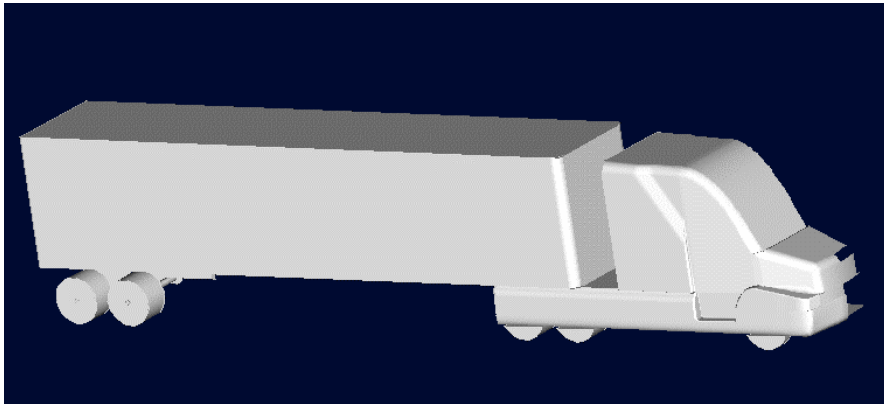
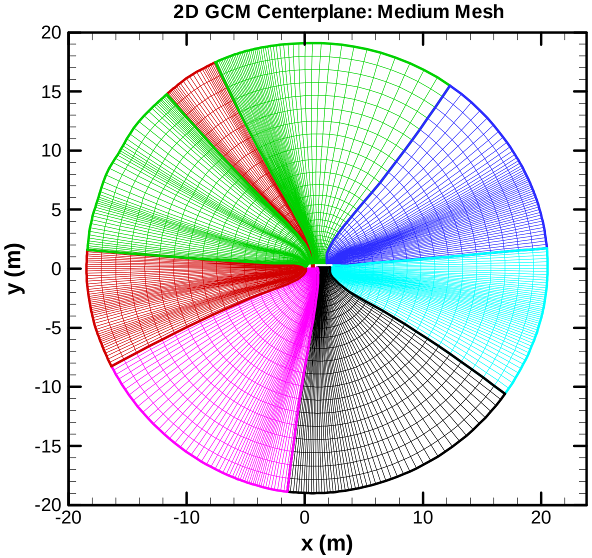
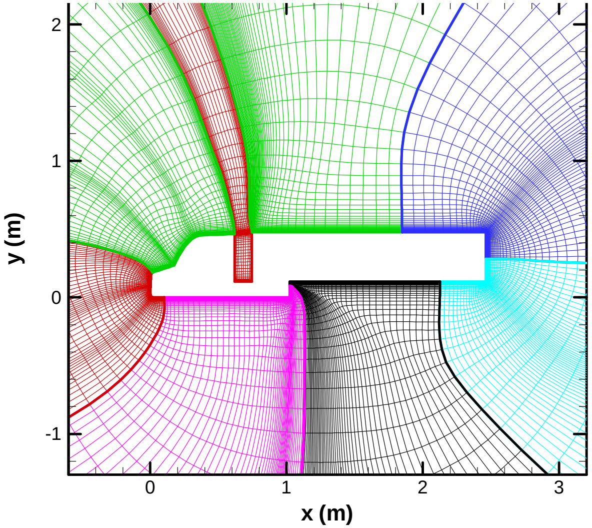
# Skin Friction on Lee Side, 10 Degree Yaw Spalart-Allmaras Compared with Experiment



# GCM: 2D Studies

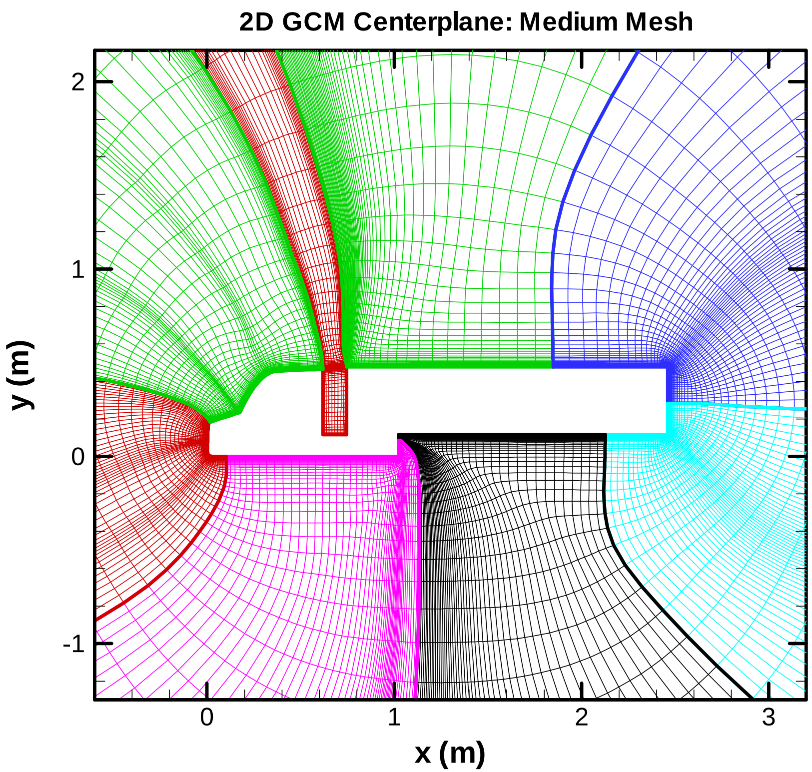




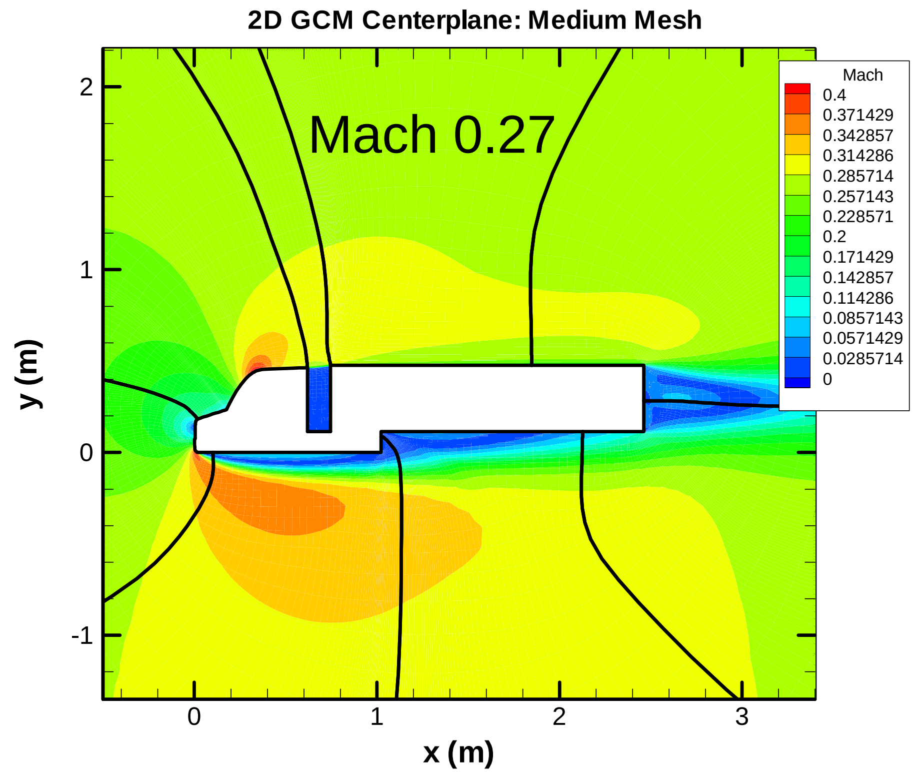




- Meshes generated (centerline cut)
- k-omega/Wilcox solution obtained
- y+ values determined

# GCM: 2D Studies




**Grid**




**Mach Contours**

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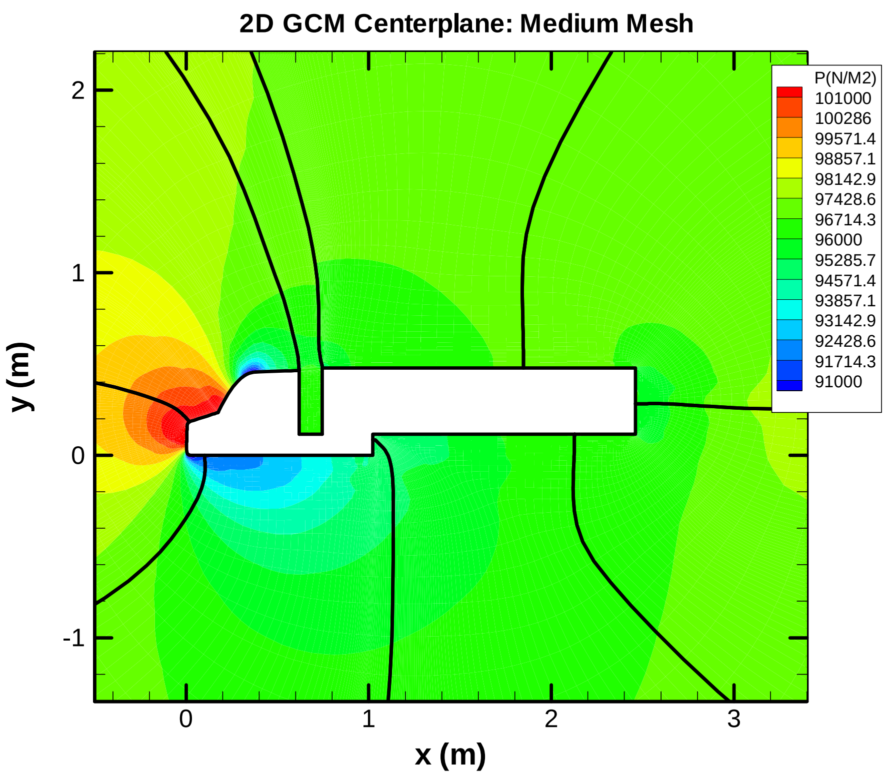
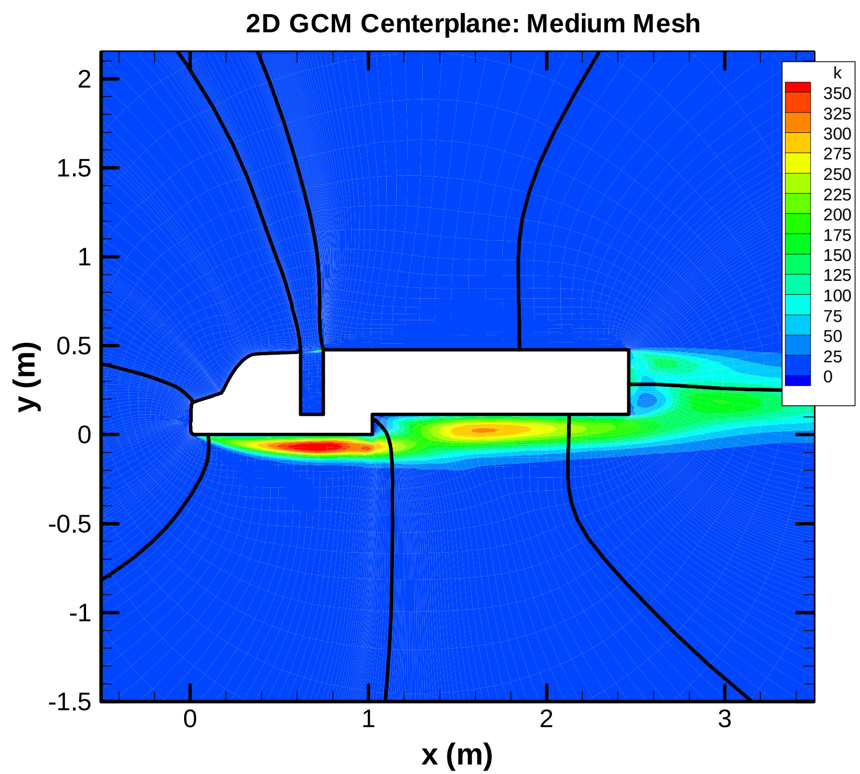
# GCM: 2D Studies

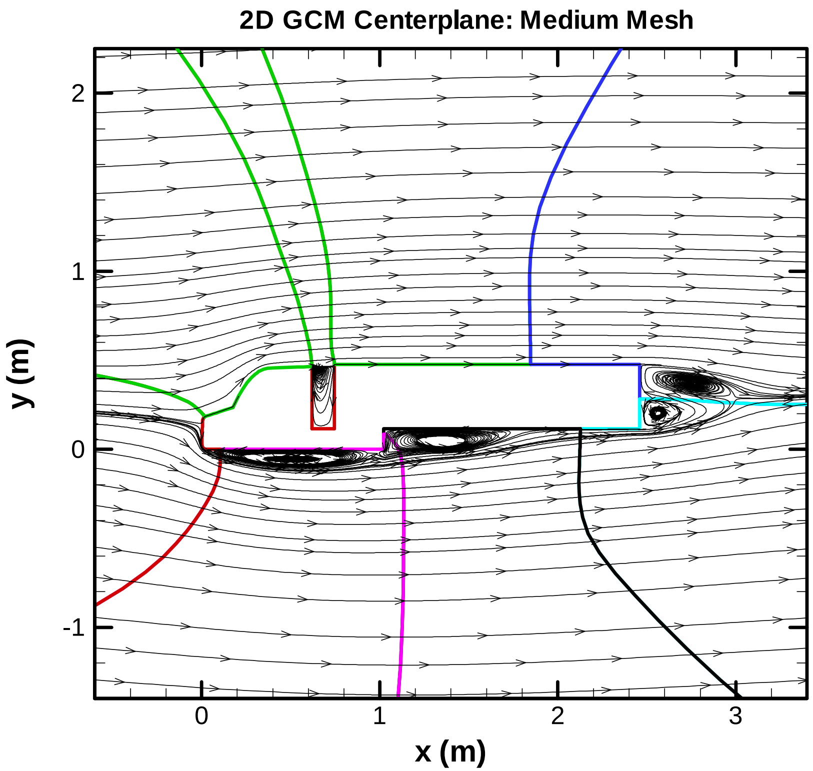


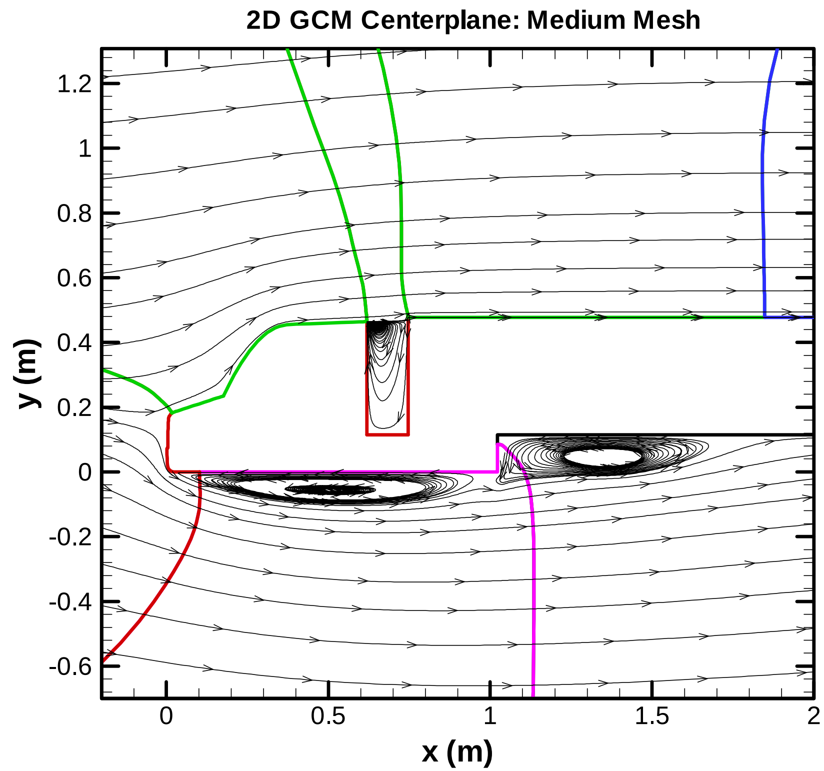

**Pressure Contours**




**TKE Contours**

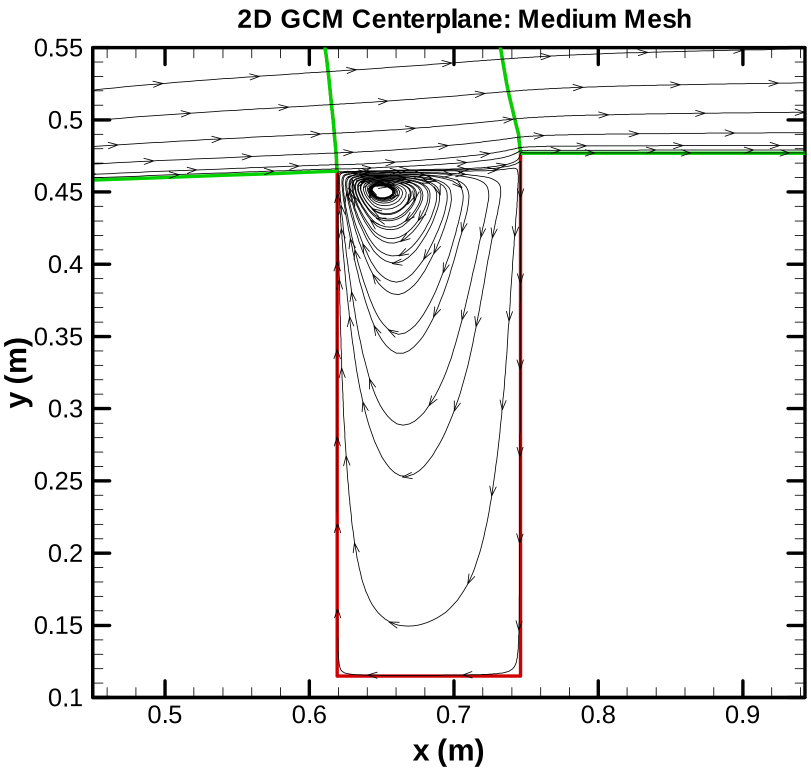
# GCM: 2D Studies

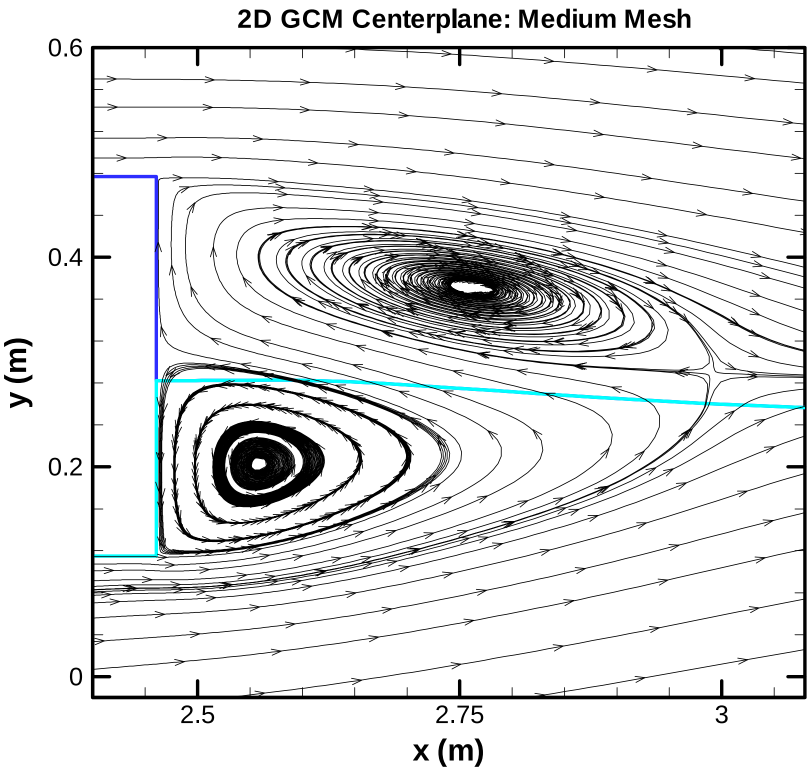



**Streamlines**

# GCM: 2D Studies




**Streamlines**

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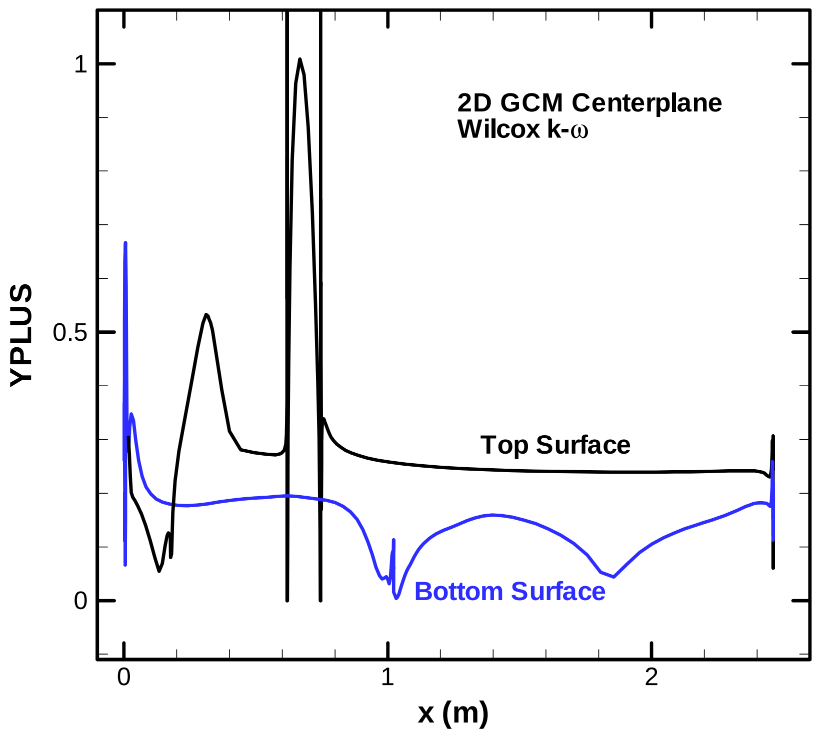
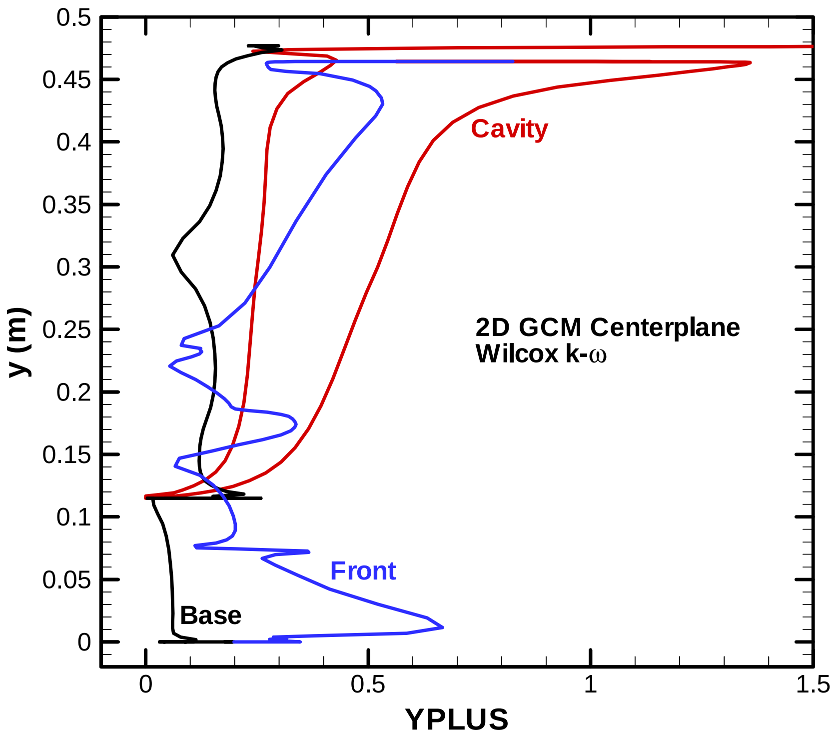
# GCM: 2D Studies

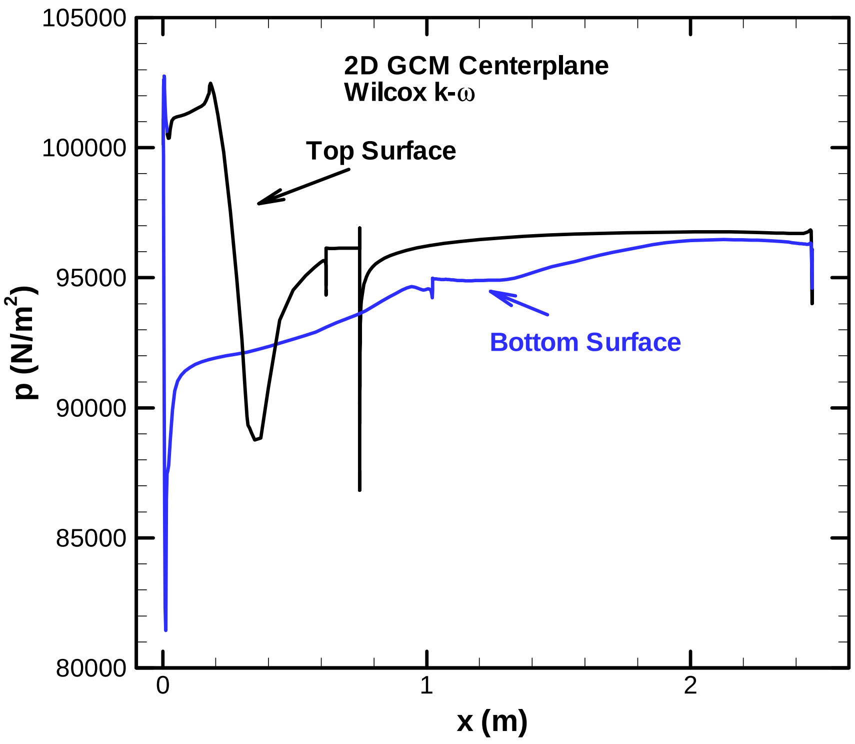


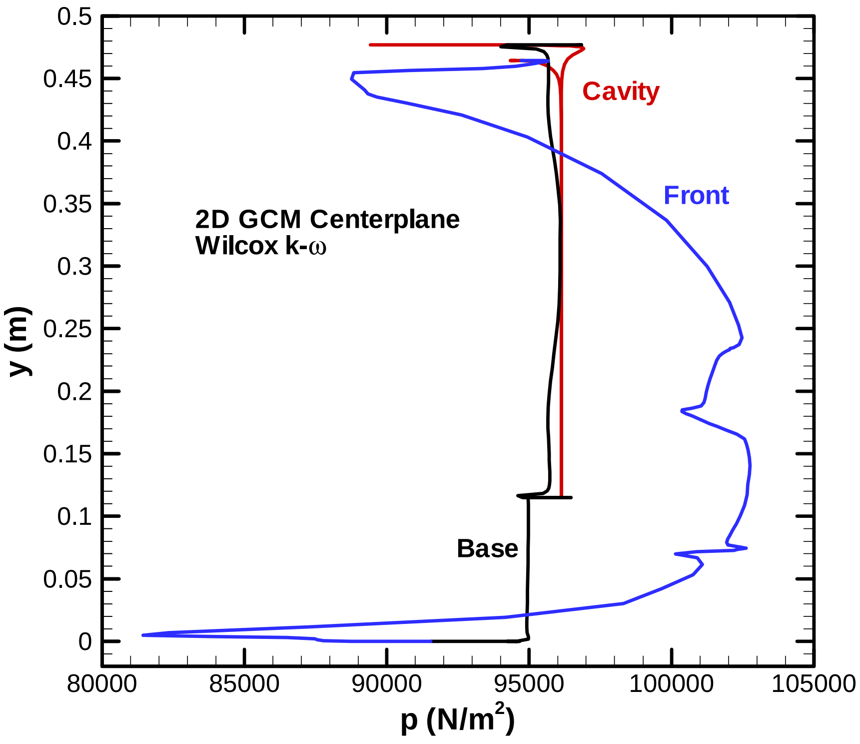

y+ (top/bottom)

y+ (front/back)

# GCM: 2D Studies

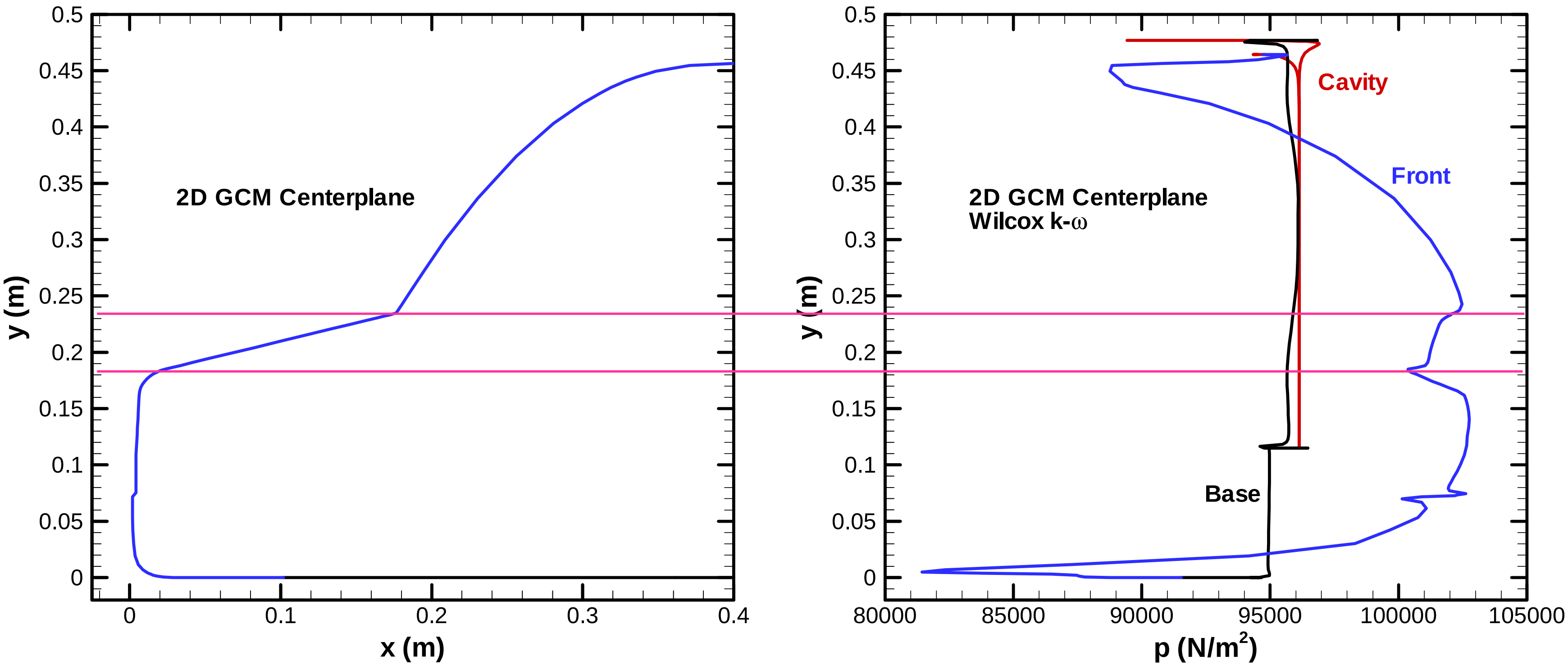



Pressure (top/bottom)



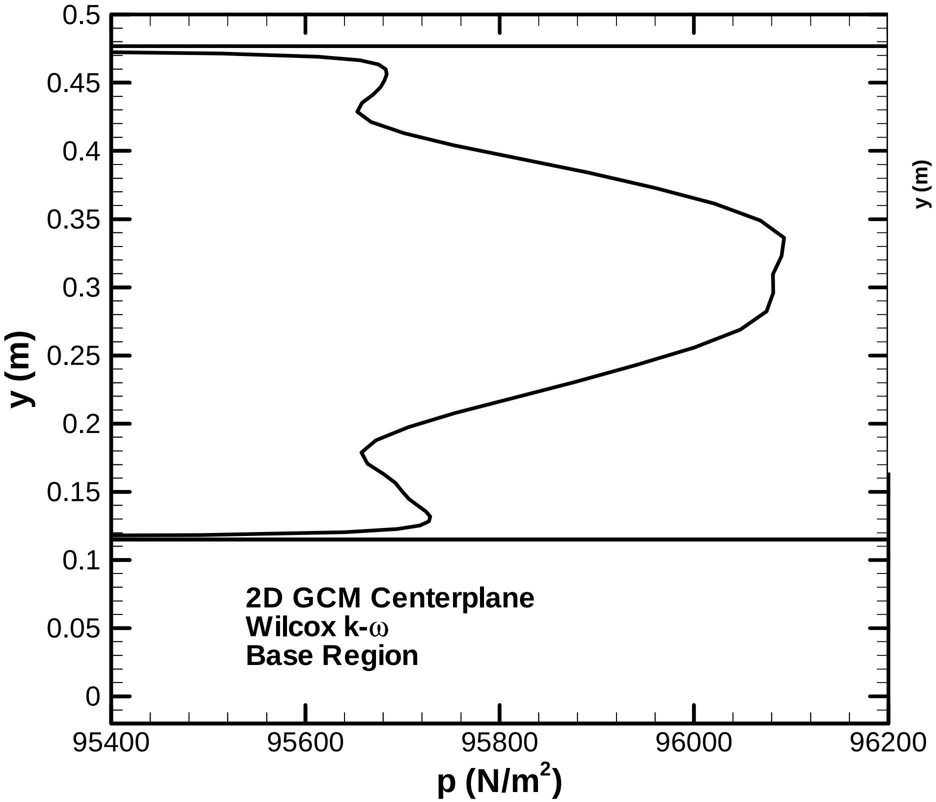

Pressure (front/back)

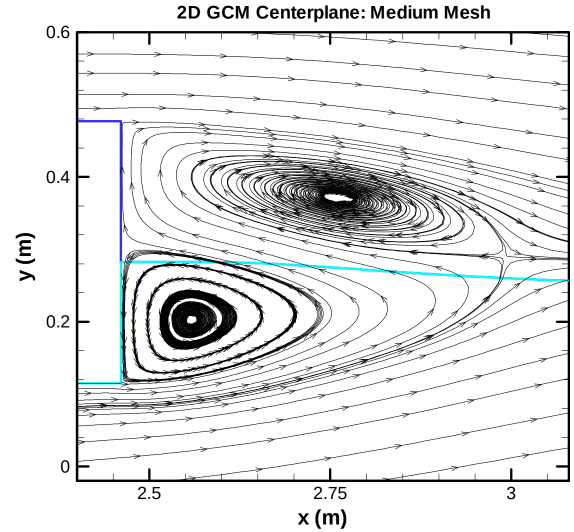
# GCM: 2D Studies



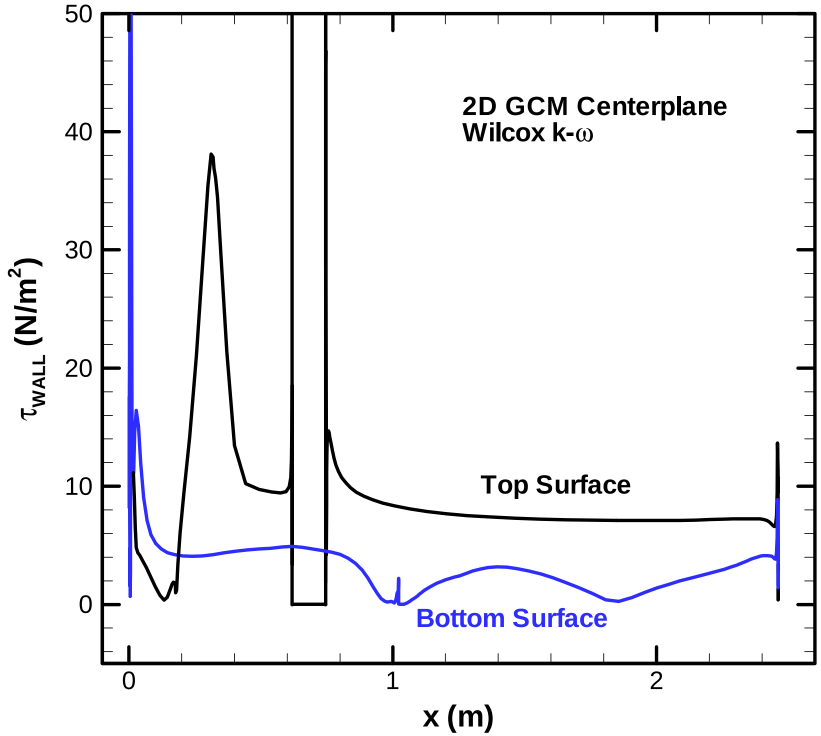

**Pressure (front/back)**

# GCM: 2D Studies

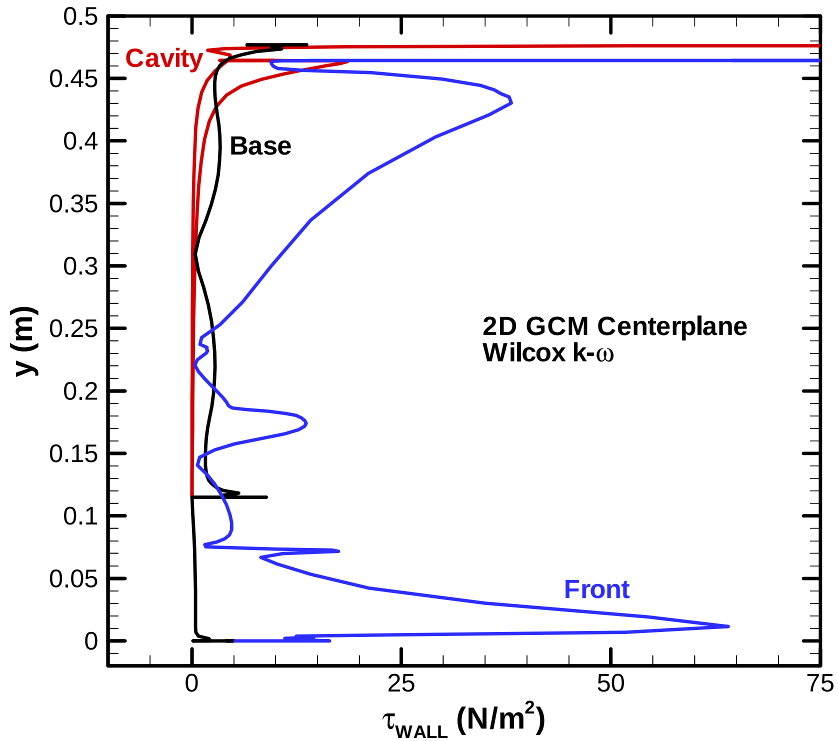


# GCM: 2D Studies




Shear Stress (top/bottom)




Shear Stress (front/back)

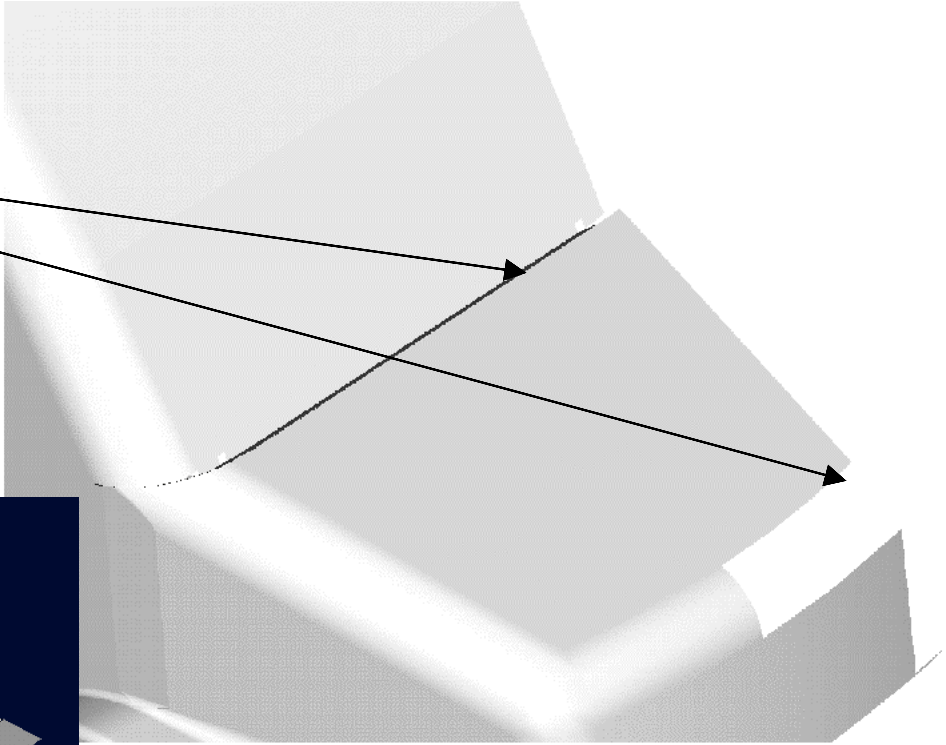
# GCM: 2D Studies

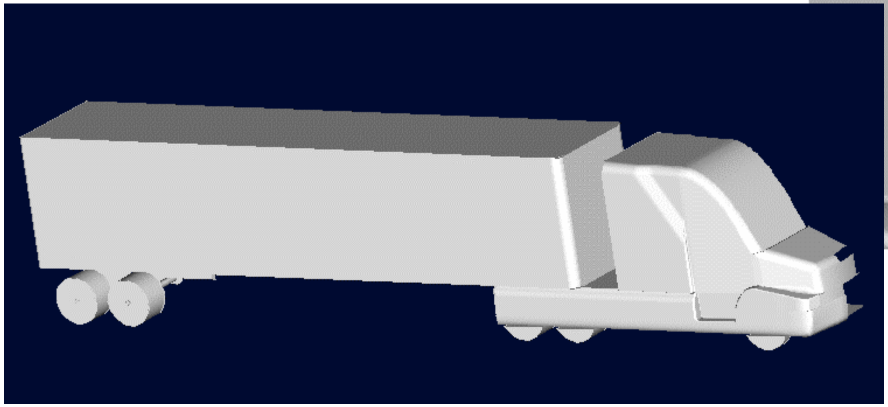
**Conclusions**

- More complex (and realistic) geometry than the GTS
- Determined appropriate wall spacing based on y+ criteria
- Significant separation on underside of truck
  - below the cab
  - below the trailer
- Underside separation (without ground plane) strongly affects the separated flow in base region
- Additional separation zone in the cab-trailer gap

# GCM: 3D

- Obtained ProE model
- Half of truck?
- Surfaces still missing
- SNL is reluctant to speculate on missing geometry surfaces





# Sandia Leveraging

- Engineering Sciences Research Foundation
  - Transition modeling
  - Hybrid RANS/LES turbulence modeling
- ASCI Material and Physical Models
  - RANS turbulence modeling
- ASCI Code Development
  - Verification and Validation methodologies/procedures
- ASCI University Alliance
  - boundary layer transition research
- ASCI Red Teraflop Computer
  - 9000 processor parallel machine
- Large dataset visualization with Parallel Visual 3
  - Bob Haimes, MIT (feature tracking)
  - data mining

# Observations

- RANS for drag prediction only makes sense if the base pressure is accurately modeled...
  - Even high fidelity, "integrate to the wall" models do not show that steady state RANS can cut it (for drag)…
    - Lower fidelity models (e.g., wall functions) designed for wall bounded flows offer no credible expectation that they better model the physics of truck base flows…
  - LES still not practical because of wall treatment
  - Hybrid RANS/LES offers a good possibility for accurate base flow prediction
- Experimental data need to be better understood and documented (NASA is doing this...)
  - Validation experiments should:
    - utilize simplified geometries (start simple and work up)
    - have well characterized freestream conditions
    - quantify uncertainties

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# Observations (continued)

- **Current RANS CFD can be used for vehicle design (e.g., airflow modification) for all surfaces (top, bottom, sides, front) except base**
  - **can optimize pressure distributions to modify aerodynamic forces for large portions of the vehicle (if used carefully)**

- **Industry needs our help connecting CFD with Aerodynamics and design (not just in terms of fluid mechanics):** ***"What does it mean to me and how can I use it in tractor/trailer design?"***

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# Conclusions and Path Forward

- Code V&V and UQ is very important (even if code applications are focused solely on design)
- Need smaller y+ values at surface to obtain accurate solutions
- May need unsteady RANS or DES to accurately predict base flow (currently not funded at SNL)
- Continue 3D GTS solutions for turbulence model study:
  - k-omega
  - k-epsilon
  - Spalart Allmaras
- Continue 3D GCM Solutions (free)
- Document, document, document!
  - 10 degree yaw solution (free)
  - 2D GTS
  - 3D GTS (FY02 Grids)
  - 2D GCM (free)

# Overview of LLNL Incompressible Flow Modeling and Development


Dora Yen Nakafuji, Jason Ortega,
Tim Dunn, Kambiz Salari, Rose McCallen

**Lawrence Livermore National Laboratory**


**Heavy Vehicle Aerodynamic Drag Working Group Meeting**
**April 3-4, 2002**

Lawrence Livermore National Laboratory


This work was performed under the auspices of the U.S. Department of Energy by the University of California, Lawrence Livermore National Laboratory under Contract No. W-7405-ENG-48.


## LLNL Project Goals

**Focus**

- **To provide industry with guidance on advanced computational methods and industry tools**
- **To identify and develop simulation techniques that can accurately predict the flowfield of heavy vehicles**
- **To investigate drag reduction strategies**

**Approach**

- **Investigate advanced simulation techniques using in-house tools that provide flexibility and access to internal resources**
- **Investigate flow structure associated with heavy vehicle aerodynamics such as gap flow and the wake**
- **Investigate feasibility of other available codes to aid industry**

## LLNL Budget for FY02

- FY02 $440 K
  - Project management
  - Engineering Foundation Conference
- Leveraging
  - ASCI code development program
    - Incompressible flow model development
  - ASCI White massively parallel computer
  - DoD/DOE Technology development program
    - Multiphase flow model development
  - LLNL Internal Tech Base Funding
    - Particle flow model development
  - NASA Ames collaboration
- Team Members
  - Dora Yen Nakafuji, Jason Ortega, Tim Dunn, Kambiz Salari, Rose McCallen

## LLNL FY02 Tasks

- Code speed up
  - Implicit/Semi-Implicit Projection methods
- Gap flow simulation
  - Stable flow structure with/without side extenders, low drag
  - Unsteady flow structure, high drag
  - Experimental data from USC and NASA
- Trailer wake simulation
  - Analysis of flow structure with/without boattail
  - Wake/Ground-plane interaction
  - Experimental data from NASA
- Full vehicle simulation with OVERFLOW
  - Tunnel simulation to determine proper outflow BC
  - GCM flow simulation in the NASA 7'x10' tunnel

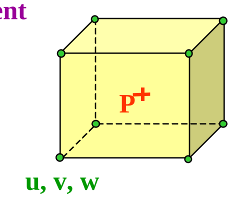
## LLNL Anticipated Deliverables for FY02

| FY02 Tasks | Oct | Nov | Dec | Jan | Feb | Mar | Apr | May | Jun | Jul | Aug | Sep |
|---|---|---|---|---|---|---|---|---|---|---|---|---|
| **1. Project Management** | | | | | | | | | | | | |
| 1A. Reports, meetings, admin support, etc. | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ |
| 1B. Industry collaborations | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ |
| 1C. Engınerring Foundation Conference | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ |
| **2. Technical Effort** | | | | | | | | | | | | |
| 2A. Simulation/Analysis using ALE3D | | | | | | | | | | | | |
| > Empty tunnel | ■ | | | ■ | ■ | ■ | | | | | | |
| > GTS flow simulation, LES, 0 yaw, M=0.27 | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ |
| > Base drag with and without boattail plates, LES | | | ■ | ■ | ■ | ■ | ■ | ■ | | | | |
| > Gap flow simulation, LES | | | | | | ■ | ■ | ■ | ■ | ■ | ■ | |
| 2B. Simulation/Analysis using OVERFLOW | | | | | | | | | | | | |
| > Benchmark | ■ | | | | | | | | | | | |
| > Empty tunnel | ■ | ■ | ■ | | | | | | | | | |
| > GTS flow simulations, *k*-? turbulence model, 0 yaw, M=0.27 | | | | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ |
| > GCM flow simulations, *k*-? turbulence model, 0 yaw | | | | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ |
| 2C. Process/Analysis of NASA GTS and GCM data | | | | ■ | ■ | ■ | | | | | | |
| 2D. Document SNL RANS results on GTS | ■ | ■ | ■ | | | | | | | | | |
| **3. Research and Development on ALE3D** | | | | | | | | | | | | |
| 3A. Benchmarks | ■ | ■ | ■ | ■ | ■ | | | | | | | |
| 3B. Turbulence modeling, LES van Driest damping, and DES | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ |
| 3C. Verification | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ |
| 3D. Speed/accuracy enhancements | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ | ■ |

## Solving 3-D Unsteady Incompressible Navier-Stokes Equations, ALE3D

**Galerkin Finite-Element Method, Q1Q0 Element**

- **8-node Hexahedral Brick Elements**
- **Tri-linear Velocity**
- **Piecewise Constant Pressure**
- **Explicit formulation**




**Implemented Implicit/Semi-Implicit projection methods to remove stability constraint on time step due to Courant and viscous restriction**

## Incompressible Flow Code Development

**Implicit Projection Method (Tim Dunn)**

**Step 1: Approximate a pressure field**

**Initialize pressure from the previous time-step**

$$\tilde{P} = P^n$$

**Step 2: Solve momentum equations for the intermediate velocity field**

$$\left[M + \Delta t\left(K + N(u^*)\right)\right]\tilde{u} = Mu^n + \Delta t\left[F - MM_L^{-1}C\tilde{P}\right]$$

**Step 3: Project to a divergent-free field**

$$\left[C^T M_L^{-1} C\right]\lambda = C^T\tilde{u} \qquad u^{n+1} = u^n - M_L^{-1}C\lambda$$

**Step 4: Update pressure**

$$p^{n+1} = p^n + \lambda / \Delta t$$

## Timing the Projection Method

**Two-dimensional wake simulation**

- **20,000 elements**
- **16 processor on IBM SP2 machine**

| | Explicit | Semi-Implicit |
|---|---|---|
| Time Step *(s)* | 3.9e-8 | 1.7e-5 |
| run time/cycle *(s)* | 3.24 | 4.77 |
| 1 second of simulation time | 961.5 days | 3.2 days |

**Semi-Implicit is about 300 times faster than explicit**

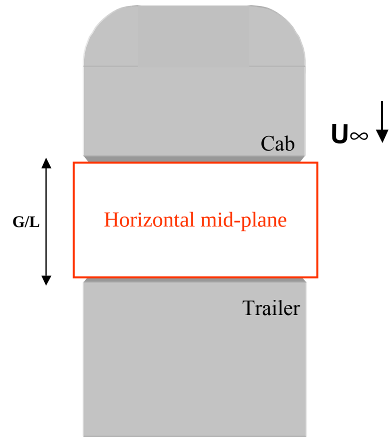
## USC Experiment, Gap Flow Investigation

PIV measurements in horizontal mid-plane

$$\mathrm{Re} = \frac{U\sqrt{A}}{\nu} = 300{,}000$$

$A$ - Frontal area

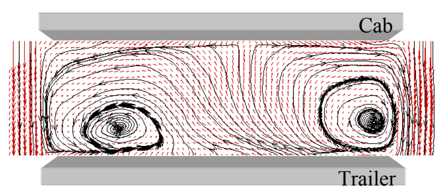
$$\frac{G}{L} = \frac{G}{\sqrt{A}}$$




## USC Experiment, Small Gap

**Time-averaged streamline patterns**

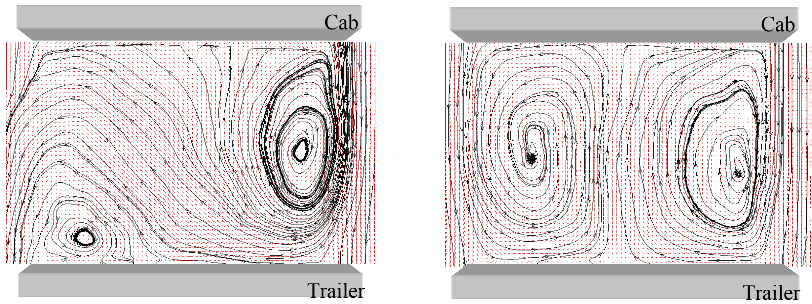
G/L = 28%




Symmetric flow

## USC Experiment, Large Gap

**Time-averaged streamline patterns, G/L = 75%**



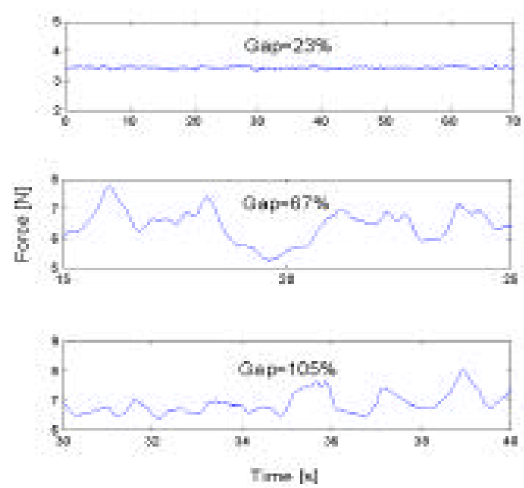

Asymmetric flow | Symmetric flow

**Gap flow structure is sensitive to the condition of the shear layer**
**Large side force may be present in the asymmetric flow case**

## USC Experiment, Time History of Drag Force

Time signature of drag force on trailer as a function of gap size

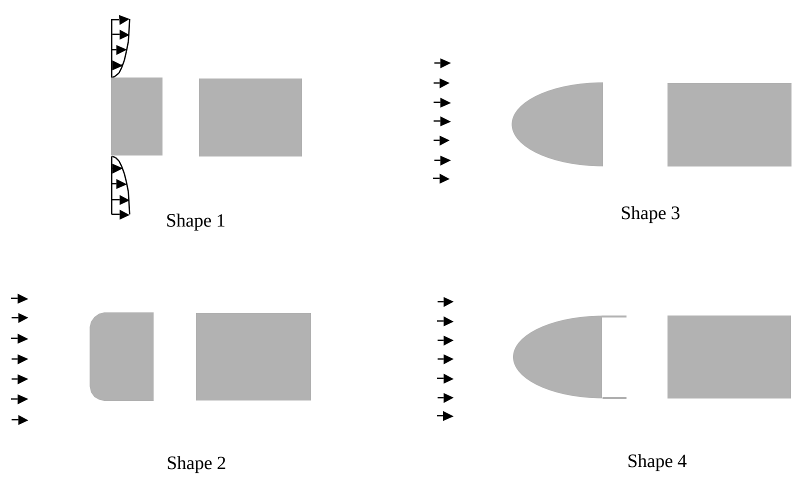
Re = 305,000



## Gap Flow Simulation, Computational Approach

- Gap flow from experimental observation is clearly three-dimensional
- Perform 2-D simulations to determine proper length and time scales needed to resolve flow structures in the gap
- Given the knowledge of the 2-D calculations perform 3-D simulations
- The computational domain is setup to capture the gap and part/all of the tractor and part of the trailer geometry

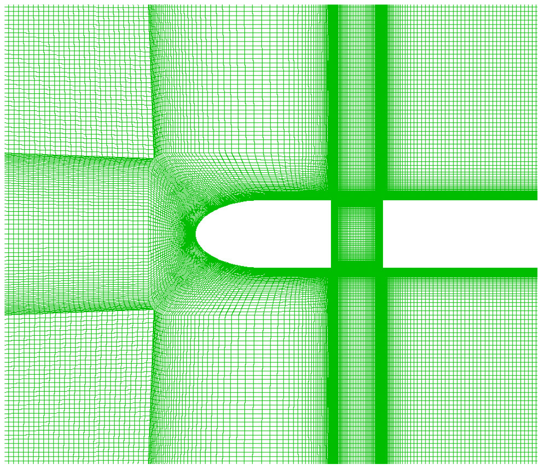
## Computational Domain and Geometry



## Computational Mesh for G/L at 72%

Unstructured Mesh

40,000 elements

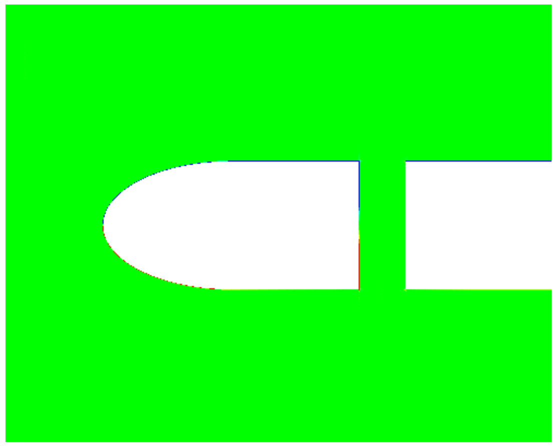


## Gap Flow Simulation Matrix

| G/L | Smagorinsky | Smagorinsky with Van Driest Damping |
|---|---|---|
| 35% | Completed | Completed |
| 72% | Completed | Completed |
| 72% with side extenders | Completed | Completed |

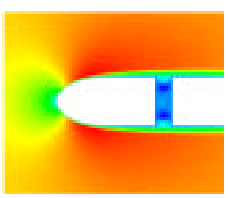
## Gap Flow Simulation, G/L of 35%

Smagorinsky

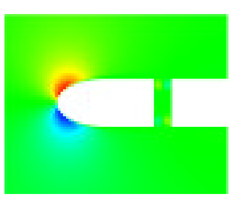


## Gap Flow Simulation, G/L of 35%

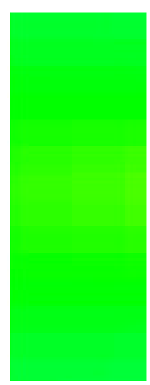
Time-averaged results, Smagorinsky



U-comp.



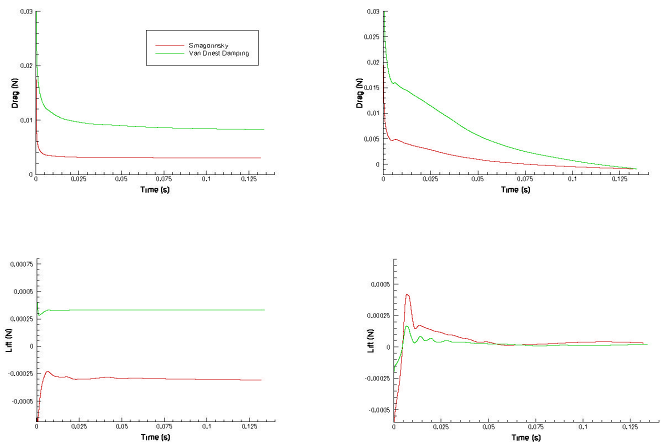
V-comp.



Pressure

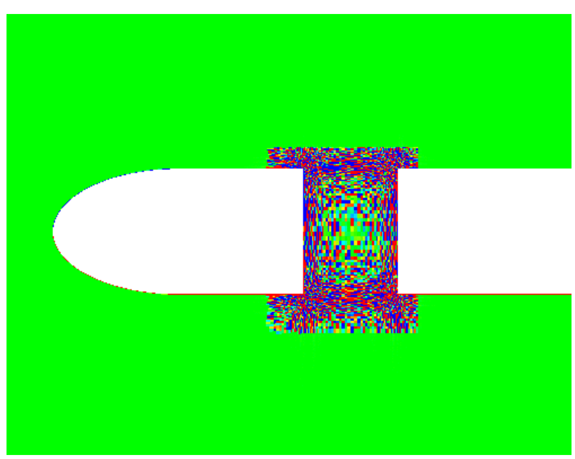
## Time History of Drag and Lift, G/L of 35%

**Tractor** | **Trailer**



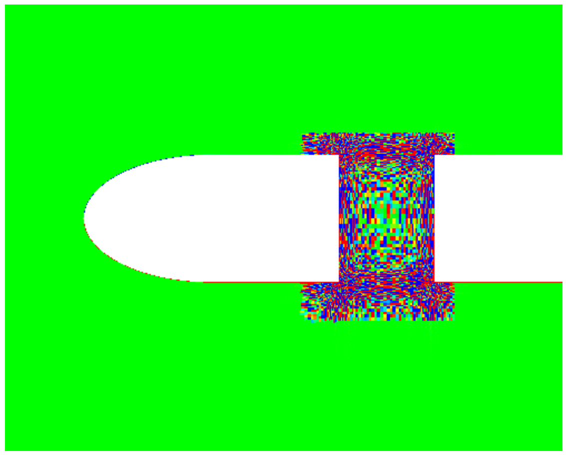

## Gap Flow Simulation, G/L of 72%

**Smagorinsky**



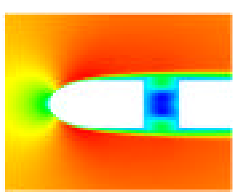
## Gap Flow Simulation, G/L of 72%

Smagorinsky with van Driest damping

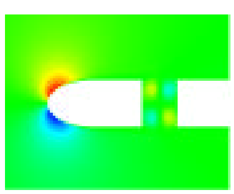


## Gap Flow Simulation, G/L at 72%

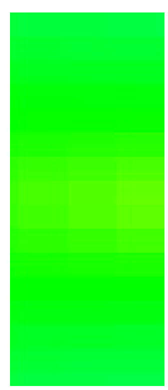
Time-averaged results, Smagorinsky



U-comp.



V-comp.

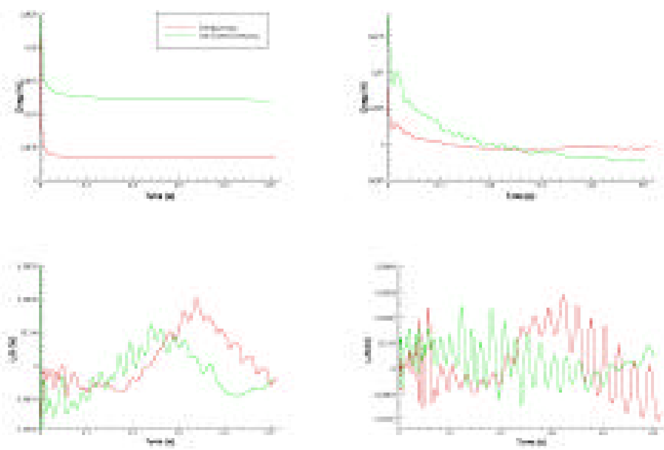


Pressure

## Time History of Drag and Lift, G/L of 72%

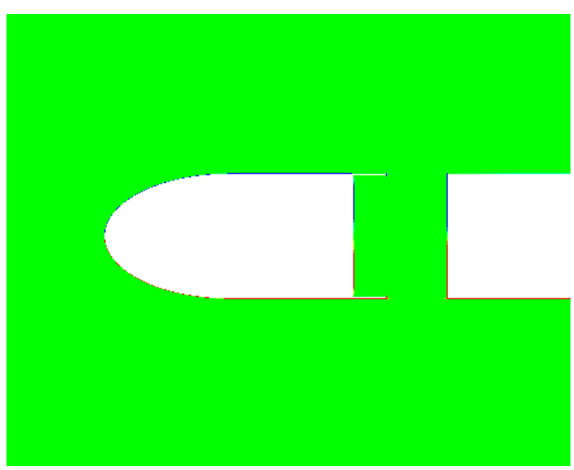
Tractor

Trailer



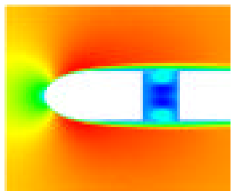
## Gap Flow Simulation, G/L at 72% with Side Extenders

Smagorinsky

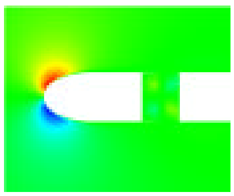


## Gap Flow Simulation, G/L at 72% with Side Extenders

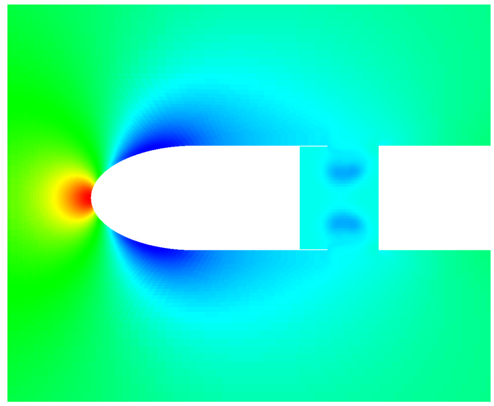
**Time-averaged results, Smagorinsky**



U-comp.



V-comp.

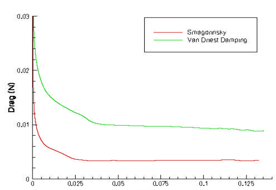


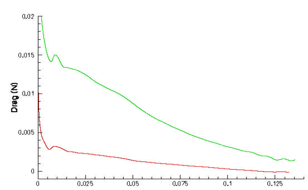
Pressure

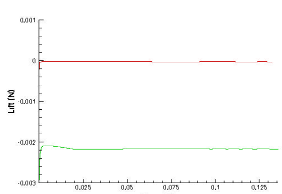
## Time History of Drag and Lift, G/L of 72% with Side Extenders

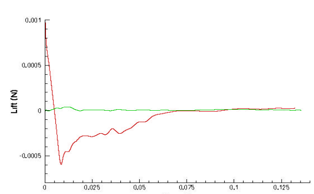
**Tractor**

**Trailer**



## Summary

- Implicit and Semi-Implicit projection methods have been implemented in ALE3D. Anticipate significant speedup with all simulations
- Initiated gap flow study with gap distances below and above the critical distance, G/L of 50%. Also, investigated the impact of side extenders on gap flow structure
- Initiated Trailer wake flow simulation with/without boattail to investigate the wake structure and its interaction with ground plane
- OVERFLOW was utilized with its overset grid capability to model NASA 7'x10' tunnel for boundary condition determination
- An overset mesh which is a modular mesh is under construction for the tractor-trailer geometry in the NASA 7'x10' tunnel

# Validation Cases and Truck Wake Simulations with ALE3D

Jason Ortega, Tim Dunn, Dora Nakafuji
Rose McCallen, Kambiz Salari

Lawrence Livermore National Laboratory

Computational Physics
Fluid Dynamic Applications

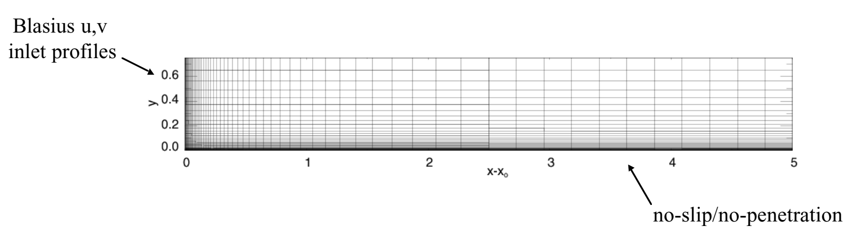
# Overview

- Validation Test Cases with ALE3D
  - Flat Plate
  - Circular Cylinder
- 2-D Truck Wake Simulations
- Summary

# Validation Cases with ALE3D

## Validation Case Flat Plate

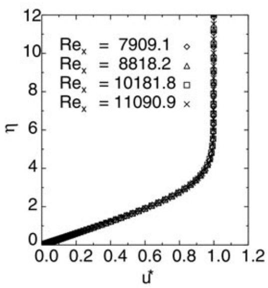
Testing ***viscous growth*** of a boundary layer and ***shear stress prediction***



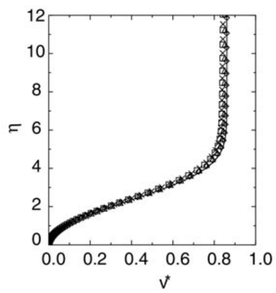

- $Re_{x\ inlet} = 2{,}000$
- ***Explicit*** and ***implicit*** time-integration schemes
- Coarse grid: 2,440 elements
- Medium grid: 9,760 elements

## Validation Case Flat Plate

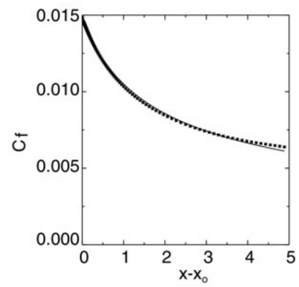
Horizontal Velocity




Vertical Velocity




Shear Stress Coefficient



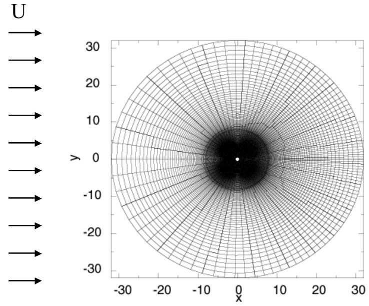

Good representation of ***laminar boundary layer development*** with ALE3D

## Validation Case Circular Cylinder

Testing ***unsteady vortex shedding*** and ***drag prediction***

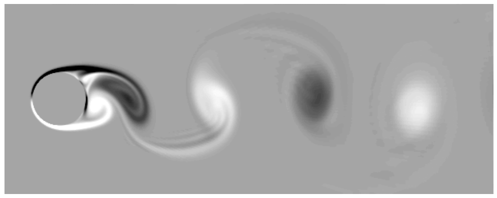
U

uniform inlet velocity profile




- $Re_d$= 1,000
- ***Explicit*** time-integration scheme
- Coarse grid: 20,000 elements
- Medium grid: 80,000 elements

## Validation Case    Circular Cylinder



Measured Quantities

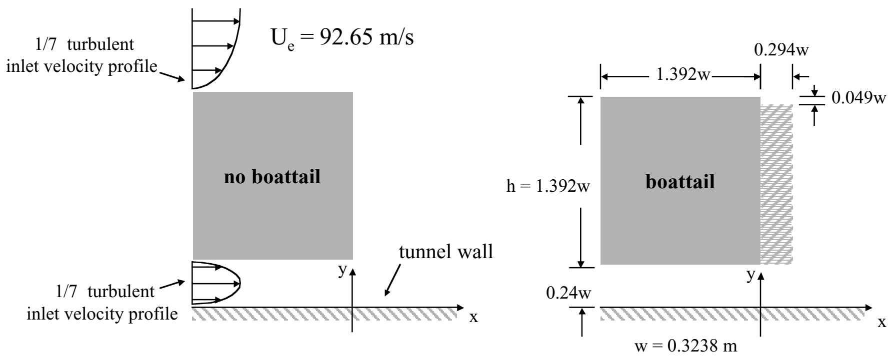
| Grid | $C_d$ | $C_l$ | $St = fD/U_\circ$ |
|---|---|---|---|
| Coarse | 1.4429 | -0.001044 | 0.2288 |
| Medium | ***1.5021*** | -0.000026 | ***0.2394*** |
| Qian & Vezza | 1.52 | -- | 0.24 |
| Blackburn *et al.* | 1.51 | -- | -- |
| Behr *et al.* | 1.53 | -- | 0.241 |
| He *et al.* | 1.5191 | -- | -- |

Capturing ***drag forces*** and ***laminar vortex shedding*** with ALE3D

## 2-D Truck Wake Simulations

## Computational Setup

Investigating ***length scales***, ***vortex dynamics***, influence of the ***ground plane***, and the ***effect of add-on devices***

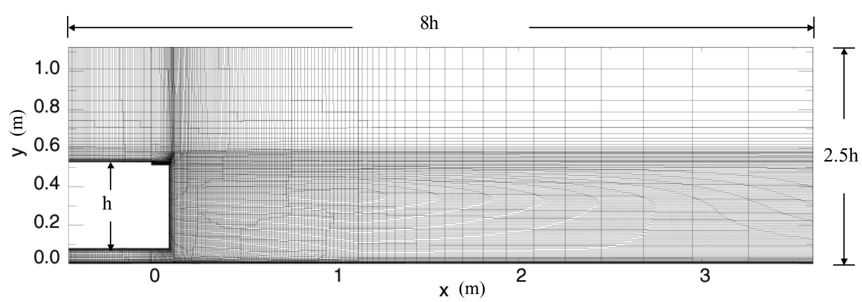



- 2-D simulation with ALE3D
- LES with Van Driest damping
- $Re_w = 2 \times 10^6$

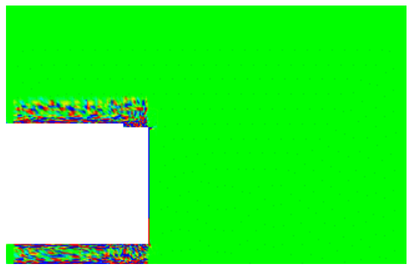
## Computational Grid

**Boattail**

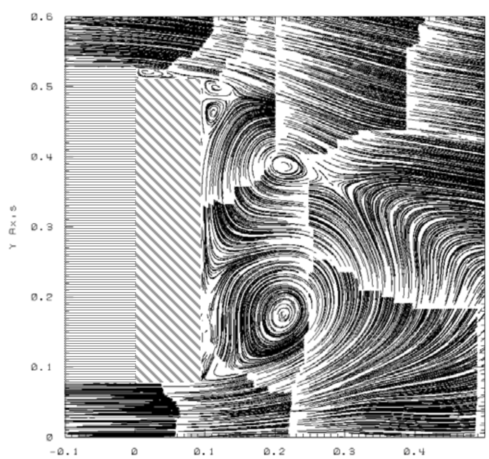
19,445 elements



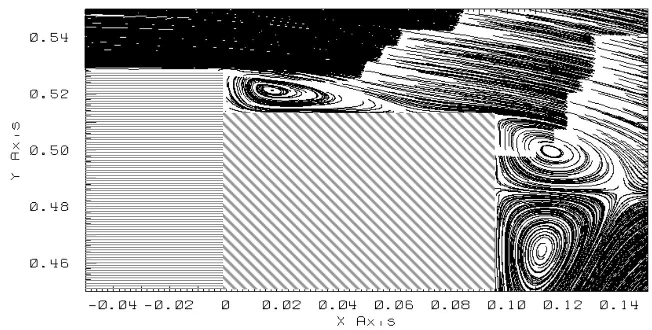
# Vorticity Measurements



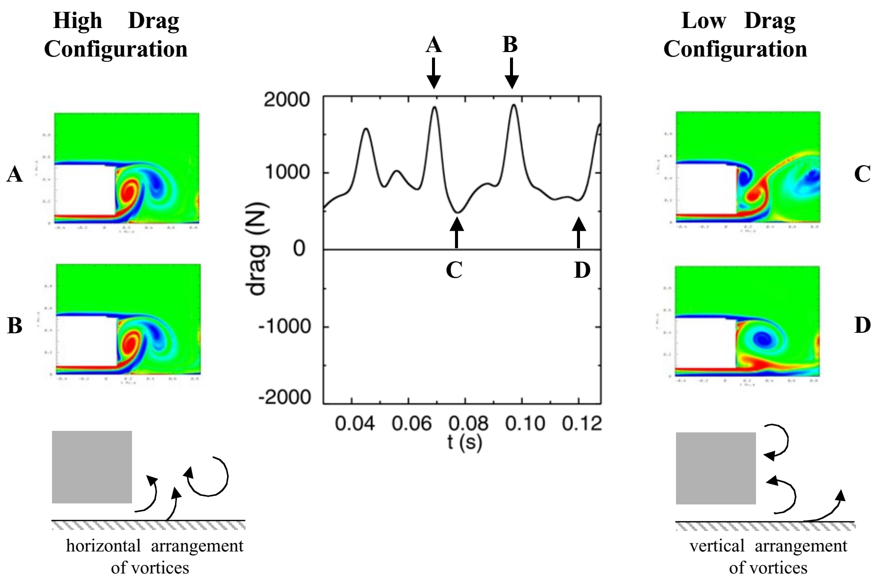
# Average Streamline Fields



# Average Streamline Fields




# Drag Measurements



# Summary

- Validation Cases with ALE3D
  - ***Velocity profiles*** and ***shear stress coefficient*** from the flat plate simulation compare favorably with those from the ***Blasius solution***
  - ***Drag coefficient*** and ***shedding frequency*** from the circular cylinder simulation show good agreement with results in the ***literature***
- 2-D Truck Wake Simulations
  - Capture the ***unsteady nature*** of ***vortex shedding*** in the wake
  - ***Drag*** is ***strongly influenced*** by the ***arrangement of the vortex patches*** in the near wake
  - Set the ***groundwork*** for future 3-D simulations by determining the ***length scales*** and ***required resolution*** of the flow field

# Full Vehicle Simulation Using OVERFLOW & Overset Tools

Dora Nakafuji, Jason Ortega, Tim Dunn,
Rose McCallen, Kambiz Salari

Lawrence Livermore
National Laboratory

# The Bridge

ALE3D/SACCARA
Turbulence Models
CFD code development

Configuration studies
Overflow/Overset Tools

Industry codes
Production expertise
Processes/methods

CONTINUING RESEARCH & DEVELOPMENT

APPLIED R&D

INDUSTRY APPLICATION & PRODUCTION DESIGN

# Motivation

- Robust and well tested RaNS code
- Provide a secondary tool for evaluating flow models and experimental results
- Incorporate Overset techniques & capabilities in simulation
- Build in modular & interchangeability into grid development process

Break-up complex geometries + Independence in generating grids = Simplification & Increased Capability

# Objectives

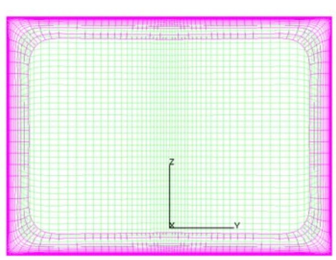
- Integrate benefits of Overflow & Overset grids
  - RaNS speed and near wall modeling capabilities
  - Gain experience using empty tunnel configuration
  - Apply tools to GCM simplified model
- Use Overset modular capability to analyze multiple truck configurations (gap, side angle) and complex geometries
- Address industry analysis needs by quantifying simplification on grid generation and establish methodologies for modular analysis

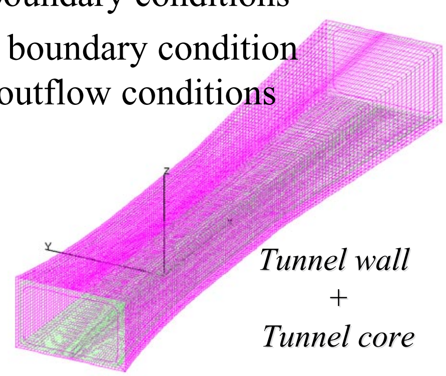
## Accomplishments

- 3-D empty tunnel simulations
  - Viscous boundary conditions along all walls
  - Overset grid (approx. 1 million pts)
- Strengthened ties with collaborators
  - Leveraged grid generation resources (NASA Ames, LLNL Overture Group)
  - Fast-tracking knowledge transfer of Chimera techniques
- Develop Overset grids GCM truck
  - Used tried&true grids (collars, caps)
  - Integrated interchangeability into grid design
  - Potentially refined & reduced grid complexity (approx. 4 million pts)

## Empty Tunnel Grid

- 3-D Overset grid – O-core with rectangular wrap on tunnel wall
- Simplified and reduce boundary conditions
- Consistent viscous wall boundary condition and appropriate inflow/outflow conditions



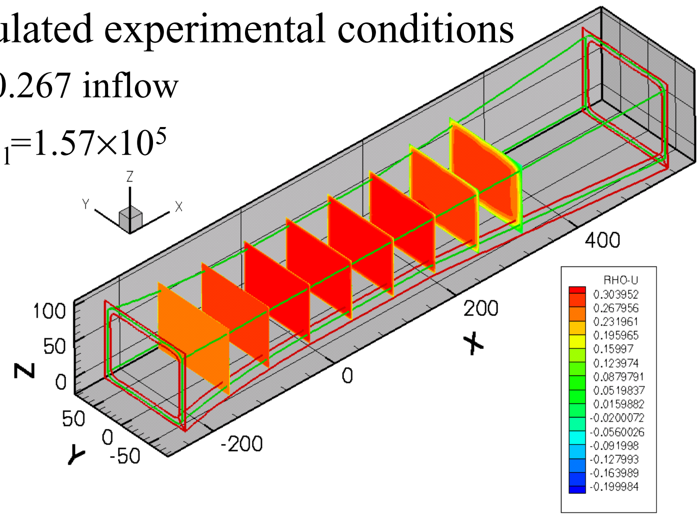


# Empty Tunnel Simulation

- Simulated experimental conditions

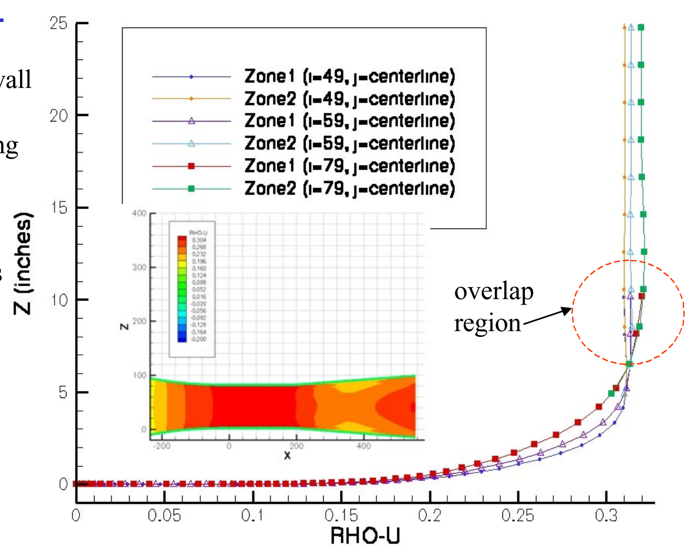
  M=0.267 inflow

  $Re_l=1.57\times10^5$



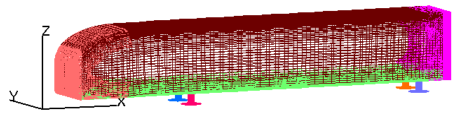

# Simulation Results

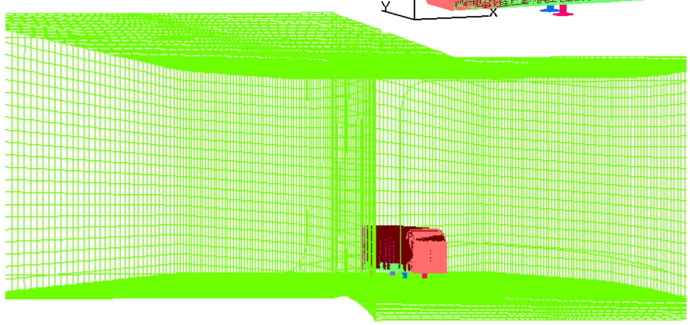
- Profiles at the wall in the tunnel test-section taken along the y-centerline
- Zones indicate multiple grids
- Overlap regions are consistent



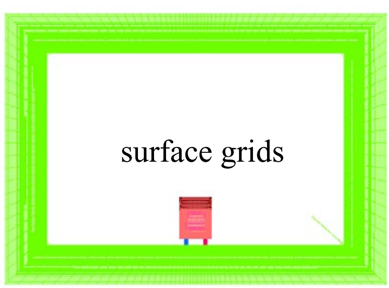
# GCM Truck Grids

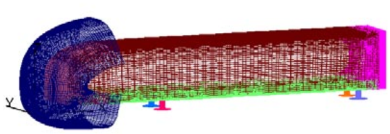
- Pre-liminary grids for truck (approx 4 mil pts)
- 8 multi-grids, optimized spacing and clustering to surface
- Minimal tunnel changes to accommodate truck



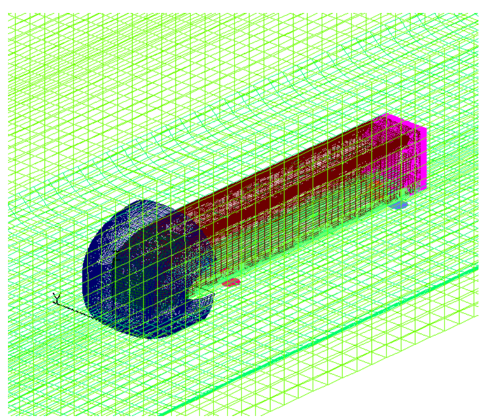


# Truck with Tunnel Grids



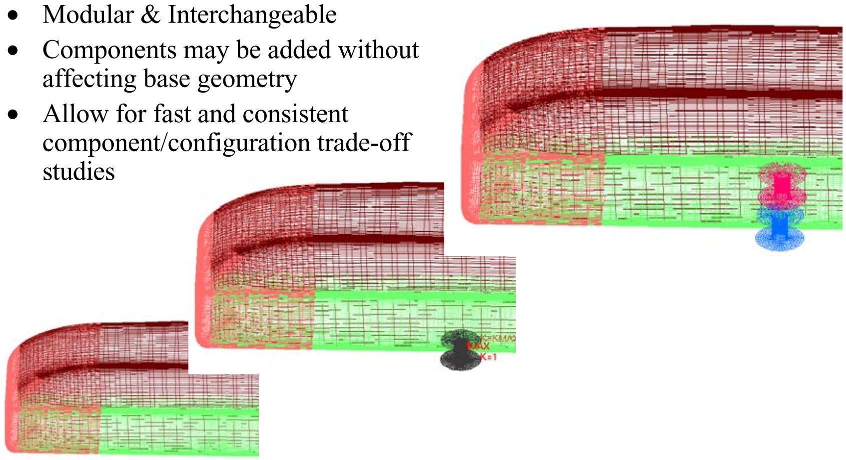



Generation of front cap volume grid



# Grid Interchangeability

- Modular & Interchangeable
- Components may be added without affecting base geometry
- Allow for fast and consistent component/configuration trade-off studies



# Caltech Heavy Vehicle Aerodynamics Computational Group

Prof. Tony Leonard

Demosthenes Kivotides, Postdoctoral Scholar

Mike Rubel, Graduate Student

Philippe Chatelain, Graduate Student

Graduate Aeronautical Laboratories · California Institute of Technology

April 3-4 2002

# Vortex Code: Essentials

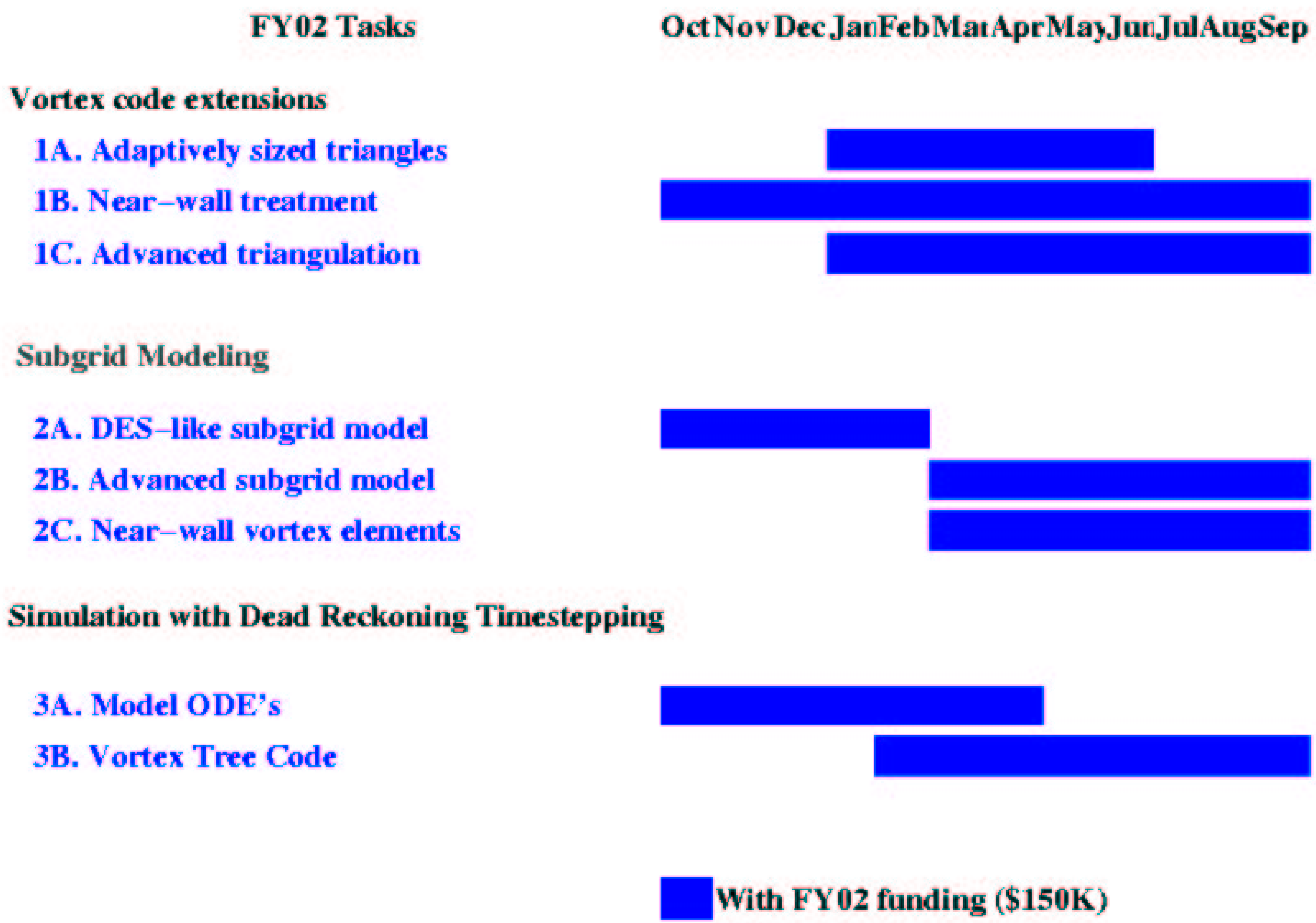
- Numerical technique to solve the Navier-Stokes equations
- Suitable for Direct Simulation and Large-Eddy Simulation
- Uses vorticity ($\vec{\omega} = \nabla \times \vec{u}$) as the solution variable
- *Lagrangian*: computational elements move with fluid velocity
- Viscous, 3-D, incompressible, with boundaries

# Vortex Code: Advantages

- Computational elements only where vorticity is nonzero
- No grid in the flow field
- Only 2-D grid on the vehicle surface
- Boundary conditions in the far field automatically satisfied

*now: examples of vortex particle codes in action*

# Caltech FY02 Planned Work



# Current Research Topics Topics

- Boundary geometry (GTS model, USC geometry, others)
- Near-wall treatment
- Dead-Reckoning time integration algorithm
- Vortex filament methods
- SGS / LES models
- Face-centered cubic lattice

# Geometry and Boundary-related Research

- Need to know information such as closest-point, closest-panel, inside/outside
- Traditionally limited to simple shapes like spheres and cubes
- GTS geometry requires more robust approach
- Implementing half-edge data structure
- Possibly novel tree-based algorithms for the above

# Near-Wall Treatment

- Particles good approximation for field in free-space, but not near wall
- Near-wall Eulerian treatment, local grid, ”thick” boundary
- Match to particles, LES further afield
- Some low-D progress; trying to expand

# The Dead Reckoning Algorithm

$$\frac{dX}{dt} = F(X, t) \qquad\qquad X(t_0) = X_0$$

- Conventional time integrator integrates every variable, every timestep
- Implemented new algorithm that automatically adjusts step size
- Closely related to dead reckoning network games algorithm
- See mass-spring animations
- How to make it work with our fast multipole tree?

# Vortex Filament Method

- Instead of particles, discretize vorticity on set of filaments
- Automatically divergence-free
- Efficient in free-space
- Boundary compatiblity work in progress
- Use to model flow in wake?

# Additional topics

- LES models for vortex methods
- FCC lattice for remeshing

# Commercial CFD Code Validation for Heavy-Vehicle Aerodynamics Simulations

David Pointer, Tanju Sofu, David Weber - Argonne National Laboratory
Everett Chu, Paul Hancock, Bob Bundy - PACCAR Technical Center

Heavy Vehicle Aerodynamic Drag Team Meeting
LLNL, April 3-4, 2002

## Background

- Next generation of computational methods/tools are currently being developed under the DOE's Heavy Vehicle Aerodynamic Drag Program
  - focus on specific turbulence and flow separation problems unique to heavy vehicle external aerodynamics
  - a wide range of turbulence modeling options
  - experimental program to support V&V efforts
- Specific elements of the program
  - long term focus
  - need for massively parallel high-performance computers
  - need for extensive verification and validation based on simple geometries

## Objectives of ANL Effort

- Assessment of commercial CFD capabilities for heavy-vehicle aerodynamics
  - extend general purpose turbulence models to aerodynamics applications
  - investigate standard turbulence models and some of the novel turbulence modeling capabilities
- Specific elements of the activity
  - near term focus (address immediate needs of manufacturers)
  - reduced reliance on high-performance computers (compatible with OEMs' computational resources)
  - realistic heavy vehicle geometries (full details of a specific design)
- Initial contacts with manufacturers indicate support and interest
  - CRADA application with Paccar Technical Center, and interactions with Freightliner
  - Strong CFD industry support (particularly from CD/adapco and EXA)

## Commercial CFD Software

- Common advantages
  - ability to model complex geometries with selective mesh refinement (unstructured grids)
  - extensive V&V work by developers and user community for a wide range of CFD applications
  - reduced need for large scale computer systems
  - development and technical support from the vendor
- Common issues
  - insufficient accuracy, high cost
  - need for CFD specialists familiar with specific software
  - need for assessment of codes' strengths/weaknesses
    - standard turbulence models generally validated for automobile industry, but assessment needed for heavy-vehicles
    - validation and assessment needed for novel turbulence models

## ANL - PACCAR CRADA

- Detailed geometry for identified vehicle configurations (Peterbilt-379 selected as the base model)
- 1/5-scale wind tunnel tests in University of Washington with selected configurations
- Assessment of STAR-CD (and possibly PowerFlow) software
- 18 month, $600K plan (equal contributions by each partner)

## ANL - PACCAR CRADA (cont.)

- Phase-I work
  - Build 1/5-scale model of the base configuration and conduct wind tunnel tests
  - Collect, organize, and process the experimental data
  - Assess standard RANS model of STAR-CD (high Reynolds number form of _-_ equations in conjunction with the “law of the wall” representation of flow)
  - Blind predictions of the flow field to avoid "tuned" solutions
- Phase-II work
  - Fine-tuning of STAR-CD model
  - Assessment of more detailed turbulence modeling options to address the limitations of the standard RANS model
  - Analysis of additional heavy-vehicle configurations (new designs)
- Possible extension of work for assessment of PowerFlow

# Summary

- Provide an independent and comprehensive evaluation of commercial CFD capabilities to address near-term goals of the Heavy-Vehicle AeroDrag Program
  - realistic and prototypic 3-D geometries and operating conditions
  - close collaboration with PACCAR to address their current needs
  - CFD industry support
- Deliver a summary of best practice guidelines for application of current commercial CFD capability to heavy vehicle industry.

21st Century Truck Partnership

*"Unprecedented Collaboration for Unparalleled Results"*

# Aerodynamic Combination Vehicle Test Update/DOE

*Georgia Tech Research Institute*
*Great Dane Trailers*
*Volvo Trucks North America*

*/Skip Yeakel, P.E.*

*Aerodynamic "SWAT Team" Meeting*
*@ Lawrence Livermore National Lab, Livermore, CA*
*April 4, 2002*

## Aerodynamic Combination Vehicle Team Members

*21st Century Truck Partnership*

- Georgia Tech Research Institute
  Atlanta, GA
- Great Dane Trailers
  Savannah, GA
- Novatek
  Atlanta, GA
- Volvo Technology of America
  Greensboro, NC
- Volvo Trucks North America
  Greensboro, NC

## "Tuning" Test Process & Prospects - Greensboro, NC

*21st Century Truck Partnership*

- **Abbreviated road course**
  - 65 mph speed limited section (~ 5.5 mi.) of U.S. Route 311 south of I-40
  - Quick cycle time - 7 test variables/14 runs completed on March 1st, 2002
  - Constant speed runs/adaptive cruise control operation--north and south
  - Minimal traffic = no runs lost (100% yield)/ March 1 (public highway)
- **Prospects**
  - "Flavor" for TRC testing but results not statistically significant
  - Volvo VN Integral Sleeper™ ("660" model - seats driver + 3 observers) -|
  - "Quick turns" = more exciting than watching paint dry (e.g. TRC) V
  - Economical = federal highway road course (a/k/a "free")
  - Better weather prospects vs. Ohio

- **Limitations**
  - Not flat (rolling hills)--hard to integrate spikes, some traffic, speed limited (65 mph), too short for statistically significant results (un-"scientific")
  - Experimentally "impure"…baseline was NOT "stock" trailer; too painful!

## TRC Hopes (and Expectations)

*21st Century Truck Partnership*

- **Akin to watching paint dry--if all goes well…a plethora of angry trucker language likely if not!**
  - No place for a cast of thousands, 800, or even...eight!
  - Watching wind blow or rain fall is neither fun nor productive.
  - ~ 450 miles per data point--requires man/machine harmony and incredible patience possessed by few.
  - Once cruise control is set, the driver has only one task…to stay within the assigned lane
- **55, 65, and 75W (or 60, 70E, and 80?--concerns!) mph test runs**
- **Results that are even half as good as predictions…no apologies needed if xx% net fuel savings can be proven!! CAUTION urged re NC "tuning" data…usefulness is software limited and should only be construed as serving the tuning purpose intended. The TRC site and "high tech" (NOT!) buckets of fuel and stopwatches are still the best (ONLY!) way to get precious, tedious, datapoints.**
- **Don't try this work at home--very few such sites around the globe!**

## Aerodynamic SWAT Team CFD Wish List

*21st Century Truck Partnership*

- Reduce cost of current CFD tools for industry/society benefit.
- Aerodynamics lasts for the life of a truck--for better or worse! Seek out optimal, and practical, design concepts with industry.
- Advance the art of the aerodynamicist--more near term blood in that turnip than in environmentally squeezed IC engines.
- Provide economical tool to judge add-on devices--a better way to separate good product concepts from snake oil.
- Maintain/expand aerodynamic R&D community and relationships; east (e.g. Langley FST, PSU+) and west (current+)…even global?!
- Correlate/coordinate with industry partners and established methodologies (road AND wind tunnel tests).
- Partner with/support other agencies for common cause via cohesive NEP (e.g. EPA/DOT “Ground Freight,” DOD “Army Transformation,” et al) under 21st Century Truck Partnership umbrella with all (16) industry partners (incl. ALL truck OEMs).

## Undesirable Aero SWAT Team Product

*21st Century Truck Partnership*

Solicit “voice of the customer”--don’t create in a vacuum!

## Undesirable Aero SWAT Team Goal

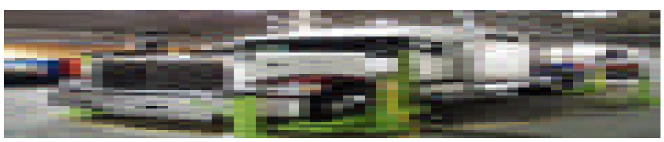
*21st Century Truck Partnership*

**Common truck design by committee--no matter how fuel efficient!**

## Aerodynamic Prospects and Importance

*21st Century Truck Partnership*

**Aerodynamics or more wasted OFFSHORE oil? It's an American choice to make.**



**Combining forces will help us get it together. We have made a good start…the best is yet to come!**


## Thanks, and be thankful for the opportunities before us!

## --Questions and Answers--

VOLVO

New Roads.™

*Aero "SWAT Team" Meeting @ LLNL, Livermore, CA*
April 4, 2002